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GROUND PROXIMITY TESTS OF THE 0.03-SCALE MODEL  
(45-0) SPACE SHUTTLE ORBITER IN THE ROCKWELL  
INTERNATIONAL NAAL LOW SPEED WIND TUNNEL  
(OA250)

by

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Prepared under NASA Contract Number NAS9-13247

by

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Houston, Texas

WIND TUNNEL TEST SPECIFICS:

Test Number: NAAL 775  
NASA Series Number: OA250  
Model Number: 45-0  
Test Dates: July 1 through 7, 1977  
Occupancy Hours: 3<sup>4</sup>

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R. Mennell, Rockwell International

ABSTRACT

Experimental aerodynamic investigations were conducted on a triple strut-mounted 0.03-scale representation of the 140A/B (modified) space shuttle orbiter configuration in the Rockwell International 7.75 x 11.00 foot low speed wind tunnel during the time period from July 1 to July 7, 1977. NASA designation for this test was OA250; facility number was NAAL 775.

The primary test objectives were to define orbiter lateral-directional stability characteristics in ground proximity and to investigate discrepancies in lateral-directional data obtained during previous NAAL tests. In order to accomplish the above objectives, the orbiter model was mounted at various heights ( $h/b$ ) above the NAAL ground plane at a fixed angle of attack ( $10^\circ$ ) and sideslipped through an angle **range of**  $-5^\circ \leq \beta \leq 5^\circ$ . Model  $h/b$  heights consisted of 0.20, 0.178, 0.125, 0.10 and 0.075 at elevon deflections of  $0^\circ$  and  $\pm 5^\circ$ . For this test period, a six-component Task internal strain-gaged balance (1.5" MK XIVA) was utilized to record all aerodynamic force data. No pressure data were recorded, and all data presented are uncorrected for base pressure effects.

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PLOTTED COEFFICIENTS SCHEDULE:

(A)  $C_Y$ ,  $C_L$ ,  $C_n$  VERSUS  $\beta$

## NOMENCLATURE

<u>SYMBOL</u>	<u>PLOT SYMBOL</u>	<u>DEFINITION</u>
a		speed of sound; m/sec, ft/sec
$C_A$	CA	forebody axial-force coefficient
$C_D$	CD	drag coefficient
$C_L$	CL	lift coefficient; $\frac{\text{lift}}{qS}$
$C_\ell$	CBL	rolling-moment coefficient; $\frac{\text{rolling moment}}{qS_b}$
$C_n$	CYN	yawing-moment coefficient; $\frac{\text{yawing moment}}{qS_b}$
$C_N$	CN	normal-force coefficient; $\frac{\text{normal force}}{qS}$
$C_m$	CLM	pitching-moment coefficient; $\frac{\text{pitching moment}}{qS_\ell \text{REF}}$
$C_Y$	CY	side-force coefficient; $\frac{\text{side force}}{qS}$
M	MACH	Mach number; V/a
p		pressure, psf
q	Q(PSF)	dynamic pressure, psf
RN/L	RN/L	unit Reynolds number; per m, per ft
$T_t$	TT	freestream total temperature °R
V		velocity, ft/sec
$\alpha$	ALPHA	angle of attack, degrees
$\beta$	BETA	angle of sideslip, degrees
$\delta_{BF}$	BDFLAP	body flap deflection
$\delta_a$	AILRON	aileron deflection
$\delta_E$	ELEVON	deflection of elevons

## NOMENCLATURE (Concluded)

<u>SYMBOL</u>	<u>PLOT SYMBOL</u>	<u>DEFINITION</u>
	XCP/L	center of pressure location, percent of body length, XMRP/LB-(CLM)(LREF)/(CN)(LB)
$\delta_R$	RUDDER	rudder deflection
$\delta_{sb}$	SPDBRK	speedbrake deflection, degrees

### Reference and C. G. Definitions

b	BREF	wing span or reference span; m, ft
$\bar{c}$		aerodynamic chord; m, ft
$l_{REF}$	LREF	reference length or wing mean
S	SREF	wing area or reference area, ft <sup>2</sup>
	MRP	moment reference point
	XMRP	moment reference point on X axis
	YMRP	moment reference point on Y axis
	ZMRP	moment reference point on Z axis

- - - - -

### SUBSCRIPTS

b	base
l	local
s	static conditions
t	total conditions
$\infty$	freestream

- - - - -

h/b	H/B	orbiter height over ground plane to orbiter wing span ratio
-----	-----	---

## CONFIGURATIONS INVESTIGATED

The model utilized for test period OA250 was a 0.03-scale representation of the 140A/B space shuttle orbiter configuration. The basic model was of the blended wing-body design utilizing a double delta wing ( $75^\circ/45^\circ \Delta_{LE}$ ), full span, dual panel elevons (unswept hingeline and 6" gaps), a centerline vertical tail with rudder and/or speed brake deflection capability, a canopy, a body flap, and an orbital maneuvering system (OMS pods) mounted on the aft fuselage sidewalls adjacent to the vertical tail. Landing gear simulation consisted of both nose and main landing gear with either full extended or full retracted capability.

The model was constructed primarily of fiberglass with wing panels of aluminum and consisted of both non-metric and metric sections. The model non-metric section was supported by a triple strut structure that was rigidly mounted to the test section floor. Each strut consisted of a threaded rod with a pinned clevis joint allowing variable h/bw settings to be obtained.

For this test period the following nomenclature was used to designate the various model components:

<u>Component</u>	<u>Description</u>
B <sub>26</sub>	Orbiter fuselage per Rockwell International lines VL70-000140A/B
C <sub>9</sub>	Orbiter canopy per Rockwell International lines VL70-000140A/B
E <sub>43</sub>	Orbiter full-span, unswept hingeline, 6" gapped elevons per Rockwell International lines VL70-000200

Configurations Investigated (Concluded)

<u>Component</u>	<u>Description</u>
F <sub>8</sub>	Orbiter body flap per Rockwell International lines VL70-000200
G <sub>15</sub>	Orbiter landing gear per Rockwell International lines VL70-000140A/B
M <sub>16</sub>	Orbiter OMS pods per Rockwell International lines VL70-000203A
N <sub>24</sub>	Orbiter main propulsion system nozzles per Rockwell International lines VL70-000140A
N <sub>28</sub>	Orbiter OMS nozzles per Rockwell International lines VL70-000140A
R <sub>5</sub>	Orbiter rudder per Rockwell International lines VL70-000146A
TC <sub>19</sub>	Orbiter tail cone fairing per Rockwell International lines BCD V70-30-330-02
V <sub>8</sub>	Orbiter vertical tail per Rockwell lines VL70-000146A (Model Drawing SS-A01360)
W <sub>116</sub>	Configuration 4, Rockwell International Drawing No. VL70-000140A, -000200

## TEST FACILITY DESCRIPTION

North American Aerodynamics Laboratory (NAAL) 7.75 x 11-foot Wind Tunnel is a continuous flow, closed circuit, single return tunnel capable of speeds up to 200 miles per hour.

The test section is vented to atmospheric pressure and is 7.75 feet high, 11 feet wide, and 12 feet long. Power is supplied by a 1250-horse-power nacelle-mounted synchronous motor driving a 19-foot, seven-blade, laminated birch propeller. Airspeed is controlled by using a magnetic clutch to vary the degree of coupling between the motor and propeller. Turbulence is minimized by a damping screen and a honeycomb section in the settling chamber upstream from the contraction cone (ratio 7.53 to 1).

Tests may be conducted using a variety of mounting systems: single strut, double strut, sting strut, reflection plane, cable suspension, or two-dimensional wall. Aerodynamic data may be measured by a planar type external balance system or sting-mounted internal balances. An Astrodata Automatic Data Acquisition System collects, multiplexes, digitizes, and records on magnetic tape 50 channels of force or pressure data or both. Data are then reduced and plotted using automatic data processing equipment and an automatic digital plotter.

The NAAL Wind Tunnel has been operating since June 1943. Calibrations are available over a wide range of test conditions.

## DATA REDUCTION

The aerodynamic force data presented in this report were measured by the Task Corporation 1.5 inch diameter MK XIVA internal strain gage balance. Corrections applied to the aforementioned data consisted of model tunnel blockage effects on test section dynamic pressure. No pressure measurements, other than tunnel parameters, were recorded and all data are uncorrected for base pressure effects.

## REFERENCES

1. SD77-SH-0176, Rockwell Report, "Pretest Information for Ground Proximity Tests of the 0.03-Scale Space Shuttle Orbiter Vehicle in the Rockwell International NAAL Low Speed Wind Tunnel Test," by E. Chee, dated June 28, 1977.
2. IL, SAS/AERO/77-291, "Test Requirements for a New Test in Rockwell/NAAL in Verify Ground Effects on Lateral/Directional Stability," dated June 8, 1977.
3. IL, SAS/WTO/75-034, Addendum #8, "NAAL Ground Plane Tests of the SSV 45-0 Model," dated June 8, 1977.
4. SAS/AERO/77-402, Rockwell International letter, "Response to OV101 ALT Aero CDR RID 30-A, 'Directional Instability During Last Three Seconds of Flight'," by W. M. Zeman, dated July 13, 1977.

TABLE I.

TEST : OA250

DATE : 7/7/77

## TEST CONDITIONS

MACH NUMBER	REYNOLDS NUMBER (per foot)	DYNAMIC PRESSURE (pounds/sq. inch)	STAGNATION TEMPERATURE (degrees Fahrenheit)
0.20	$1.42 \times 10^6/\text{ft.}$	$60 \text{#/ft.}^2$	$80^\circ \rightarrow 120^\circ \text{F}$

BALANCE UTILIZED: 1.5" TASK MK XIV A

	CAPACITY:	ACCURACY:	COEFFICIENT TOLERANCE:
NF	<u>130 lb/gage</u>	<u>0.25%</u>	_____
SF	<u>80 lb/gage</u>	<u>0.25%</u>	_____
AF	<u>40 lb/gage</u>	<u>0.25%</u>	_____
PM	_____	_____	_____
RM	<u>150 in-lb.</u>	<u>0.25%</u>	_____
YM	_____	_____	_____

COMMENTS:

## TEST : OA250 NAAK 725

TABLE II.  
DATA SET/RUN NUMBER COLLATION SUMMARY

DATE : 7/7/77

DATA SET IDENTIFIER	CONFIGURATION	SCHED.		NO. OF RUNS	MACH NUMBERS (OR ALTERNATE INDEPENDENT VARIABLE)
		$\alpha$	$\beta$		
00001	1 -	10	A	+5	.20
0002	1 - G/5	0	0	0	1
0003	1	1			2
0004		+5			3
0005		0			4
0006		-5			5
0007		0			6
0008		+5			7
0009		-5			8
0010		-5			9
0011		+5			10
0012		+5			11
0013		0			12
0014		0			13
0015		0			14
0016		0			15
0017		+5			16
0018		-5			17
					18
1					
7					
13					
19					
25					
31					
37					
43					
49					
55					
61					
67					
73					
76					

12

ORIGINAL PAGE IS  
OF POOR QUALITY $\alpha$  OR  $\beta$   
SCHEDULES

$\beta(A) = -5, -4, -2, -1, 0, 1, 2, 4, 5$

(1) = Beta Schedules  
 (2) = Beta Schedules

1 = Beta Schedules  
 2 = Beta Schedules



TABLE III. MODEL DIMENSIONAL DATA

MODEL COMPONENT: BODY - B<sub>26</sub>

GENERAL DESCRIPTION: Configuration 140A/B Orbiter fuselage. NOTE: B<sub>26</sub> is identical to B<sub>24</sub> except underside of fuselage has been refaired to accept W<sub>116</sub>.

MODEL SCALE: 0.030 MODEL DRAWING: SS-A00147, Release 12

DRAWING NUMBER: VL70-000143B, -000200, -000205, -006089, -000145.  
VL70-000140A, -000140B

DIMENSIONS:	<u>FULL SCALE</u>	<u>MODEL SCALE</u>
Length (OML: Fwd. Sta. X <sub>0</sub> = 235), In.	1293.3	38.799
Length (IML: Fwd. Sta. X <sub>0</sub> = 238), In.	1290.3	38.709
Max Width (@ X <sub>0</sub> = 1528.3), In.	264.0	7.920
Max Depth (@ X <sub>0</sub> = 1564), In.	250.00	7.500
Fineness Ratio	0.264	0.264
Area - Ft <sup>2</sup>		
Max. Cross-Sectional	340.88	0.3068

TABLE III. MODEL DIMENSIONAL DATA (Continued)

MODEL COMPONENT: CANOPY - C<sub>9</sub>

GENERAL DESCRIPTION: Configuration 3A. Canopy used with fuselage  
B26.

MODEL SCALE: 0.030

MODEL DRAWING: SS-A00147, Release 12

DRAWING NUMBER:

DIMENSIONS:	<u>FULL SCALE</u>	<u>MODEL SCALE</u>
Length ( $x_0 = 434.643$ to 578), In.	143.357	4.301
Max Width (@ $x_0 = 513.127$ ), In.	152.412	4.572
Max Depth (@ $x_0 = 485.0$ ), In.	25.000	0.750

TABLE III. MODEL DIMENSIONAL DATA (Continued)

MODEL COMPONENT: SLOTTED ELEVON (6-inch gap) - E<sub>43</sub>GENERAL DESCRIPTION: Configuration 140A/B orbiter elevon. NOTE: E<sub>43</sub>  
is a slotted version of E<sub>26</sub>. Data are for one side.

MODEL SCALE: 0.030

DRAWING NUMBER: VL70-000200, -006089, -006092

DIMENSIONS:	<u>FULL SCALE</u>	<u>MODEL SCALE</u>
Area, Ft <sup>2</sup>	210.00	0.189
Span (equivalent), In.	349.2	10.476
Inb'd equivalent chord, In.	118.004	3.540
Outb'd equivalent chord, In.	55.192	1.656
Ratio movable surface chord/ total surface chord		
At Inb'd equiv. chord	0.2096	0.2096
At Outb'd equiv. chord	0.4004	0.4004
Sweep Back Angles, degrees		
Leading Edge	0.00	0.00
Trailing Edge	-10.056	-10.056
Hingeline	0.00	0.00
Area Moment (Product of Area & $\bar{c}$ ), Ft <sup>3</sup>	1587.25	0.043
Mean Aerodynamic Chord, In.	90.7	2.721

TABLE III. MODEL DIMENSIONAL DATA (Continued)

MODEL COMPONENT: BODY FLAP - F8

GENERAL DESCRIPTION: Configuration 140A/B orbiter body flap.

Hingeline located at  $X_0 = 1528.3$ ,  $Z_0 = 284.3$

MODEL SCALE: 0.030

MODEL DRAWING: SS-A00147, Release 12

DRAWING NUMBER: VL70-000140A, -000145

DIMENSIONS:	<u>FULL SCALE</u>	<u>MODEL SCALE</u>
Length ( $X_0 = 1520$ to $X_0 = 1613$ ), In.	93.00	2.79
Max Width, In.	262.00	7.86
Max Depth (@ $X = 1520$ ), In.	23.00	0.69
Fineness Ratio		
Area - $\text{Ft}^2$		
Planform	150.525	0.406
Base	41.847	0.113

TABLE III. MODEL DIMENSIONAL DATA (Continued)

MODEL COMPONENT: G<sub>15</sub>

GENERAL DESCRIPTION: Main and nose landing gear door and struts.

MODEL SCALE: 0.03

DRAWING NUMBER: VL70-000140/SS-A00149

DIMENSIONS:	<u>FULL SCALE</u>	<u>MODEL SCALE</u>
<b>NOSE GEAR:</b>		
No. Wheels	2	2
Wheel axis:		
Fuselage Station, In.	374.741	11.242
Waterline, In.	221.506	6.645
Pivot axis:		
Fuselage Station	375.506	11.265
Waterline	298.000	8.940
Strut diameter, In.	7.716	0.231
Wheel diameter, In.	32.000	0.960
Wheel Width, In.	8.790	0.264
Wheel Centerline-to-centerline width, In.	22.000	0.660
Side door:		
Height (follows body contour), In.	18.518	0.556
Leading Edge location, F.S.	272.049	8.161
Trailing Edge location, F.S.	381.926	11.458
<b>MAIN GEAR (Two Struts):</b>		
No. of wheels (per strut)	2	2
Wheel axis:		
Fuselage Station, In.	1171.086	35.133
Waterline, In.	175.432	5.263
Pivot axis:		
Fuselage Station, In.	1180.000	35.400
Waterline, In.	283.012	8.490
Main Strut Diameter, In.	9.568	0.287
Wheel Diameter, In.	44.198	1.326
Wheel Width, In.	16.049	0.481
Wheel centerline-to-centerline width, In.	36.000	1.080
Side door:		
Length, In.	148.148	4.444
Width, In.	61.728	1.852

TABLE III. MODEL DIMENSIONAL DATA (Continued)

MODEL COMPONENT: OMS POD - M<sub>16</sub>

GENERAL DESCRIPTION: Configuration 140C orbiter OMS pod - short pod.

MODEL SCALE: 0.030

DRAWING NUMBER: VL70-008401 -008410

DIMENSIONS:	<u>FULL SCALE</u>	<u>MODEL SCALE</u>
Length (OMS Fwd. Sta. X <sub>0</sub> = 1310.5), In.	258.50	7.755
Max Width (@X <sub>0</sub> = 1511), In.	136.8	4.104
Max Depth (@X <sub>0</sub> = 1511), In.	74.70	2.241
Fineness Ratio	2.484	2.484
Area - Ft <sup>2</sup>		
Max Cross-Sectional	58.864	0.053

TABLE III. MODEL DIMENSIONAL DATA (Continued)

MODEL COMPONENT: MPS NOZZLES - N<sub>24</sub>

GENERAL DESCRIPTION: Configuration 140A/B orbiter MPS nozzles

MODEL SCALE: 0.030

MODEL DRAWING: SS-A00147, Release 12

DRAWING NUMBER: VL70-005030A, -000140A

DIMENSIONS:

FULL SCALE      MODEL SCALE

MACH NO.

Length - In.

Gimbal Point to Exit Plane  
Throat to Exit Plane

157.0      4.71  
99.2      2.976

Diameter - In.

Exit

91.00      2.73

Area - Ft<sup>2</sup>

Exit

45.17      0.0407

Gimbal Point (Station) - In.

Upper Nozzle

X

1445.      43.35

Y

0      0

Z

443.00      13.29

Lower Nozzles

X

1468.17      44.045

Y

+53.00      +1.59

Z

342.64      10.279

Null Position - Deg.

Upper Nozzle

Pitch

16      16

Yaw

0      0

Lower Nozzle

Pitch

10      10

Yaw

3.5      3.5

TABLE III. MODEL DIMENSIONAL DATA (Continued)

MODEL COMPONENT: OMS NOZZLES - N<sub>28</sub>

GENERAL DESCRIPTION: Configuration 140A/B orbiter OMS nozzles.

MODEL SCALE: 0.030

DRAWING NUMBER: VL70-000140A (Location) SS-A00106, Release 5 (Contour)

DIMENSIONS:	<u>FULL SCALE</u>	<u>MODEL SCALE</u>
-------------	-------------------	--------------------

MACH NO.

Gimbal Point (Station), In.

Left Nozzles

X <sub>o</sub>	1518.0	45.54
Y <sub>o</sub>	-88.0	-2.65
Z <sub>o</sub>	492.0	14.76

Right Nozzles

X <sub>o</sub>	1518.0	45.54
Y <sub>o</sub>	88.0	2.64
Z <sub>o</sub>	492.0	14.76

Null Position - Deg.

Left Nozzles

Pitch	15°49'	15°49'
Yaw	12°17'	12°17'

Right Nozzles

Pitch	15°49'	15°49'
Yaw	12°17'	12°17'

TABLE III. MODEL DIMENSIONAL DATA (Continued)

MODEL COMPONENT: RUDDER - R<sub>5</sub>

GENERAL DESCRIPTION: Configuration 140C orbiter rudder (identical to Configuration 140A/B rudder)

MODEL SCALE: 0.030

DRAWING NUMBER: VL70-000146B, -000095

DIMENSIONS:	<u>FULL SCALE</u>	<u>MODEL SCALE</u>
Area, Ft <sup>2</sup>	100.15	0.090
Span (equivalent), In.	201.00	6.030
Inb'd equivalent chord, In.	91.585	2.748
Outb'd equivalent chord, In.	50.833	1.525
Ratio movable surface chord/ total surface chord		
At Inb'd equiv. chord	0.400	0.400
At Outb'd equiv. chord	0.400	0.400
Sweep Back Angles, degrees		
Leading Edge	34.83	34.83
Trailing Edge	26.25	26.25
Hingeline	34.83	34.83
Area Moment (Product of area & $\bar{c}$ ), Ft <sup>3</sup>	610.92	0.0165
Mean Aerodynamic Chord, In.	73.2	2.196

TABLE IIIa. - MODEL DIMENSIONAL DATA - Continued.

MODEL COMPONENT: Tailcone - TC19

GENERAL DESCRIPTION: Afterbody tailcone used with body B26 to simulate the orbiter ferry configuration.

SCALE MODEL = .03

DRAWING NUMBER: SS-A01681  
VL70-30-330-02

DIMENSIONS: FULL SCALE      MODEL SCALE

Length (TE @ $X_O$ 1900), in.	<u>459.3</u>	<u>13.779</u>
-------------------------------	--------------	---------------

Max. Width ( $X_O$ 1523), in.	<u>303</u>	<u>9.090</u>
-------------------------------	------------	--------------

Max. Depth ( $X_O$ 1466), in.	<u>286</u>	<u>8.380</u>
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Area, $\text{ft}^2$		
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Max. Cross-Sectional ( $X_O$ 1523)	<u>504.6</u>	<u>.0454</u>
------------------------------------	--------------	--------------

Frontal	<u>469.9</u>	<u>0.423</u>
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TABLE III. MODEL DIMENSIONAL DATA

MODEL COMPONENT: VERTICAL - V<sub>8</sub>GENERAL DESCRIPTION: Configuration 140C orbiter vertical tail  
(Identical to configuration 140A/B vertical tail).

MODEL SCALE: 0.030

DRAWING NUMBER: VL70-000140C, -000146B

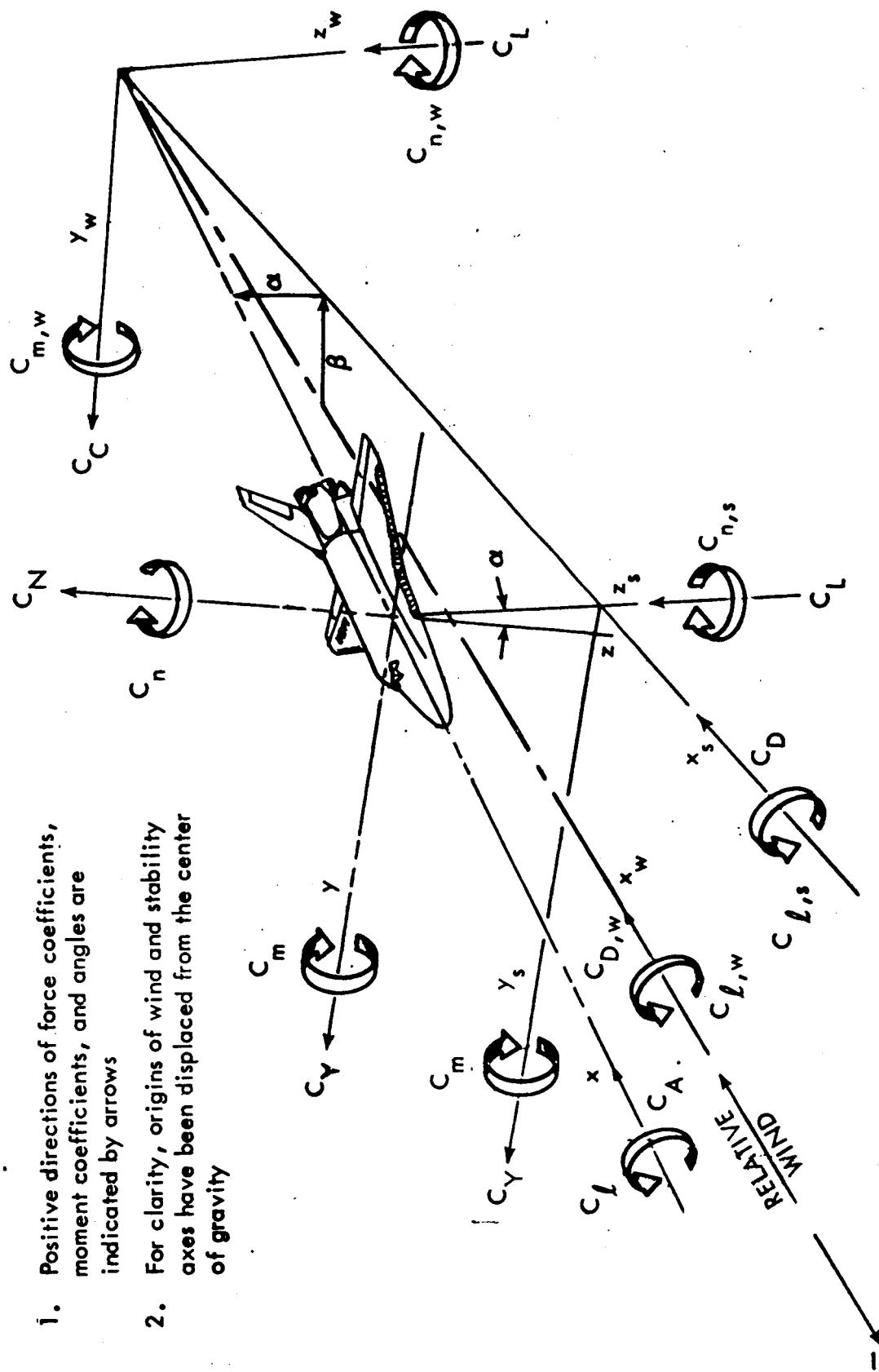
DIMENSIONS:	<u>FULL SCALE</u>	<u>MODEL SCALE</u>
<b>TOTAL DATA</b>		
Area (Theo.), ft <sup>2</sup> Planform	413.253	0.372
Span (Theo.), in.	315.72	9.472
Aspect Ratio	1.675	1.675
Rate of Taper	0.507	0.507
Taper Ratio	0.404	0.404
Sweep Back Angles, degrees		
Leading Edge	45.000	45.000
Trailing Edge	26.25	26.25
0.25 Element Line	41.13	41.13
Chords:		
Root (Theo.) WP	268.50	8.055
Tip (Theo.) WP	108.47	3.254
MAC	199.81	5.994
Fus. Sta. of .25 MAC	1463.35	43.901
W.P. of .25 MAC	635.52	19.066
B.L. of .25 MAC	0.0	0.0
Airfoil Section		
Leading Wedge Angle, deg.	10.00	10.00
Trailing Wedge Angle, deg.	14.92	14.92
Leading Edge Radius	3.00	0.060
Void Area	13.17	0.0019
Blanketed Area		

TABLE III. MODEL DIMENSIONAL DATA (Concluded)  
 MODEL COMPONENT: WING-W<sub>116</sub>  
 GENERAL DESCRIPTION: Configuration 4. NOTE: Identical to W<sub>114</sub> except airfoil thickness. Dihedral angle is along trailing edge of wing.

MODEL SCALE: 0.030	DRAWING NO.: VL70-000140A, -000200	FULL SCALE	MODEL SCALE
<b>DIMENSIONS:</b>			
<b>TOTAL DATA</b>			
Area (Theo.), ft <sup>2</sup>			
Planform	2690.00	2.421	
Span (Theo.), in.	936.68	28.10	
Aspect Ratio	2.265	2.265	
Rate of Taper	1.177	1.177	
Taper Ratio	0.200	0.200	
Dihedral Angle, degrees	3.500	3.500	
Incidence Angle, degrees	0.500	0.500	
Aerodynamic Twist, degrees	3.000	3.000	
Sweep Back Angles, degrees			
Leading Edge	45.000	45.000	
Trailing Edge	- 10.056	- 10.056	
0.25 Element Line	35.209	35.209	
Chords:			
Root (Theo.) B.P.O.O.	689.24	20.677	
Tip, (Theo.) B.P.	137.85	4.136	
MAC	474.81	14.244	
Fus. Sta. of .25 MAC	1136.83	34.105	
W.P. of .25 MAC	290.58	8.717	
B.L. of .25 MAC	182.13	5.464	
<b>EXPOSED DATA</b>			
Area (Theo.), ft <sup>2</sup>	1751.50	1.576	
Span, (Theo.), in. BP108	720.68	21.620	
Aspect Ratio	2.059	2.059	
Taper Ratio	0.245	0.245	
Chords			
Root BP108	562.09	16.863	
Tip 1.00 b/2	137.85	4.136	
MAC	392.83	11.785	
Fus. Sta. of .25 MAC	1185.98	35.579	
W.P. of .25 MAC	294.30	8.829	
B.L. of .25 MAC	251.77	7.553	
Airfoil Section (Rockwell Mod NASA) XXXX-64			
Root b/2 =	0.113	0.113	
Tip b/2 =	0.120	0.120	
Data for (1) or (2) Sides			
Leading Edge Cuff			
Planform Area, ft <sup>2</sup>	113.18	0.102	
Leading Edge Intersects Fus M.L. @ Sta	500.00	15.00	
Leading Edge Intersects Wing @ Sta	1024.00	30.72	

**Notes:**

1. Positive directions of force coefficients, moment coefficients, and angles are indicated by arrows
2. For clarity, origins of wind and stability axes have been displaced from the center of gravity



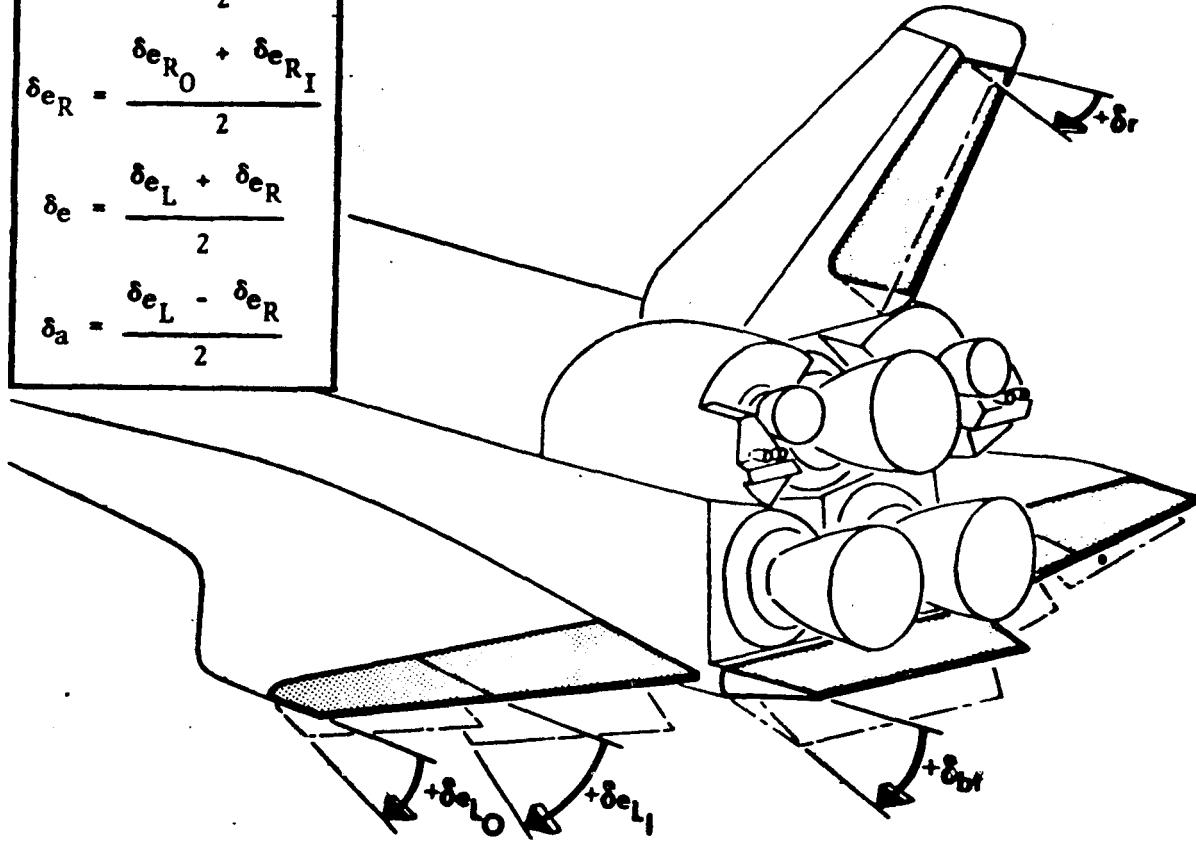
a. Orbiter Axis Systems  
Figure 1. Axis systems and sign conventions.

$$\delta e_L = \frac{\delta e_{L_0} + \delta e_{L_I}}{2}$$

$$\delta e_R = \frac{\delta e_{R_0} + \delta e_{R_I}}{2}$$

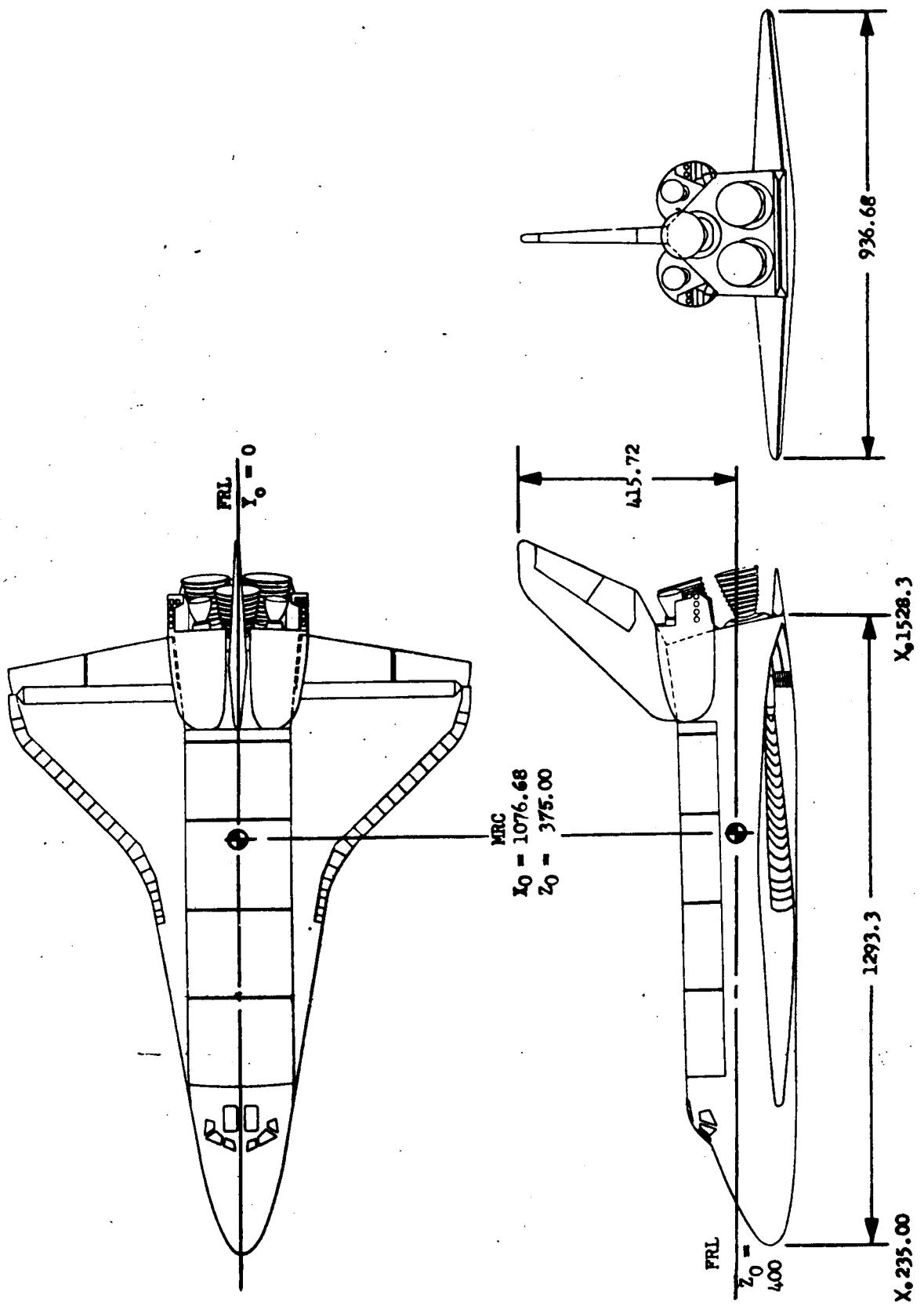
$$\delta e = \frac{\delta e_L + \delta e_R}{2}$$

$$\delta_a = \frac{\delta e_L - \delta e_R}{2}$$

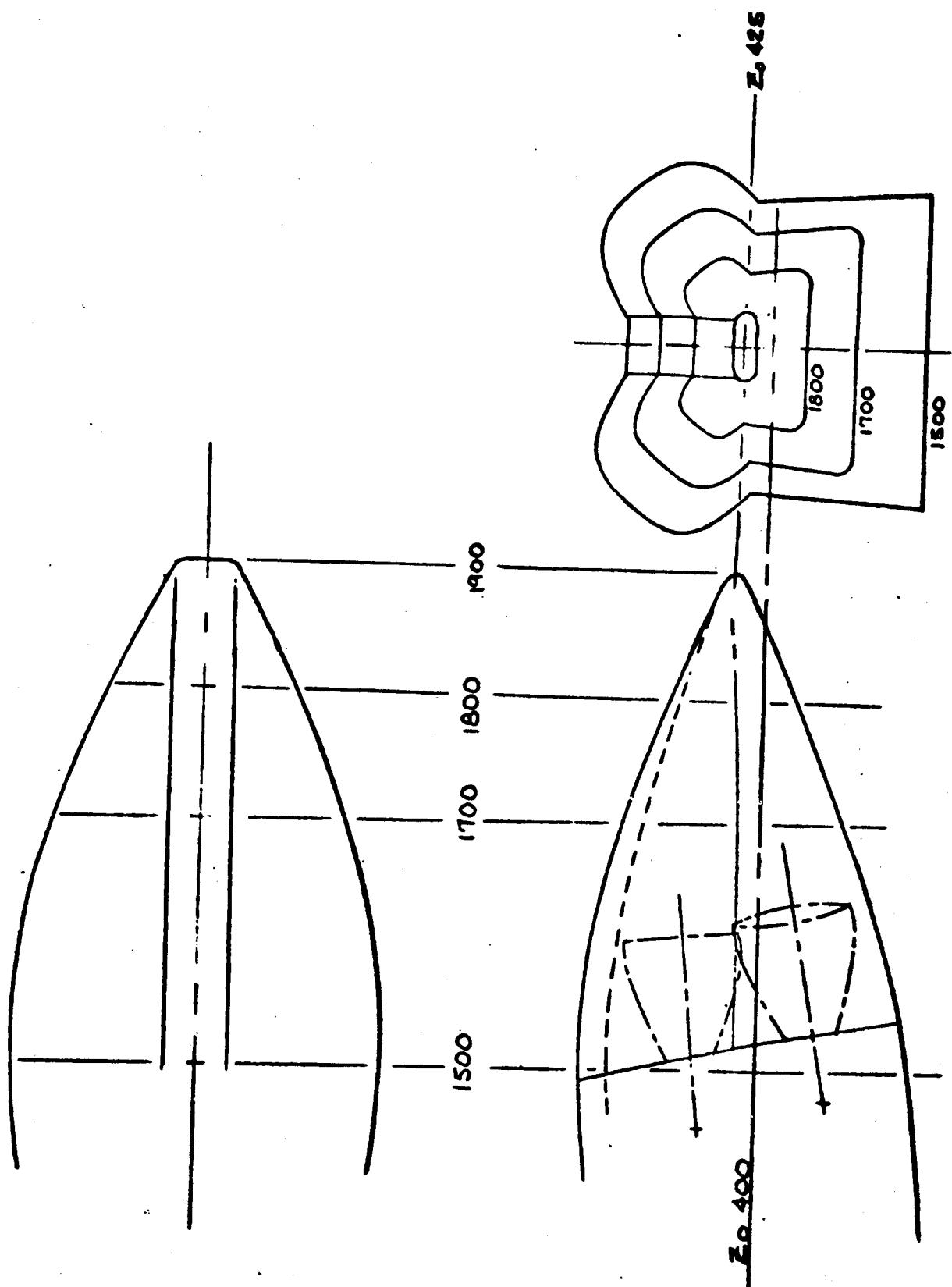


Positive Deflection of	Angle	Aero Forces and Moments	Hinge Moment
Rudder, $\delta_r$	$+β$ , $-ψ$	$+C_Y$ , $-C_n$	$-C_{h_r}$
Elevon, $\delta_e$	$-α$ , $-θ$	$-C_m$	$-C_{h_e}$
Right, $\delta e_R$	$-φ$	$-C_l$	$-C_{h_{eR}}$
Left, $\delta e_L$	$+φ$	$+C_l$	$-C_{h_{eL}}$
Aileron, $\delta_a$	$+φ$	$+C_l$	
Body Flap, $\delta_{bf}$	$-α$ , $-θ$	$-C_m$	$-C_{hb_f}$

b. Control Surface Deflections  
Figure 1. Concluded.



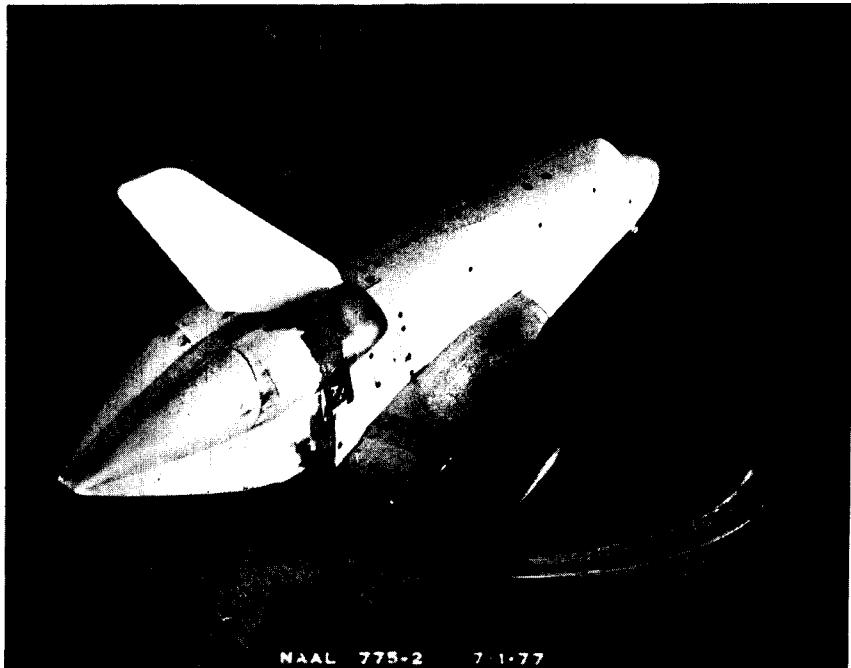
a. Three-View of LEOA/B Modified Orbiter Configuration  
 Figure 2. Model sketches.



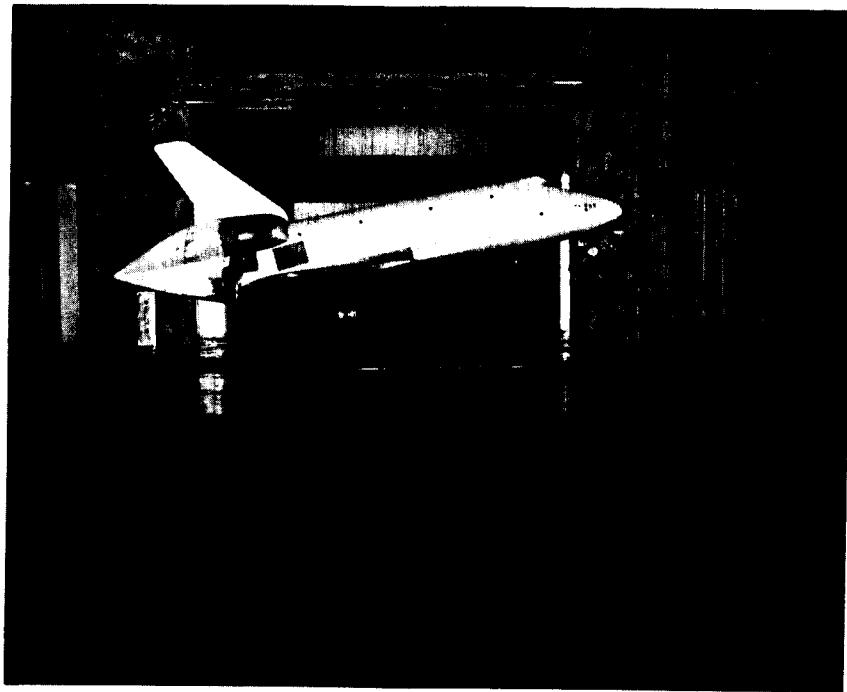
b. Tail Cone Configuration - TC19  
Figure 2. Concluded.



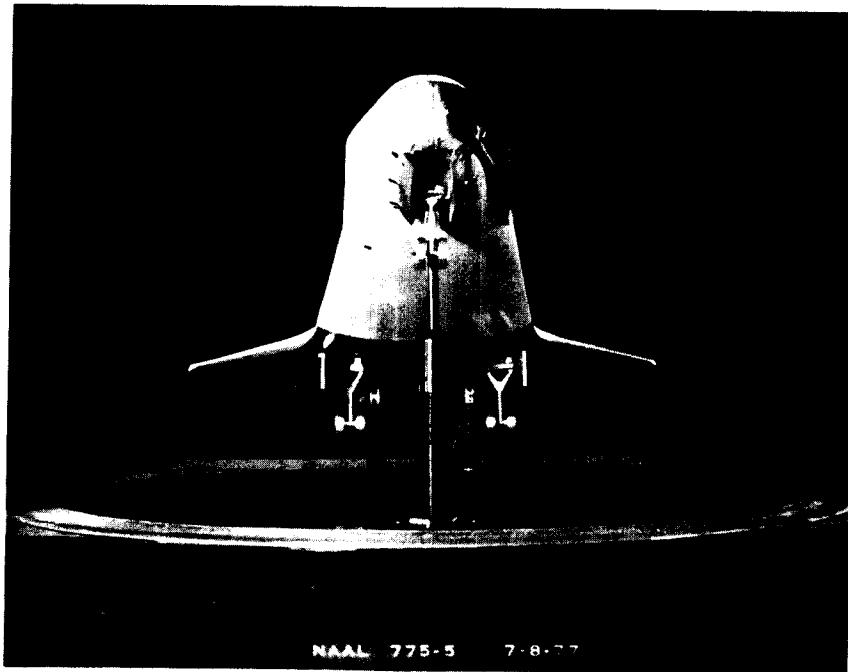
a. NAAL Installation, Three-Quarter Front View  
Configuration B<sub>26</sub> C<sub>9</sub> G<sub>15</sub> M<sub>16</sub> F<sub>8</sub> W<sub>116</sub> E<sub>43</sub> V<sub>8</sub> R<sub>5</sub> TC<sub>19</sub>



b. NAAL Installation, Rear View Configuration  
B<sub>26</sub> C<sub>9</sub> G<sub>15</sub> M<sub>16</sub> F<sub>8</sub> W<sub>116</sub> E<sub>43</sub> V<sub>8</sub> R<sub>5</sub> TC<sub>19</sub>  
Figure 3. Model photographs.



c. NAAL Installation, Side View Configuration  
B<sub>26</sub> C<sub>9</sub> G<sub>15</sub> M<sub>16</sub> F<sub>8</sub> W<sub>116</sub> E<sub>43</sub> V<sub>8</sub> R<sub>5</sub> TC<sub>19</sub>



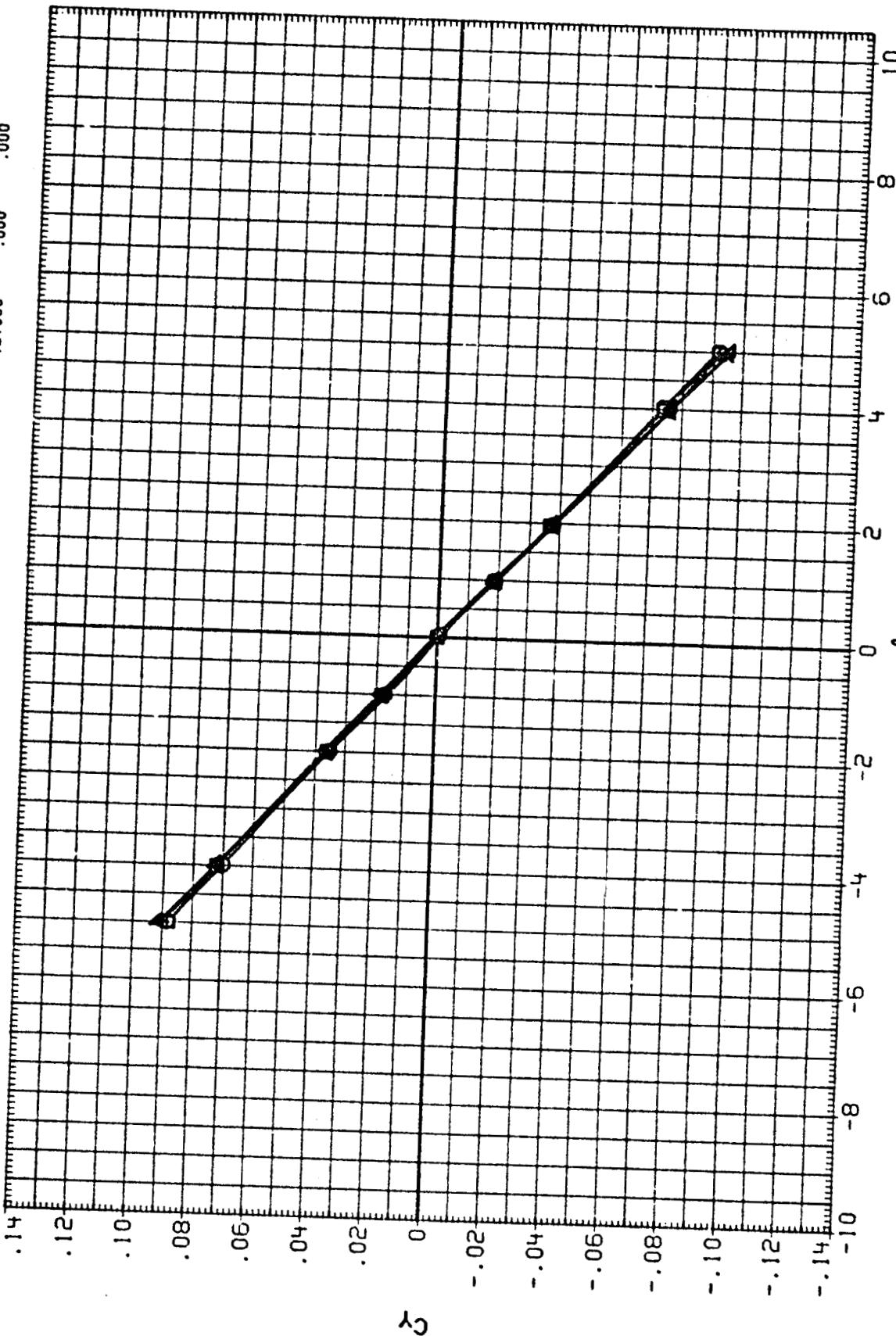
d. NAAL Installation, Front View Configuration  
B<sub>26</sub> C<sub>9</sub> G<sub>15</sub> M<sub>16</sub> F<sub>8</sub> W<sub>116</sub> E<sub>43</sub> V<sub>8</sub> R<sub>5</sub> TC<sub>19</sub>  
Figure 3. Concluded.

**DATA FIGURES**

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RF0007	O A250 B26C9G15M16F BH116E43VBR5TC19
RF0014	O A250 B26C9G15M16F BH116E43VBR5TC19
RF0016	O A250 B26C9G15M16F BH116E43VBR5TC19
RF0021	O A250 B26C9G15M16F BH116E43VBR5TC19

ELEVON	H/8	AIRCON	ALPHA	B0FLAP	RUDDER	SPOILER
.000	.200	.000	10.000	-12.000	.000	.000
.000	.178	.000	10.000	-12.000	.000	.000
.000	.156	.000	10.000	-12.000	.000	.000
.000	.100	.000	10.000	-12.000	.000	.000
.000	.075	.000	10.000	-12.000	.000	.000

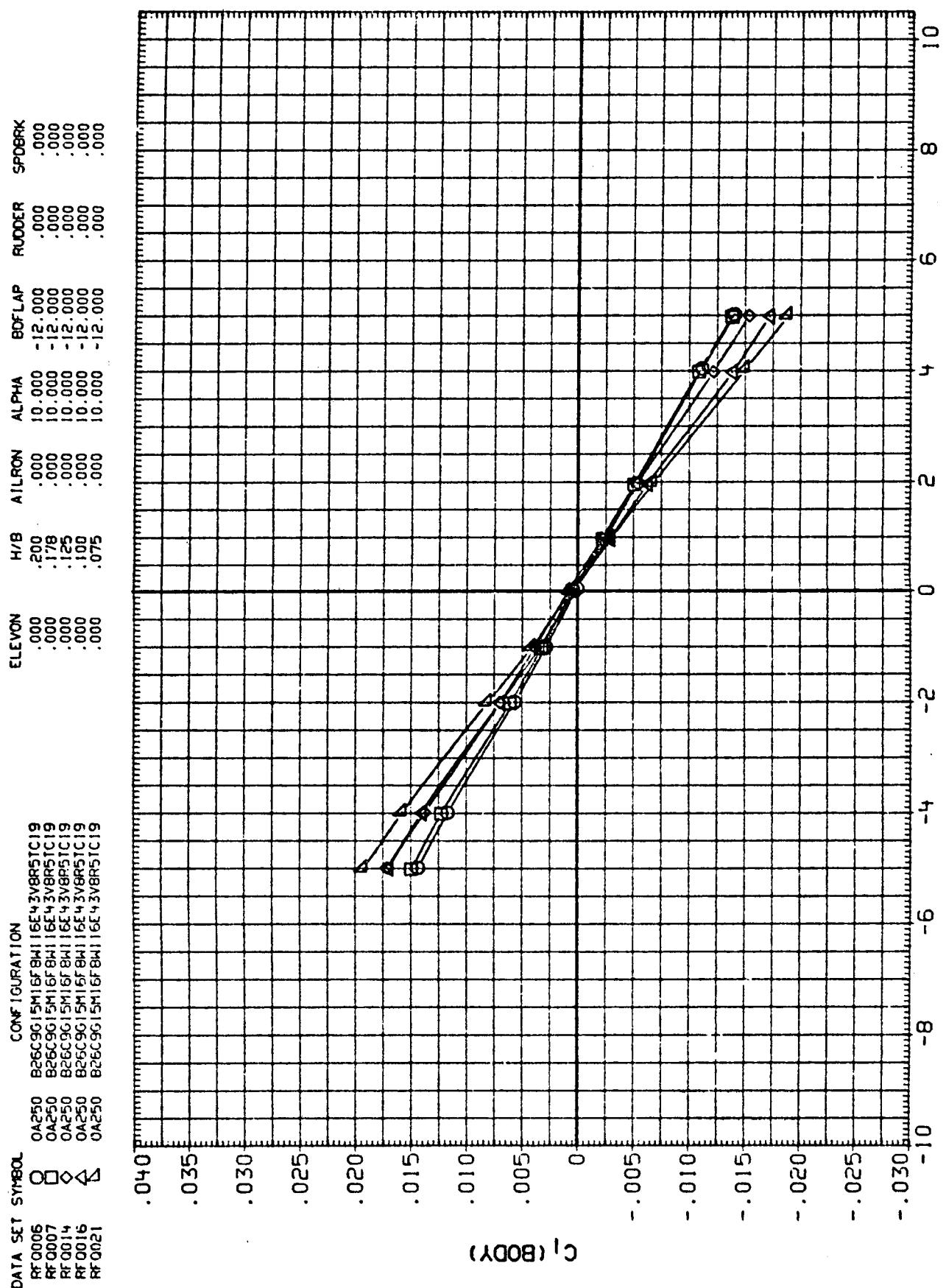


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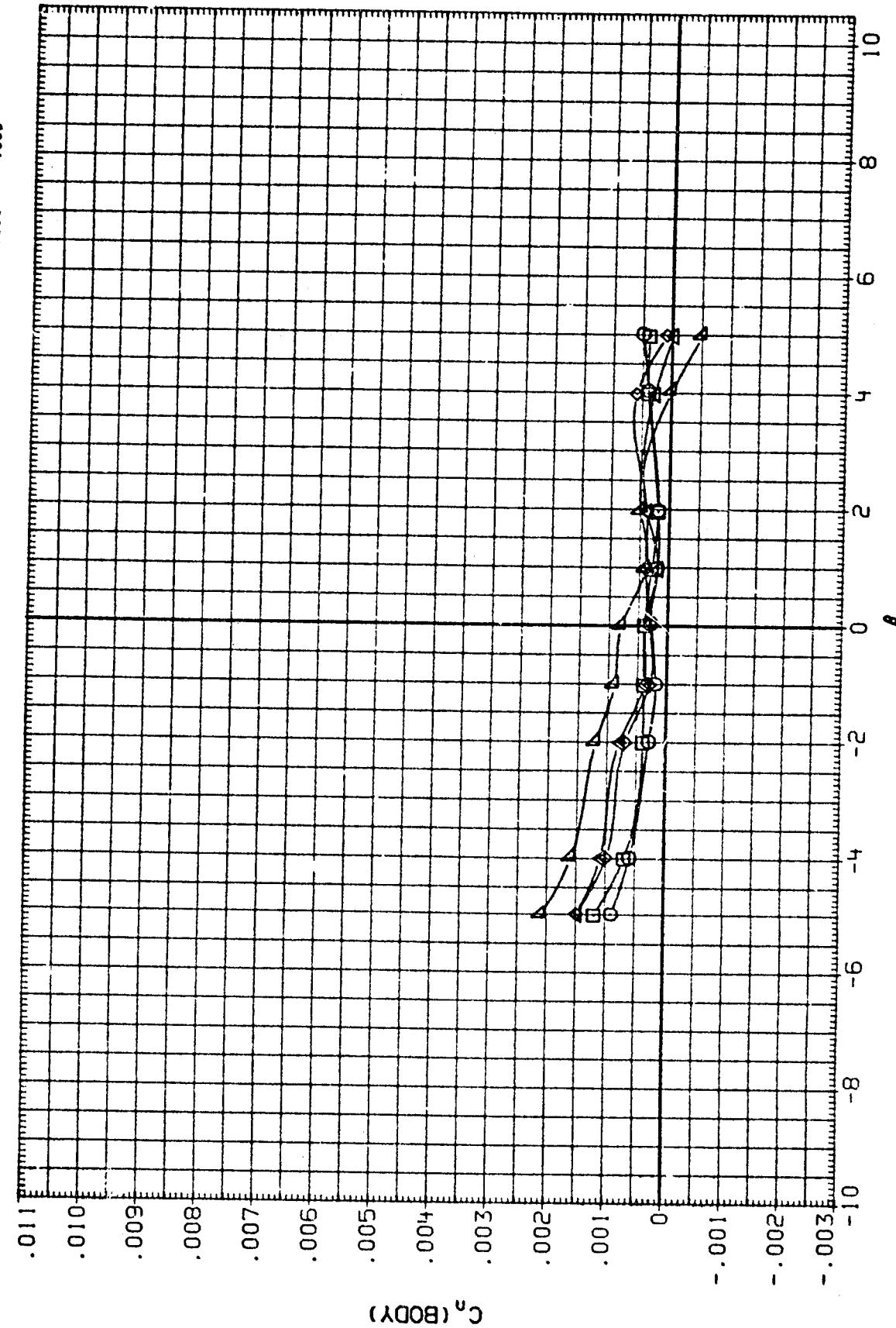


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RFQ0014	◇	B26C9615M16F8W116E43V8R5IC19
RFQ0016	△	B26C9615M16F8W116E43V8R5IC19
RFQ0021	▽	B26C9615M16F8W116E43V8R5IC19

ELEVON H/B AIRRON ALPHA BOFLAP RUDDER SPDBRK

.000	.200	.000	.000	-12.000	.000
.000	.178	.000	.000	-12.000	.000
.000	.125	.000	.000	-12.000	.000
.000	.100	.000	.000	-12.000	.000
.000	.075	.000	.000	-12.000	.000



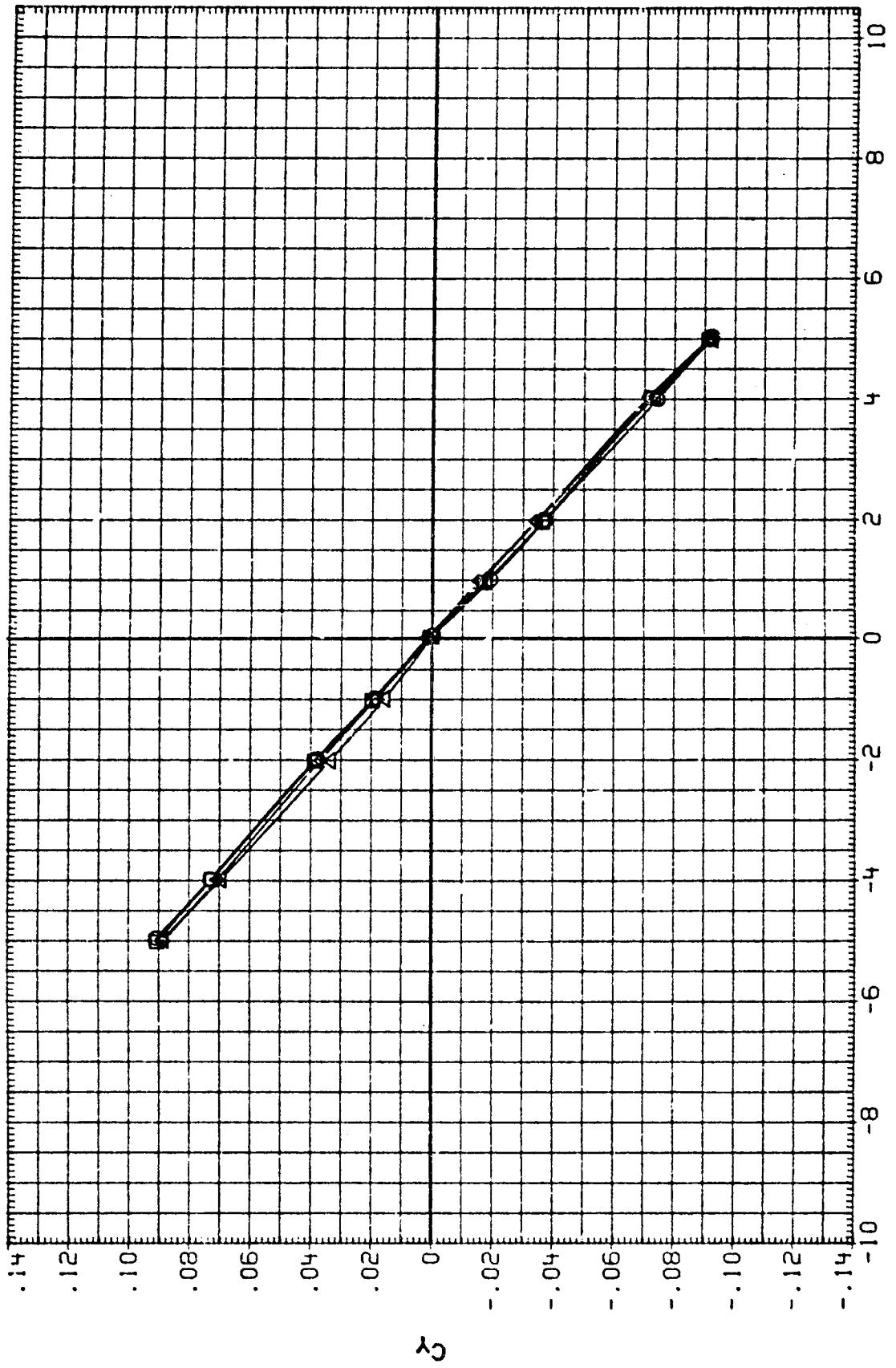
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RF0017	△ 0A250 B26C9G15M16FBH116E43VBR5TC19
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	ELEVON	H/B	AILRDN	ALPHA	BOFLAP	RUDDER	SPDBRK
RF0003	5.000	.200	.000	10.000	-12.000	.000	.000
RF0008	5.000	.178	.000	10.000	-12.000	.000	.000
RF0017	5.000	.100	.000	10.000	-12.000	.000	.000
RF0022	5.000	.075	.000	10.000	-12.000	.000	.000

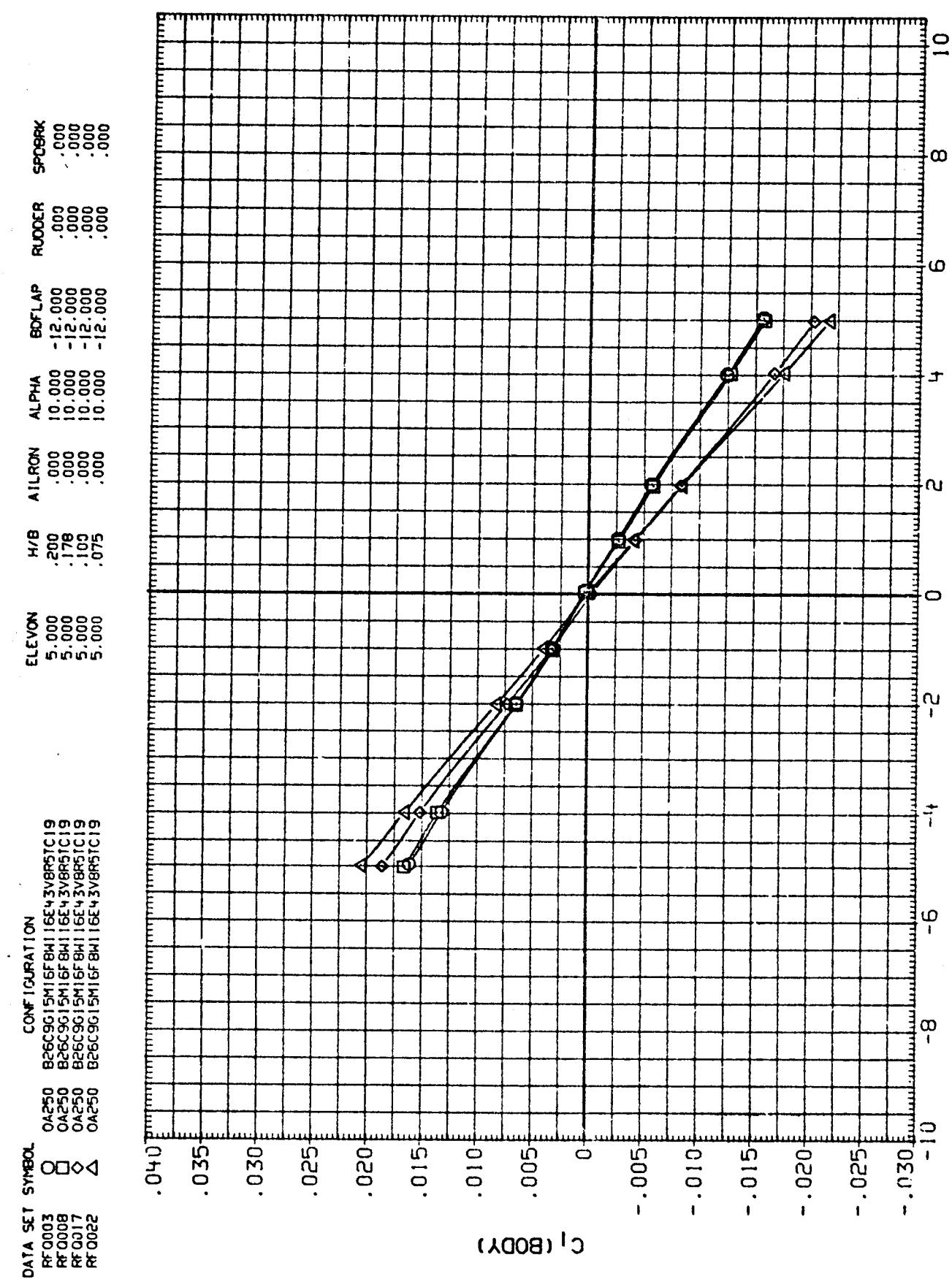


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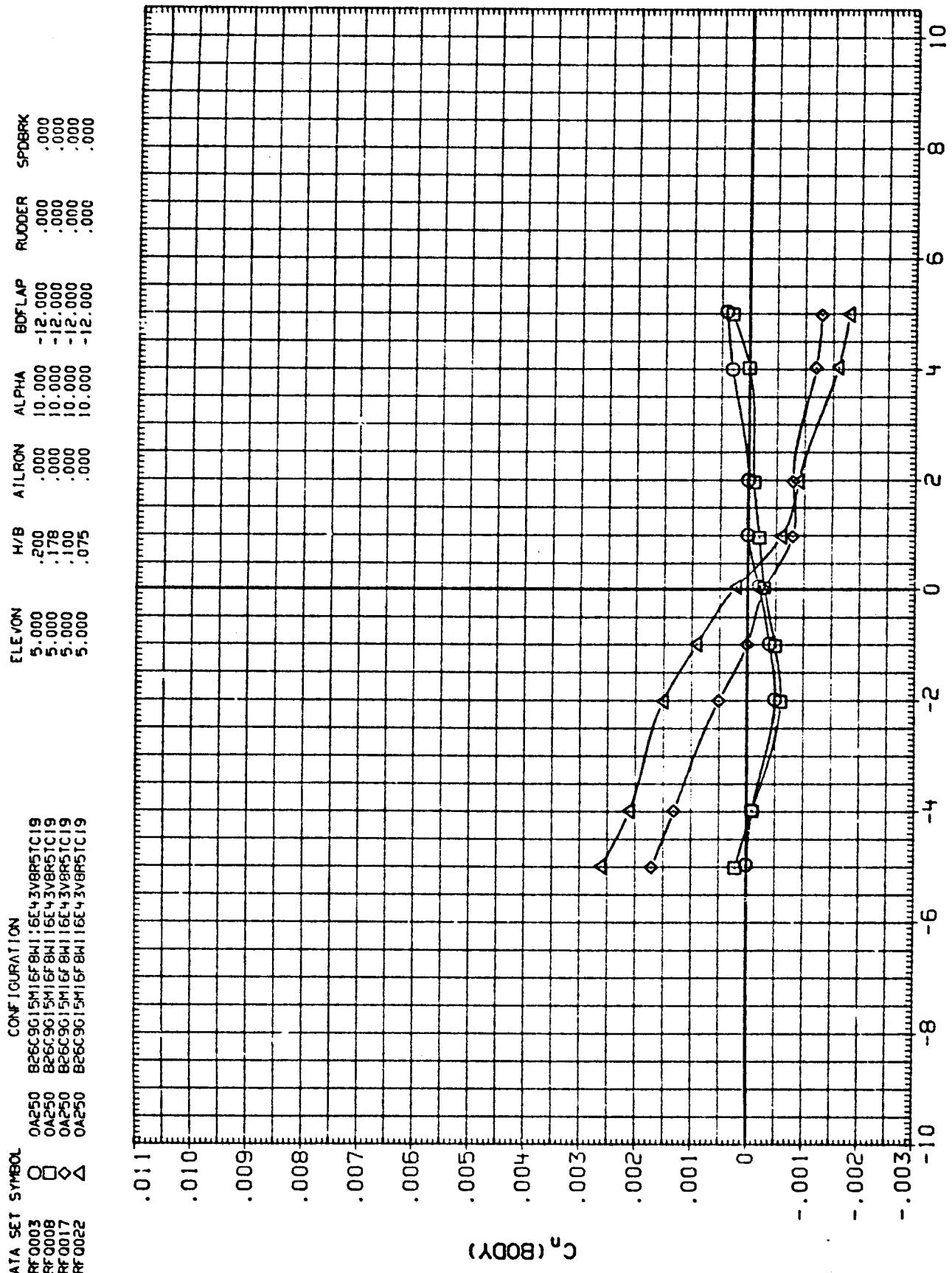
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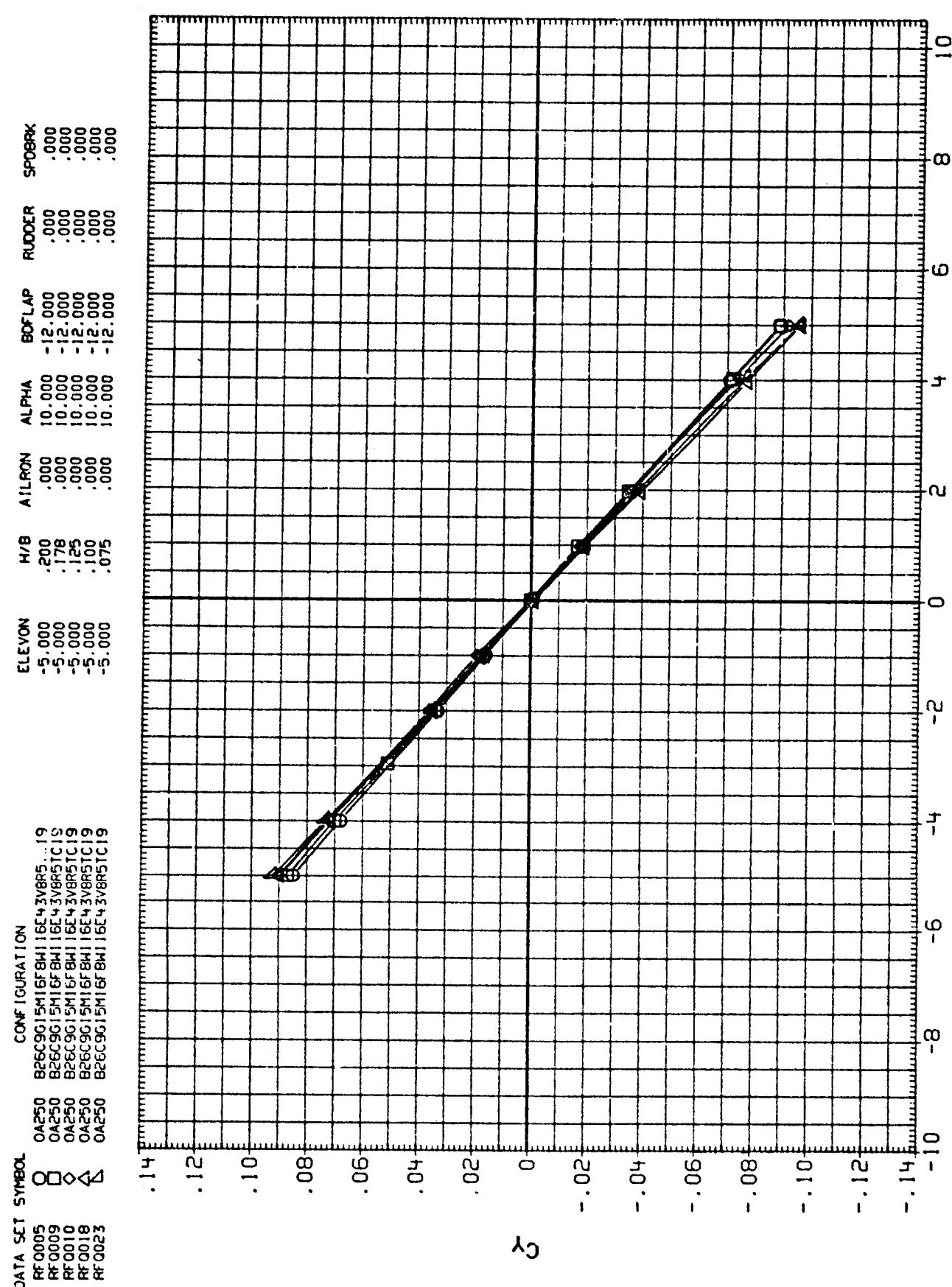
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(A) MACH = .20

GROUND PROXIMITY EFFECT, ELEVON DEFLECTION = 5.0  
 (A) MACH = .20



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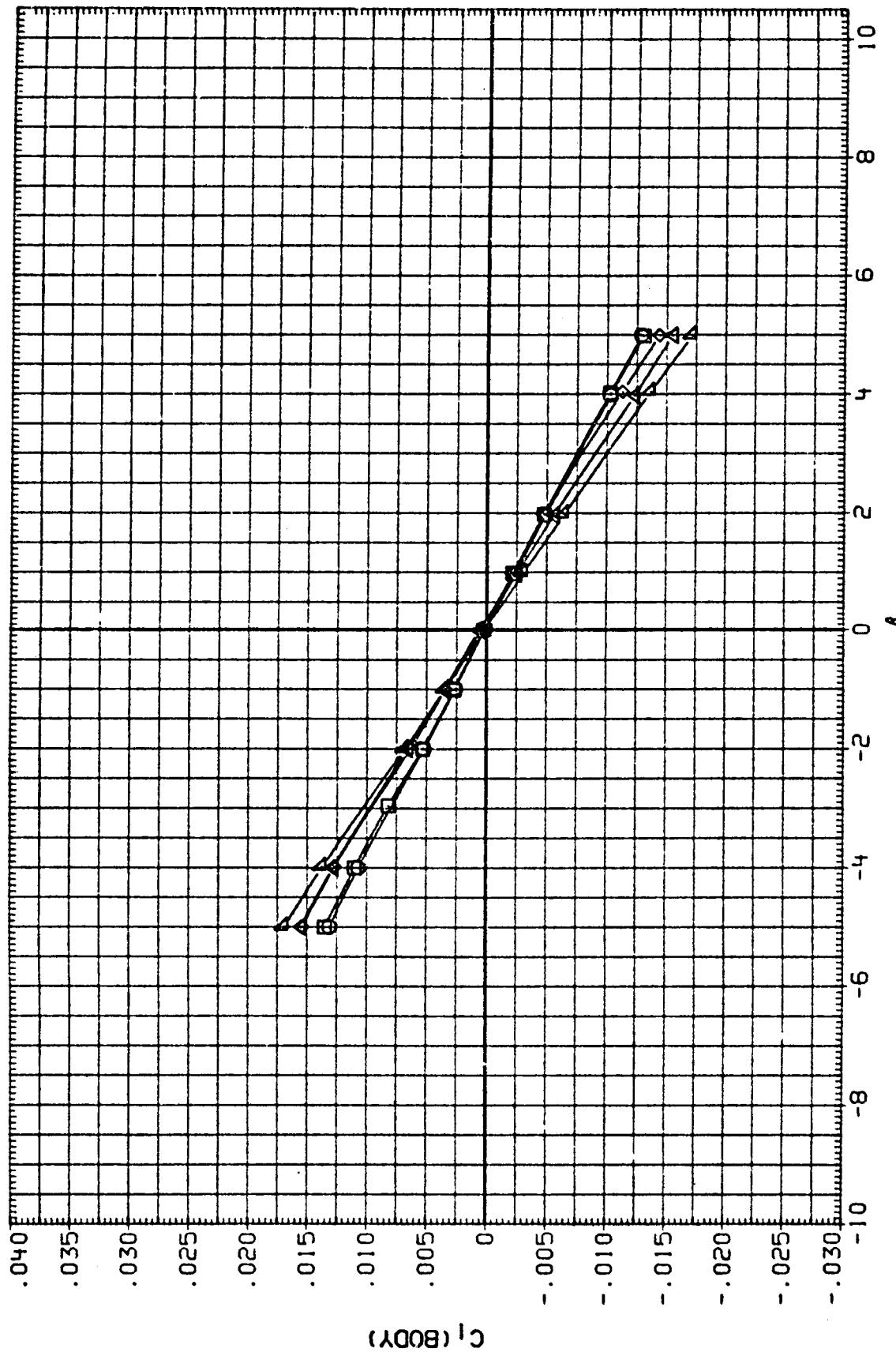


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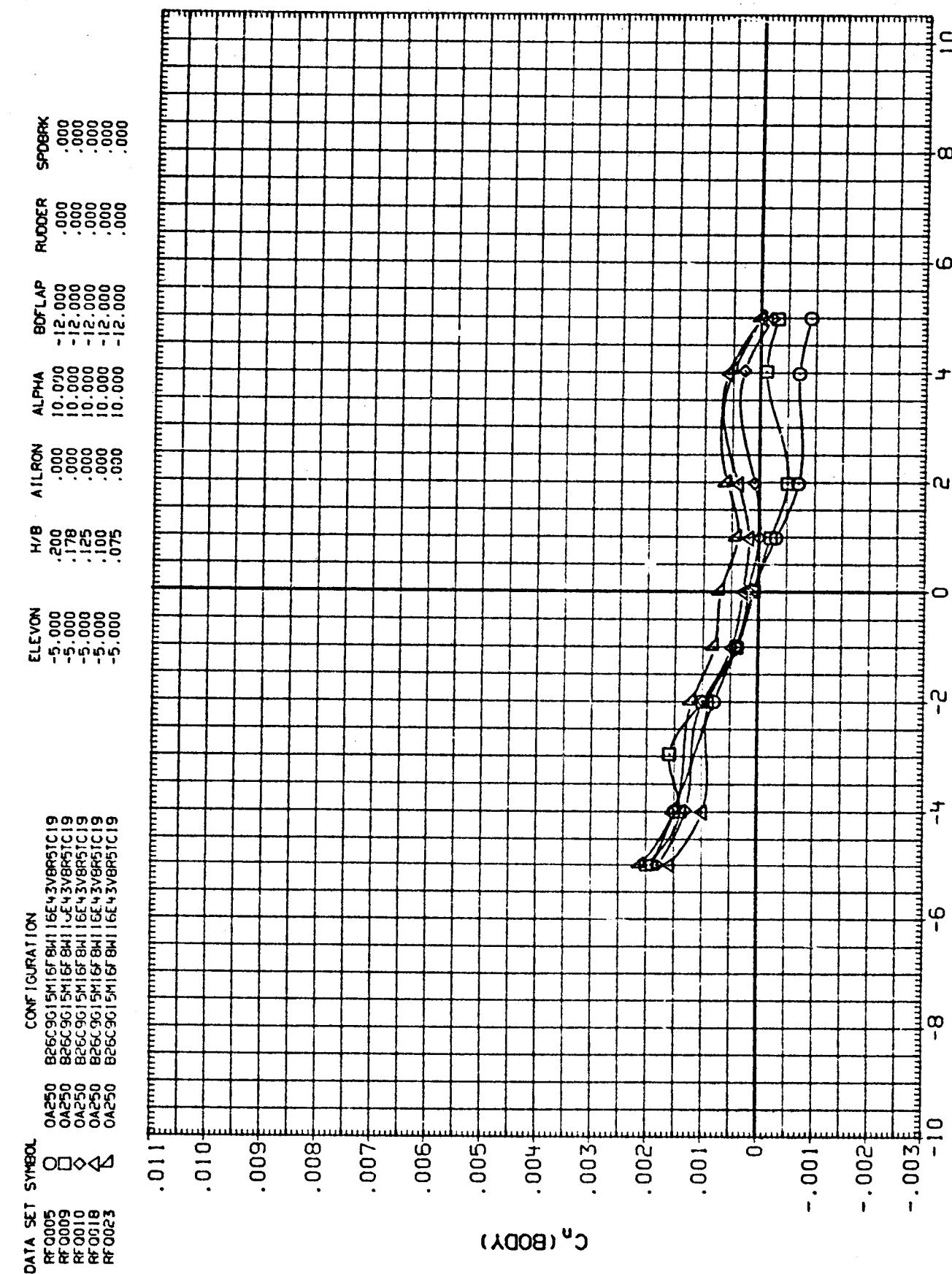
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RF Q009	□ A250 B26C9615M16F8W116E43V8R5TC19
RF Q010	◇ A250 B26C9615M16F8W116E43V8R5TC19
RF Q018	△ A250 B26C9615M16F8W116E43V8R5TC19
RF Q023	▲ A250 B26C9615M16F8W116E43V8R5TC19

	ELEVON	H/B	AIRRON	ALPHA	BLFLAP	RUDER	SPDBRK
RF Q005	-5.000	.200	.000	10.000	-12.000	.000	.000
RF Q009	-5.000	.178	.000	10.000	-12.000	.000	.000
RF Q010	-5.000	.125	.000	10.000	-12.000	.000	.000
RF Q018	-5.000	.100	.000	10.000	-12.000	.000	.000
RF Q023	-5.000	.075	.000	10.000	-12.000	.000	.000



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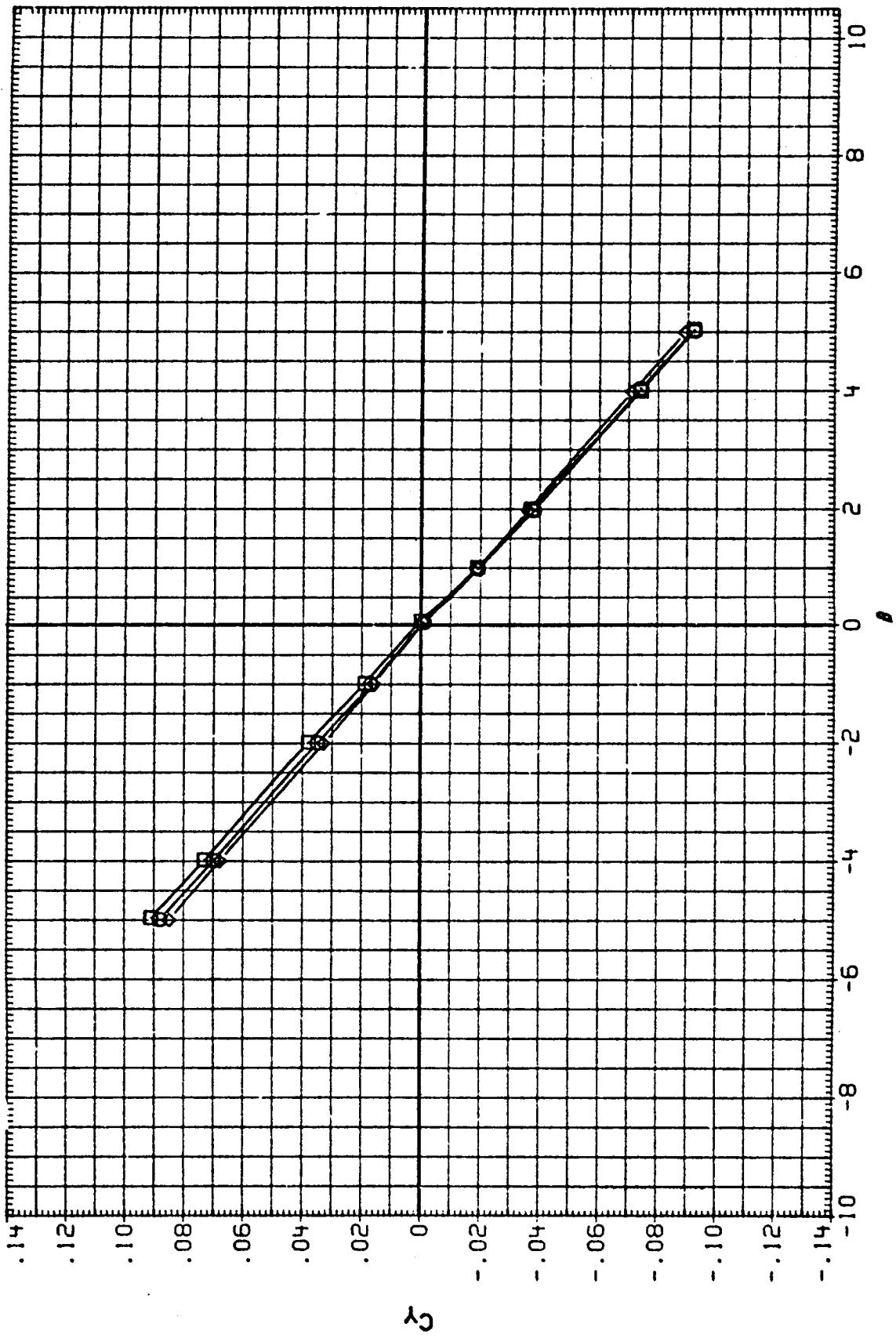
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GROUND PROXIMITY EFFECT, ELEVON DEFLECTION = -5.0

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RFQ003	O	0A250	B26C9615M16F8W116E43V8R5TC19	.200	.000	10.000	-12.000	.000	.000
RFQ005	D	0A250	B26C9615M16F8W116E43V8R5TC19	.200	.000	10.000	-12.000	.000	.000

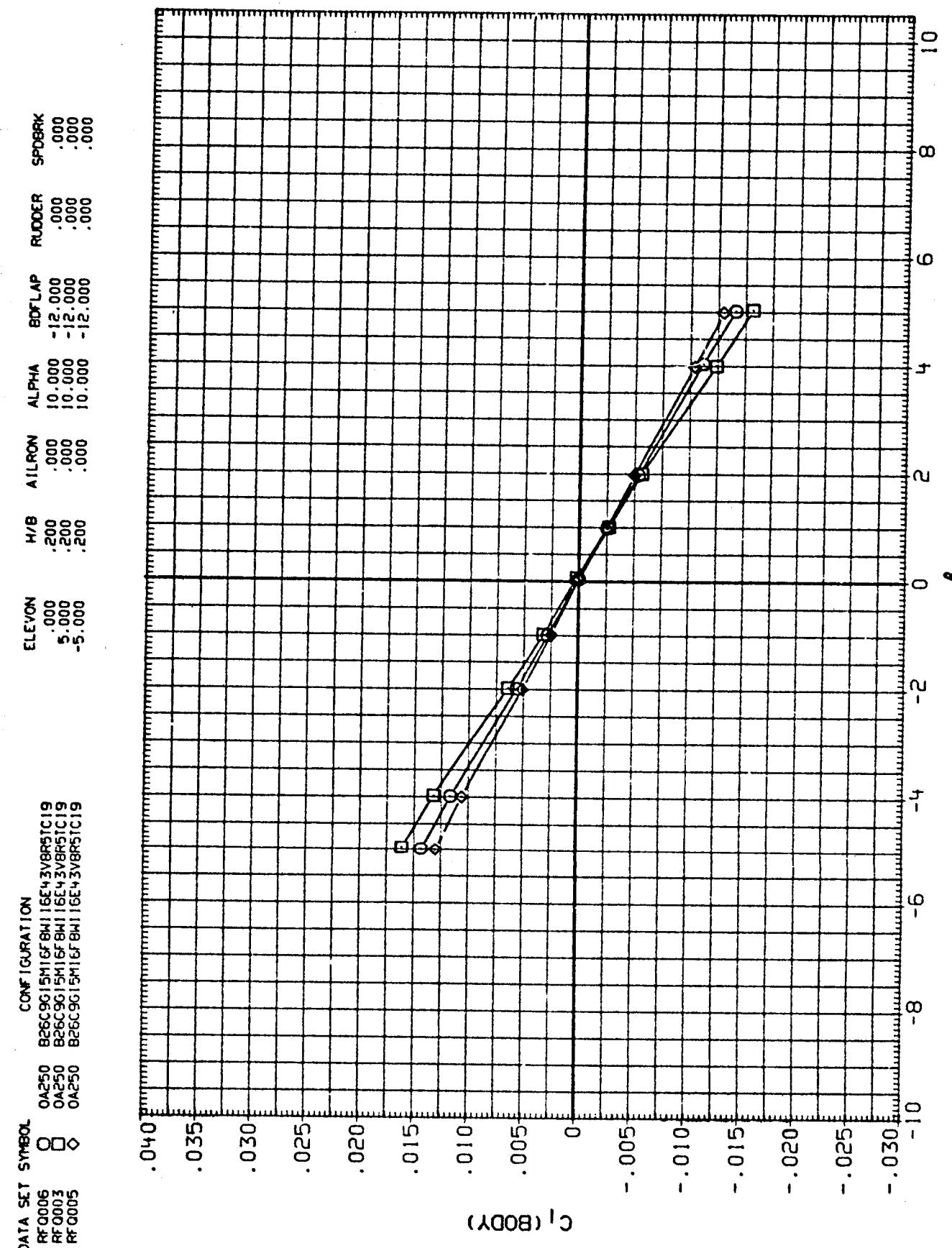


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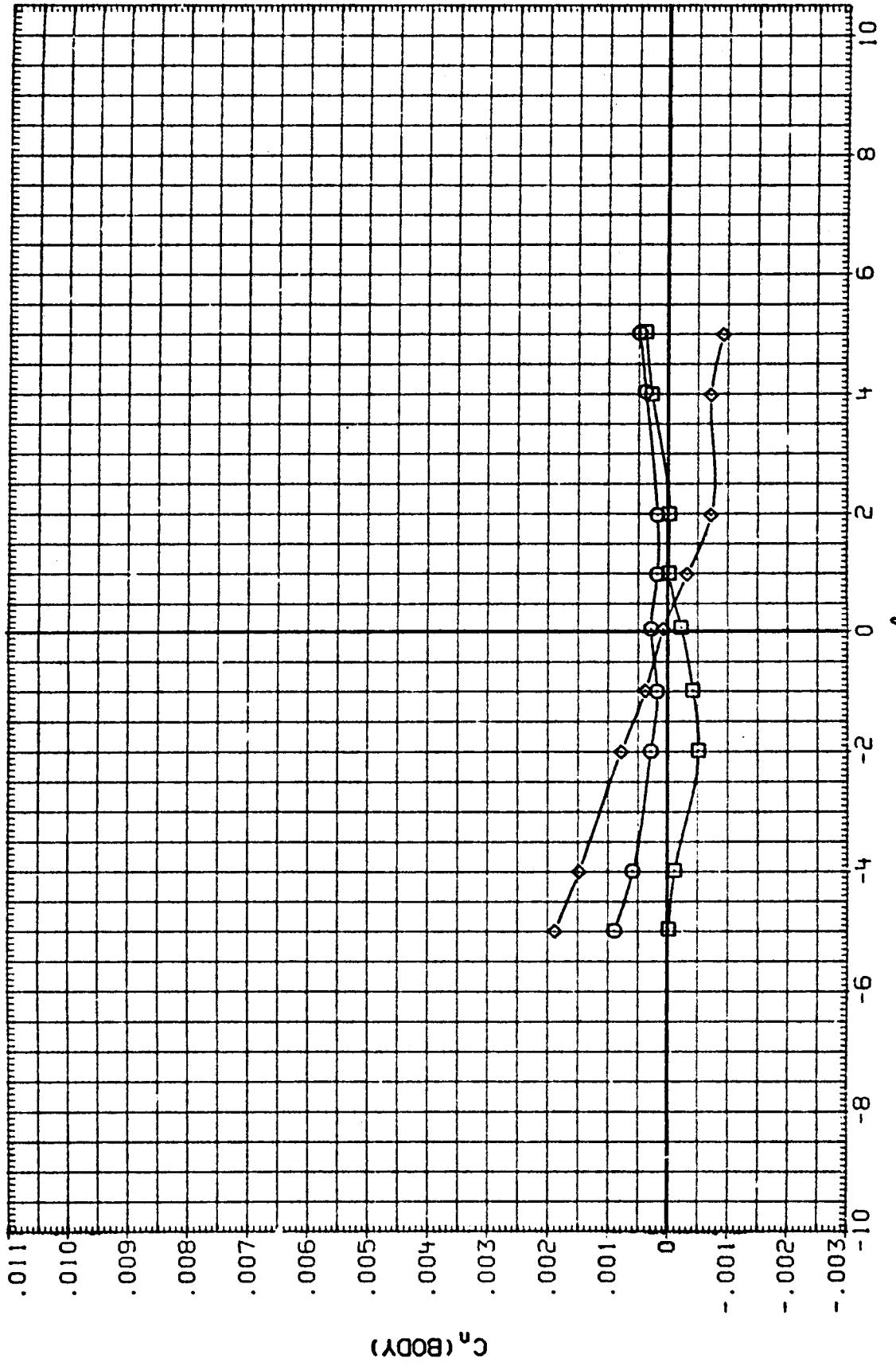
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PAGE 10



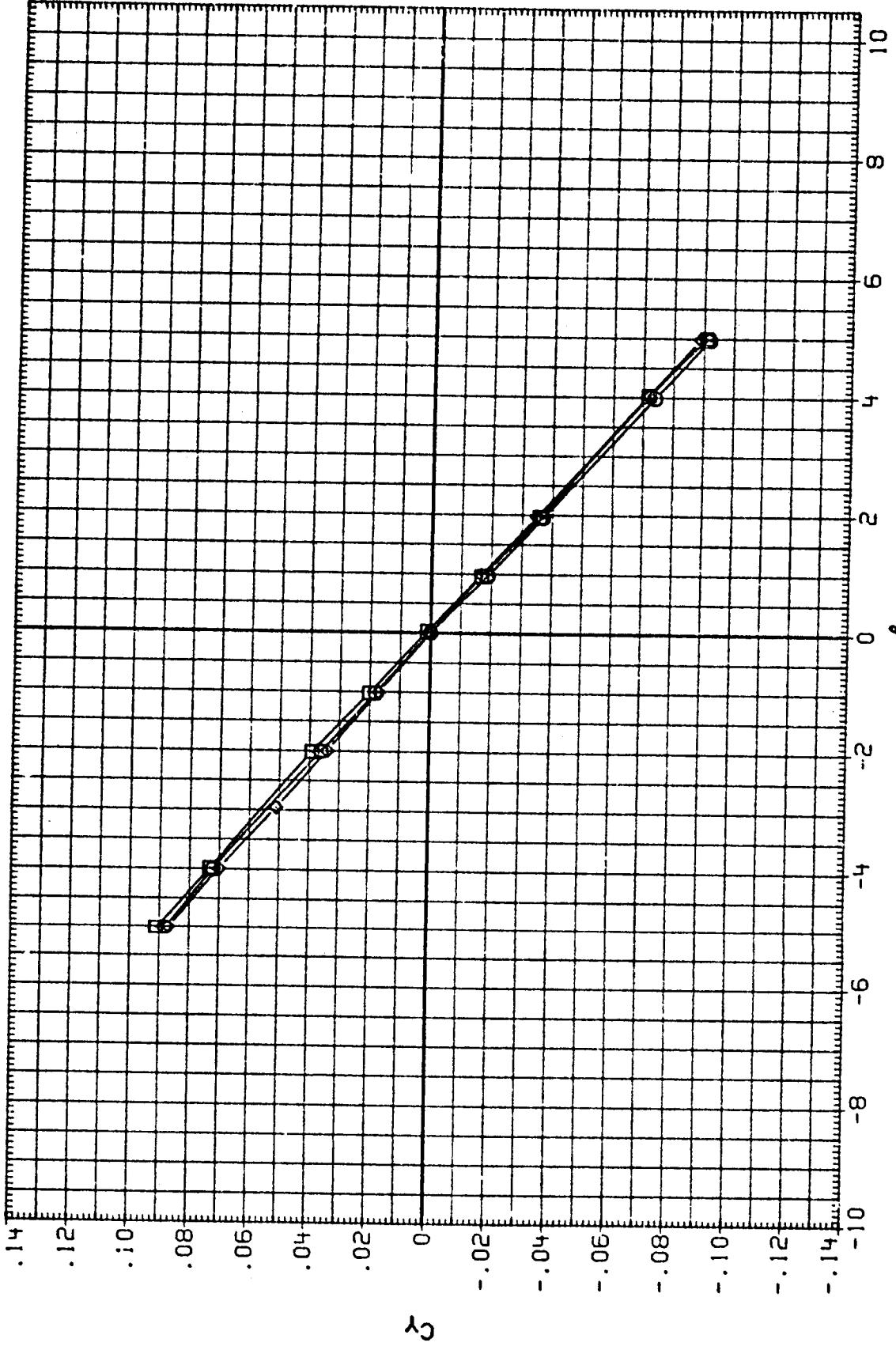
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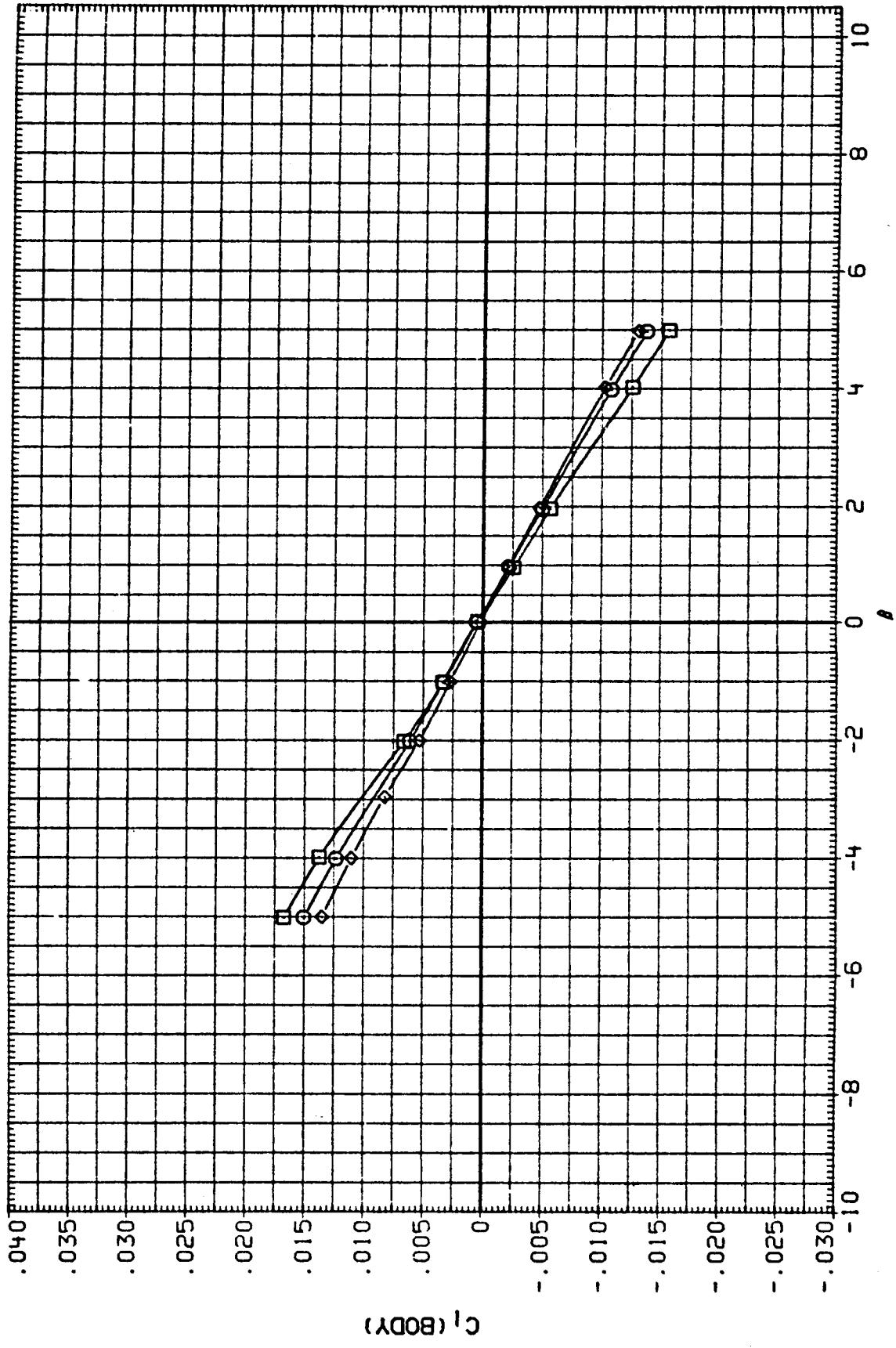
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RF0008	□	OA250	B26C9G15M16FBW116E43V8R5TC19
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(A) MACH = .20

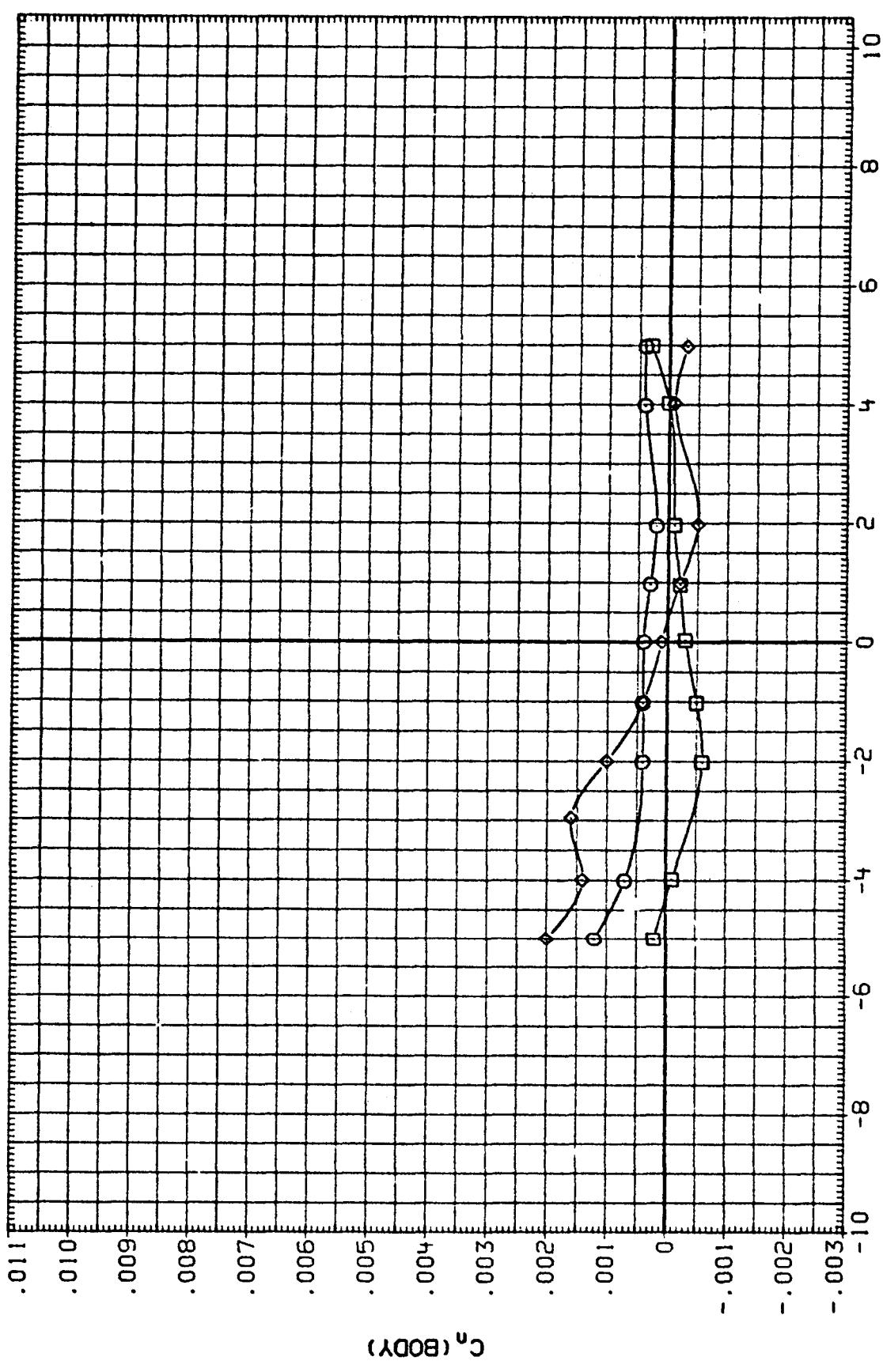
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EFFECT OF ELEVON DEFLECTION, H/B = 0.178

(A) MACH = .20

## EFFECT OF ELEVON DEFLECTION, H/B = 0.178

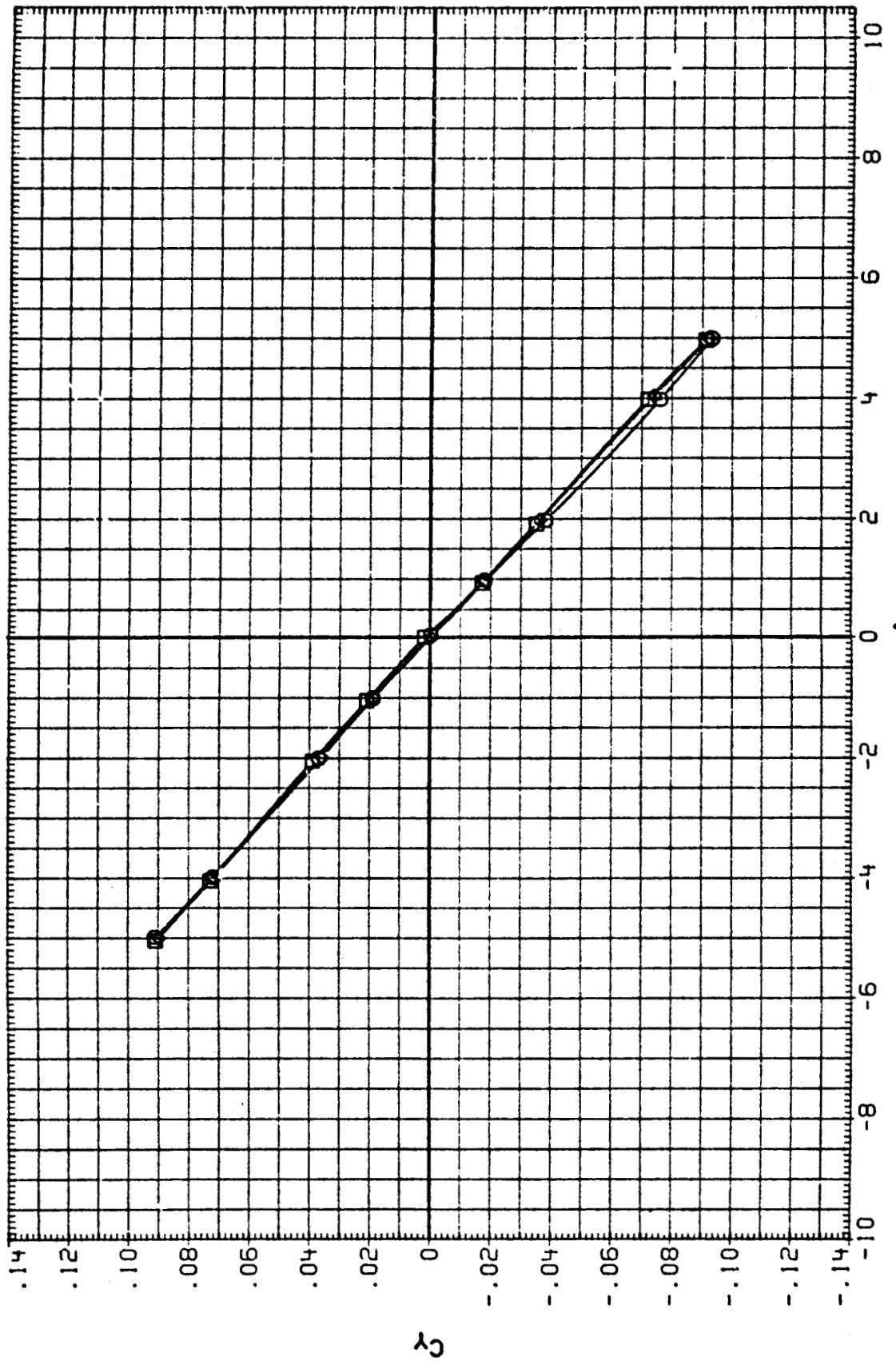


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RF 0010	◊	0A250	B26C9G15H16F8W116E43V8R5TC19

ELEVON H/B AILRDN ALPHA BOFLAP RUDDER SPDRK

.000	.125	.000	10.000	-12.000	.000	.000
5.000	.125	.000	10.000	-12.000	.000	.000
-5.000	.125	.000	10.000	-12.000	.000	.000



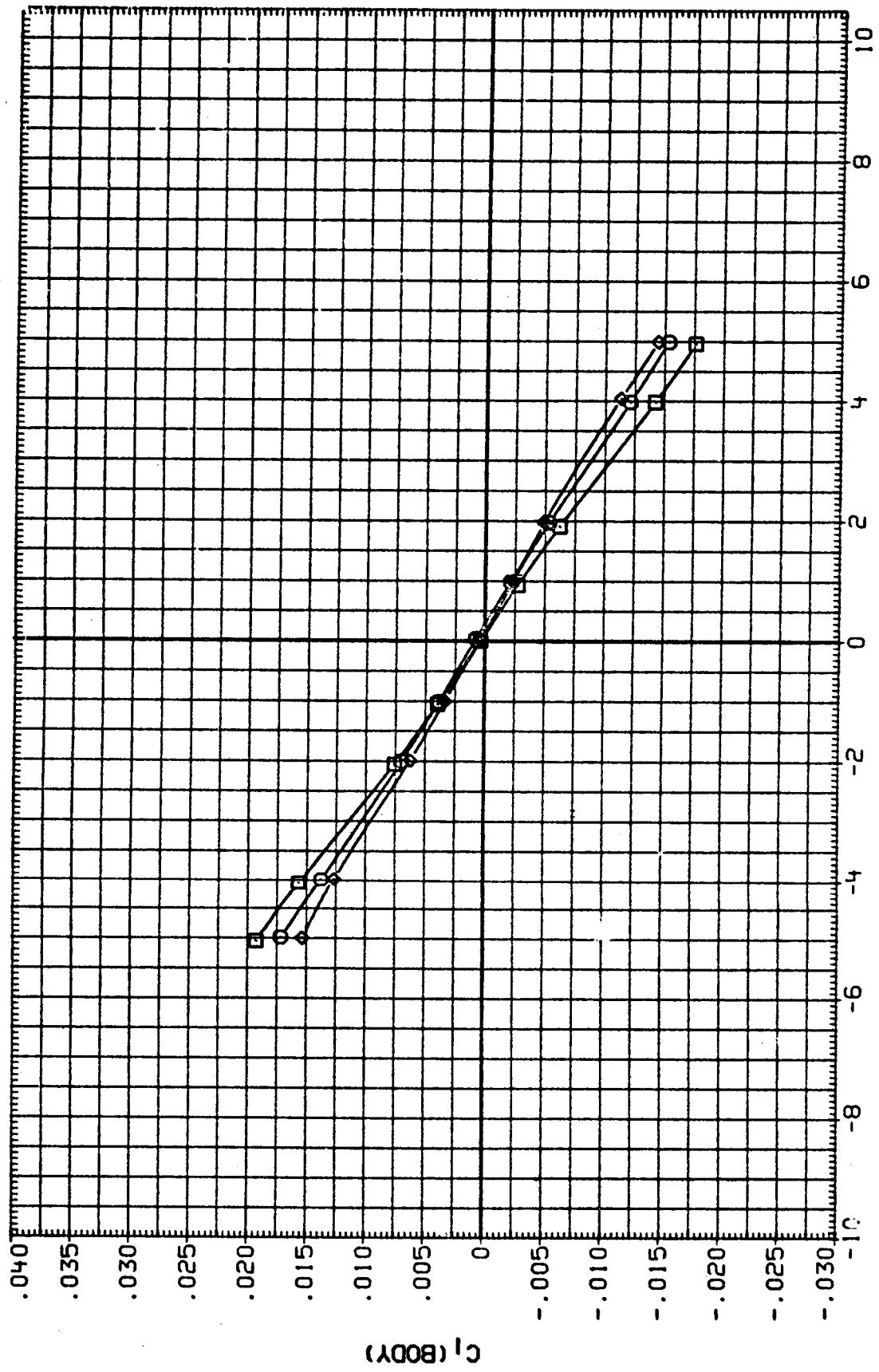
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(A) MACH = .20

PAGE 16

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ELEVON	H/B	AILRON	ALPHA	SOFLAP	RUDDER	SPDBRK
.000	.125	.000	.000	-12.000	.000	.000
.500	.125	.000	.000	-12.000	.000	.000
-5.000	.125	.000	.000	-12.000	.000	.000

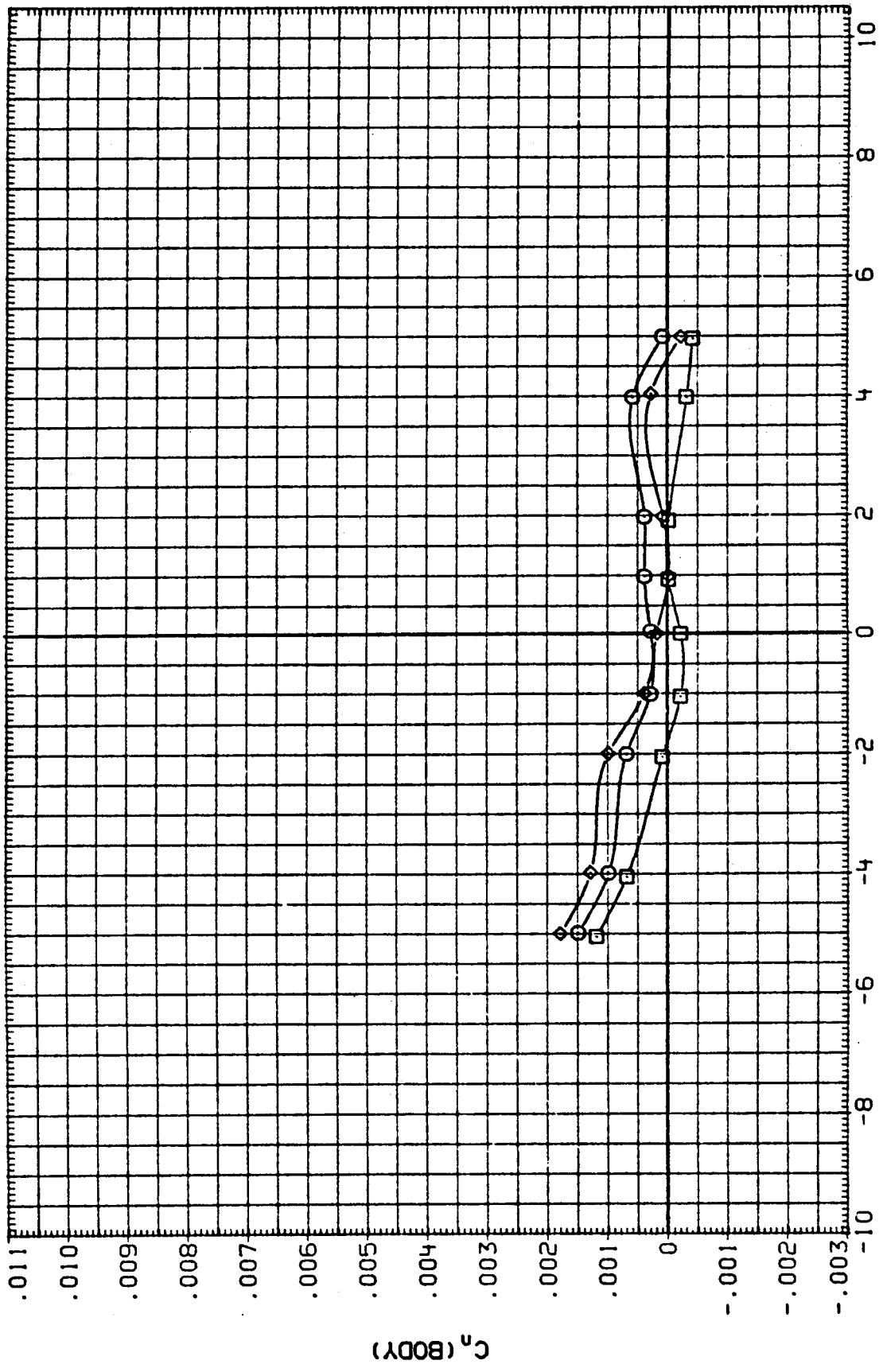


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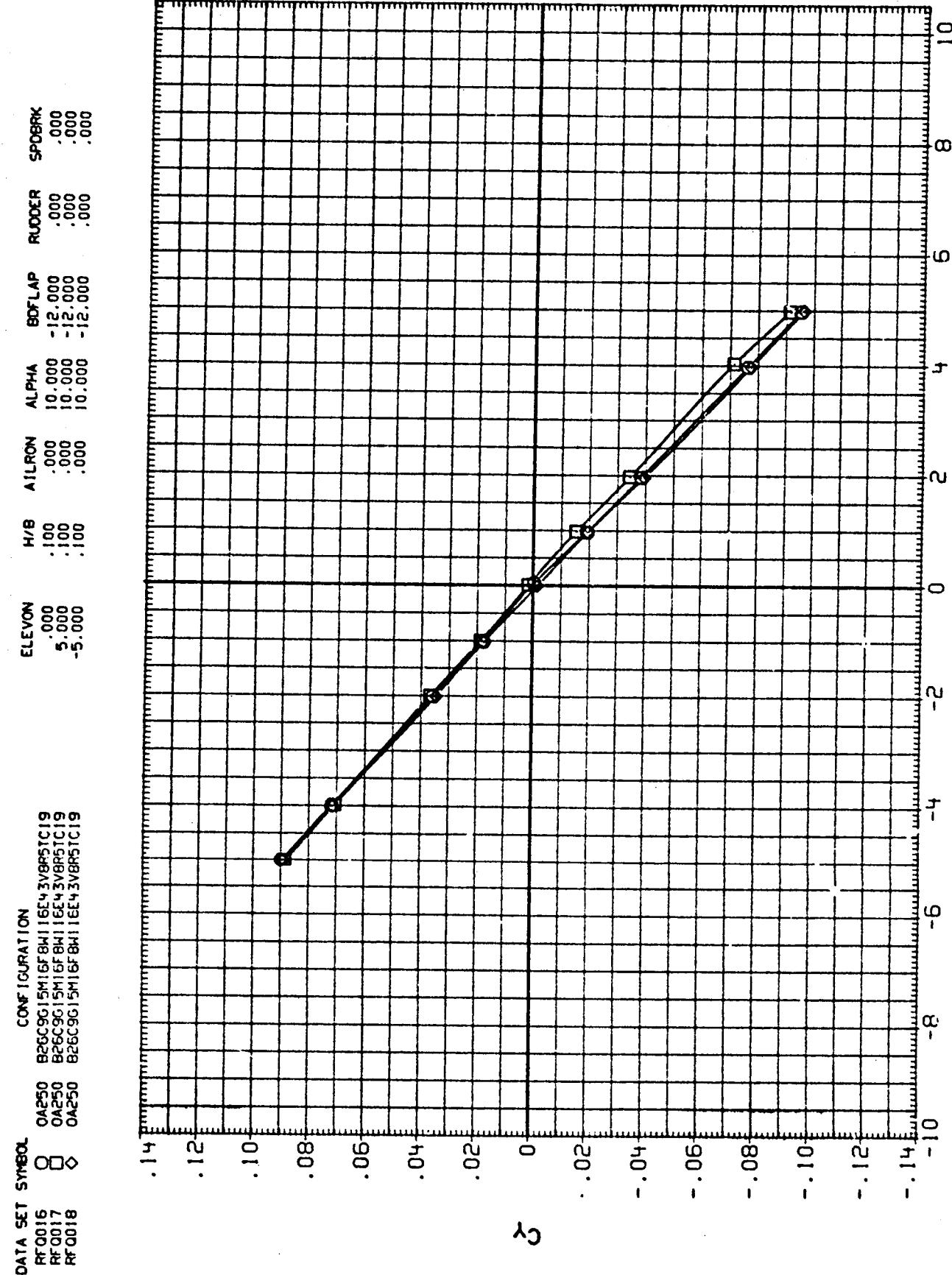
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EFFECT OF ELEVON DEFLECTION, H/B = 0.125

(A) MACH = .20

PAGE 18



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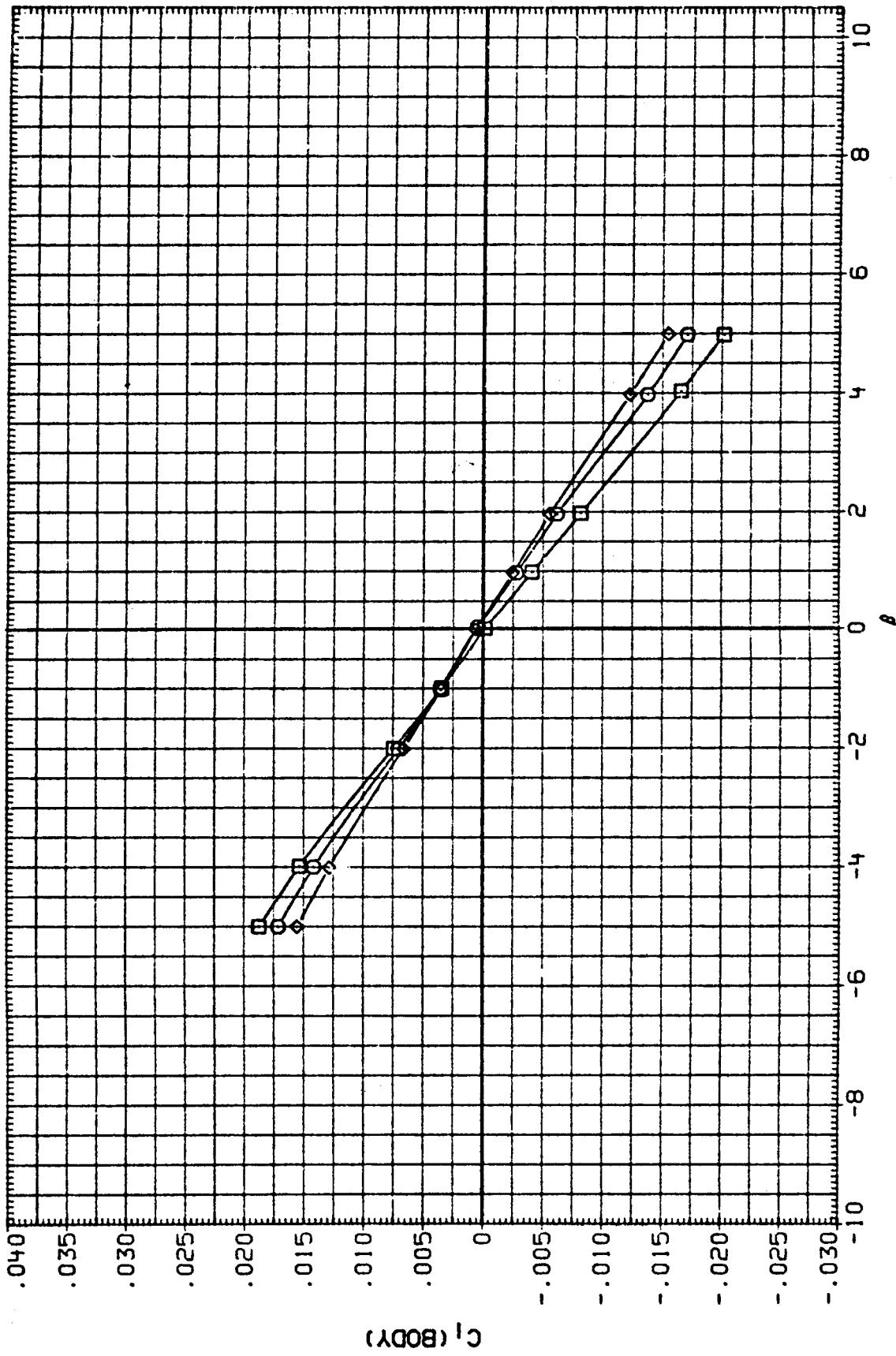
RF0016	O
RF0017	□
RF0018	◇

CONFIGURATION

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B26C9G15M16FBW116E43V8R5TC19
B26C9G15M16FBW116E43V8R5TC19

ELEVON H/B AILERON ALPHA BOFLAP RUDDER SPDBRK

.000 .100 .000 10.000 -12.000 .000
5.000 .100 .000 10.000 -12.000 .000
-5.000 .100 .000 10.000 -12.000 .000



EFFECT OF ELEVON DEFLECTION, H/B = 0.10

(A) MACH = .20

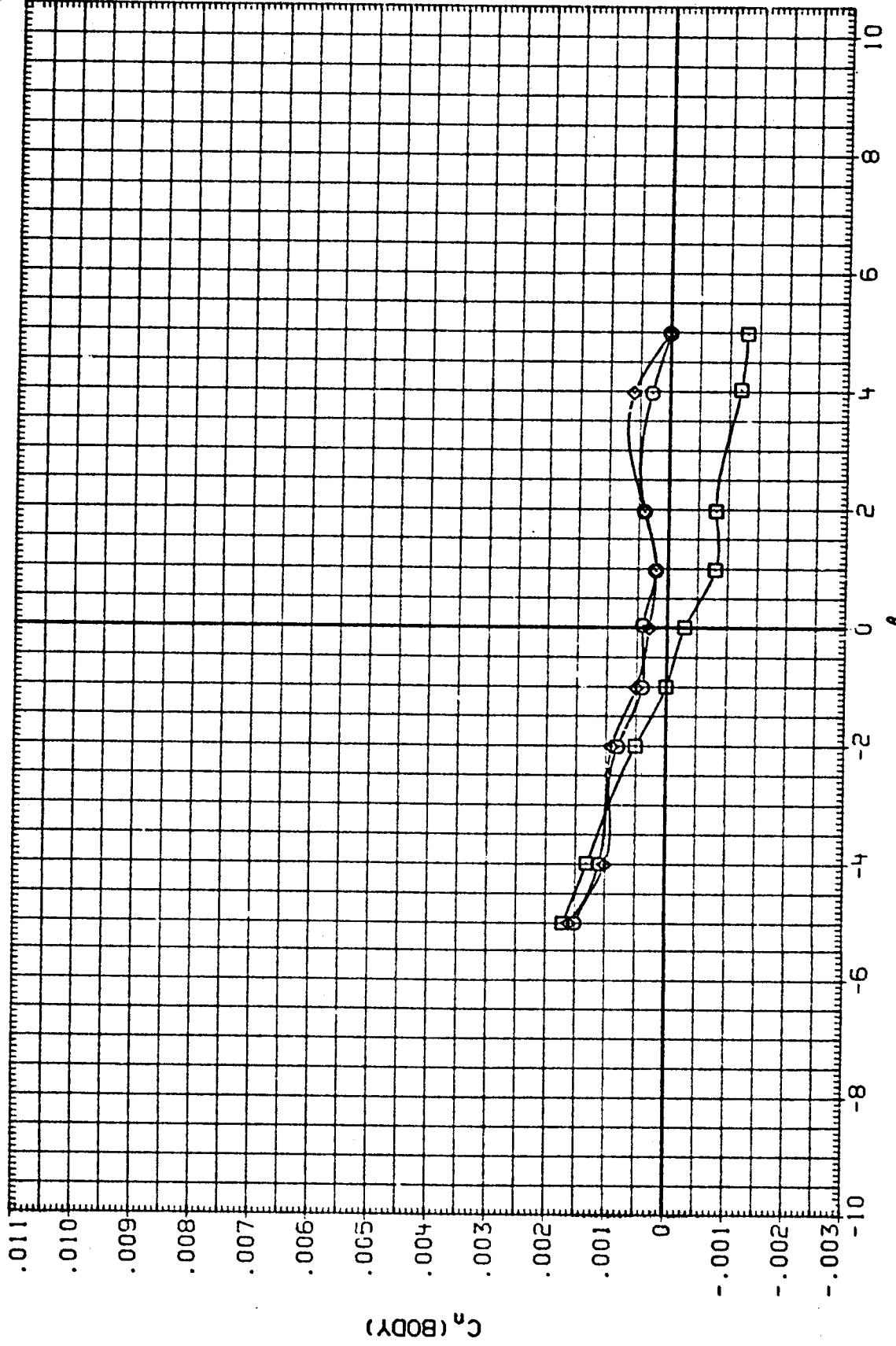
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ELEVON H/B AIRRON ALPHA EDFLAP RUDDER SPOILER

.000	.100	.000	10.000	-12.000	.000
5.000	.100	.000	10.000	-12.000	.000
-5.000	.100	.000	10.000	-12.000	.000



EFFECT OF ELEVON DEFLECTION, H/B = 0.10

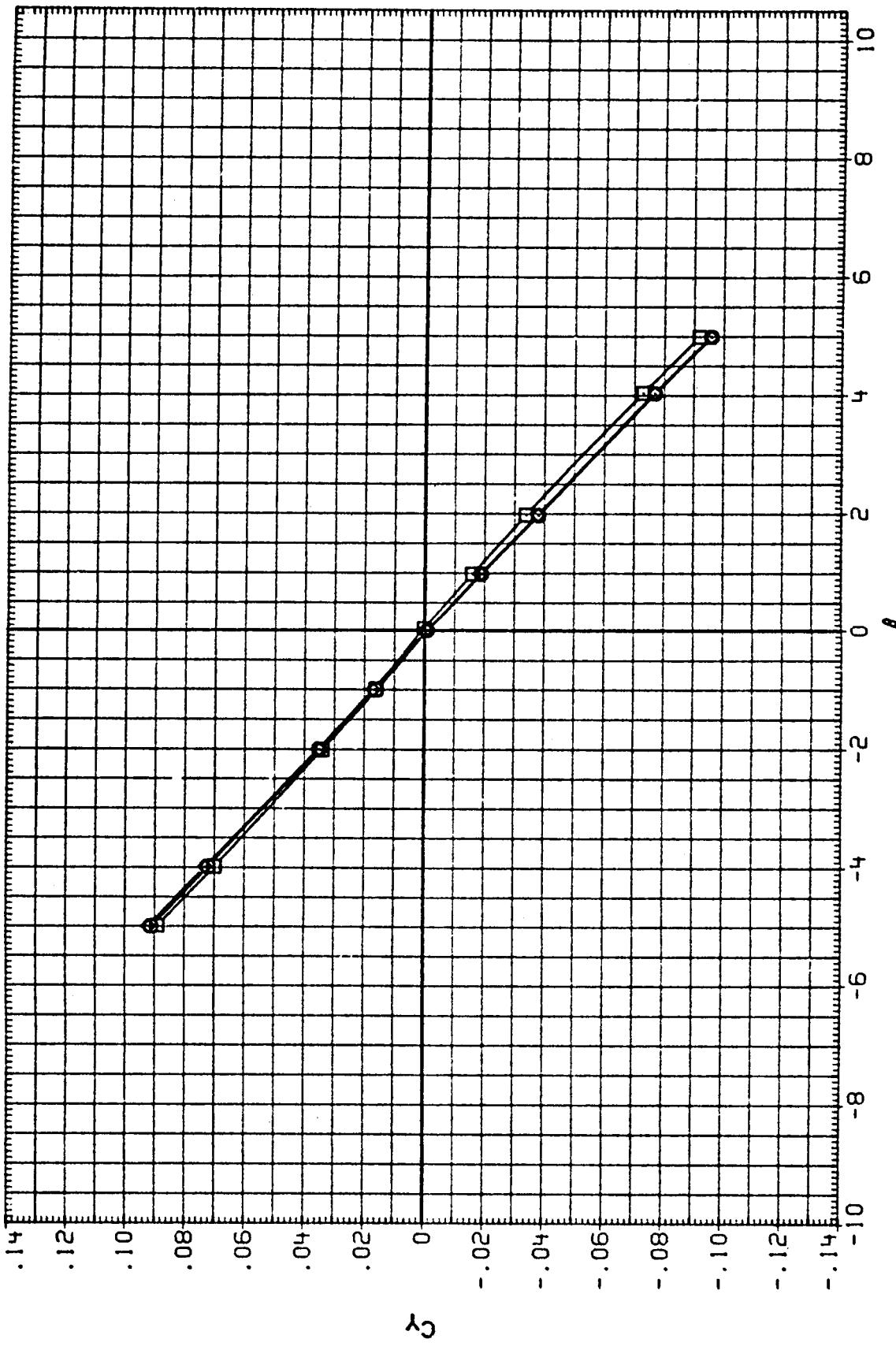
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RF 0022	□	0A250	E26C3015M16FBW116E4 3VBR51C19
RF 0023	◊	0A250	E26C9015M16FBW116E4 3VBR51C19

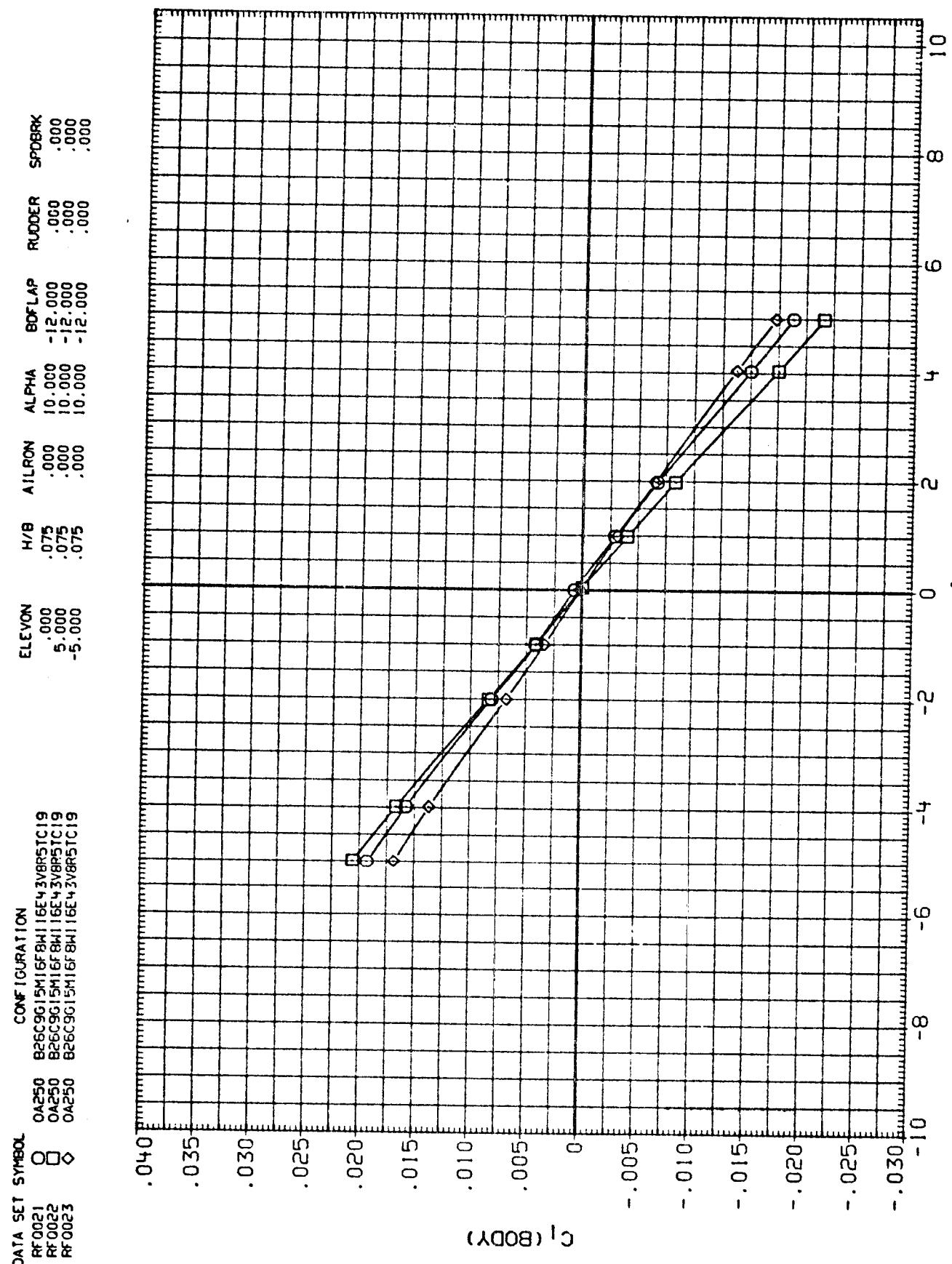
CONFIGURATION

ELEVON	H/B	AIRLON	ALPHA	BLFLAP	RUDDER	SPDBRK
.000	.075	.000	10.000	-12.000	.000	.000
5.000	.075	.000	10.000	-12.000	.000	.000
-5.000	.075	.000	10.000	-12.000	.000	.000



EFFECT OF ELEVON DEFLECTION, H/B = 0.075

(A) MACH = .20

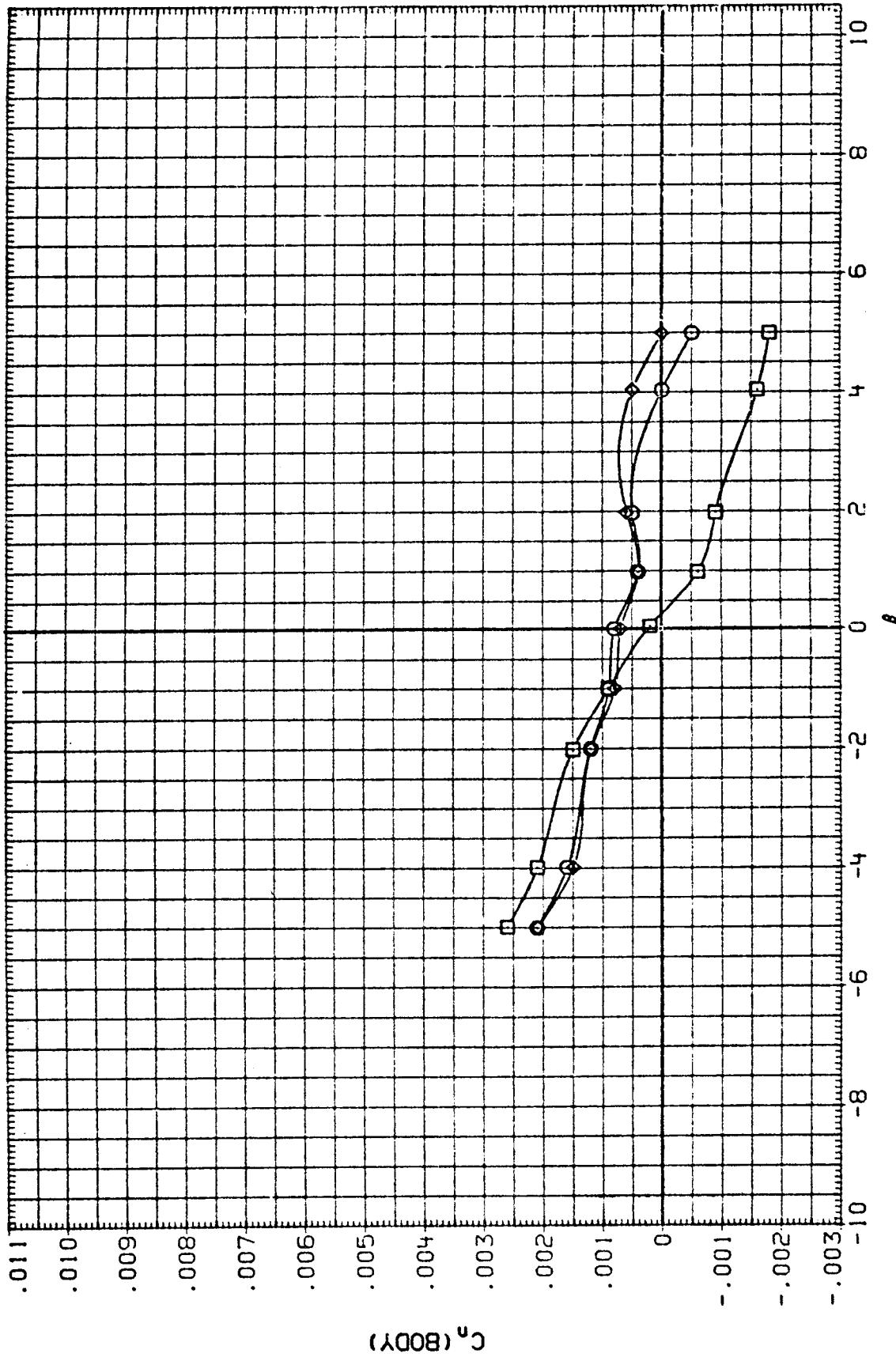


EFFECT OF ELEVON DEFLECTION, H/B = 0.075

(A)MACH = .20

DATA SET SYMBOL CONFIGURATION  
 RF0021 O B26C9615M16FBW16E43VBRSTC19  
 RF0022 □ B26C9615M16FBW16E43VBRSTC19  
 RF0023 ◇ B26C9615M16FBW16E43VBRSTC19

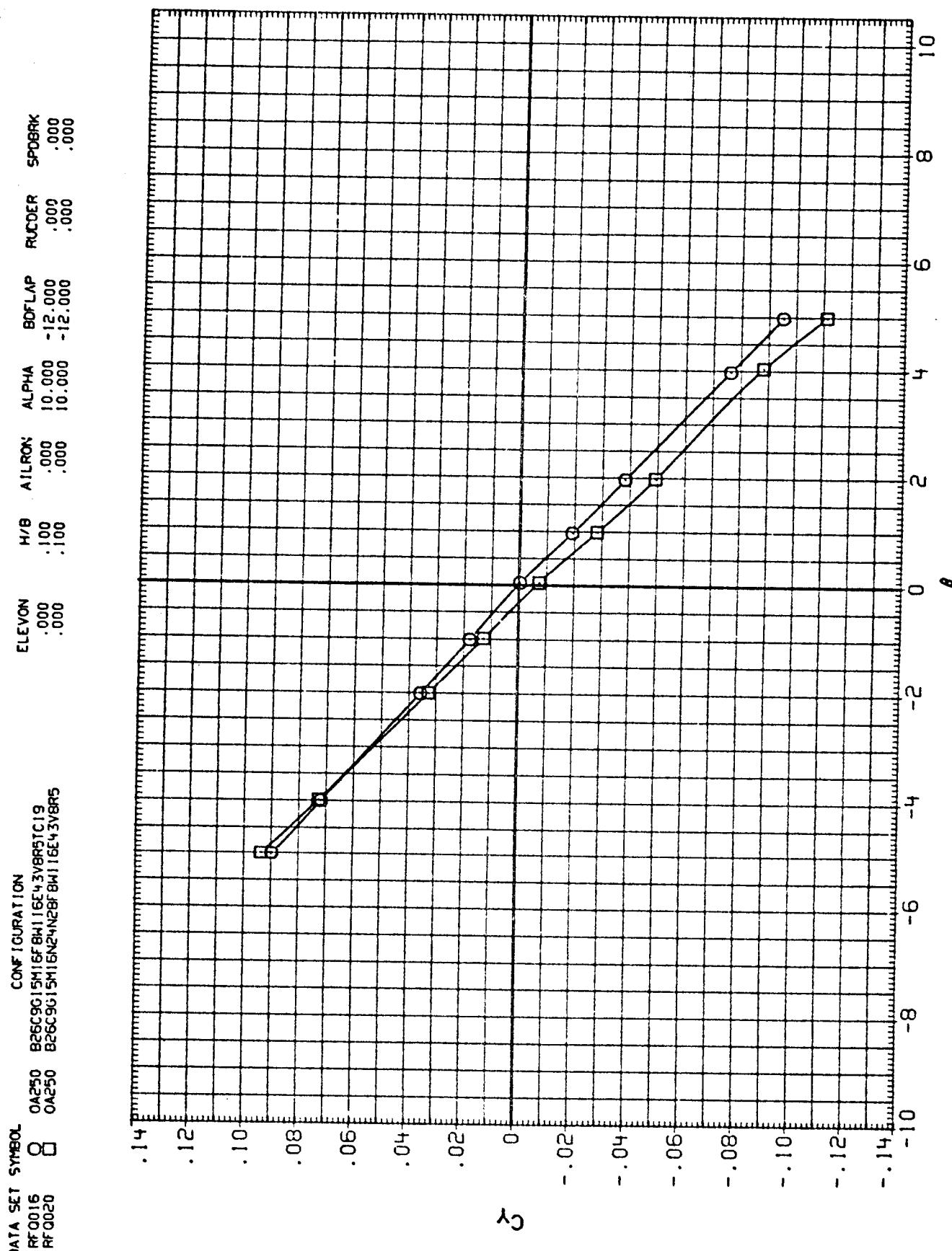
ELEVON H/B AILRDN ALPHA BOFLAP RUDDER SPDRK  
 .000 .000 10.000 -12.000 .000 .000  
 5.000 .075 .000 10.000 -12.000 .000 .000  
 -5.000 .075 .000 10.000 -12.000 .000 .000



EFFECT OF ELEVON DEFLECTION, H/B = 0.075

(A) MACH = .20

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DATA SET SYMBOL

CONFIGURATION

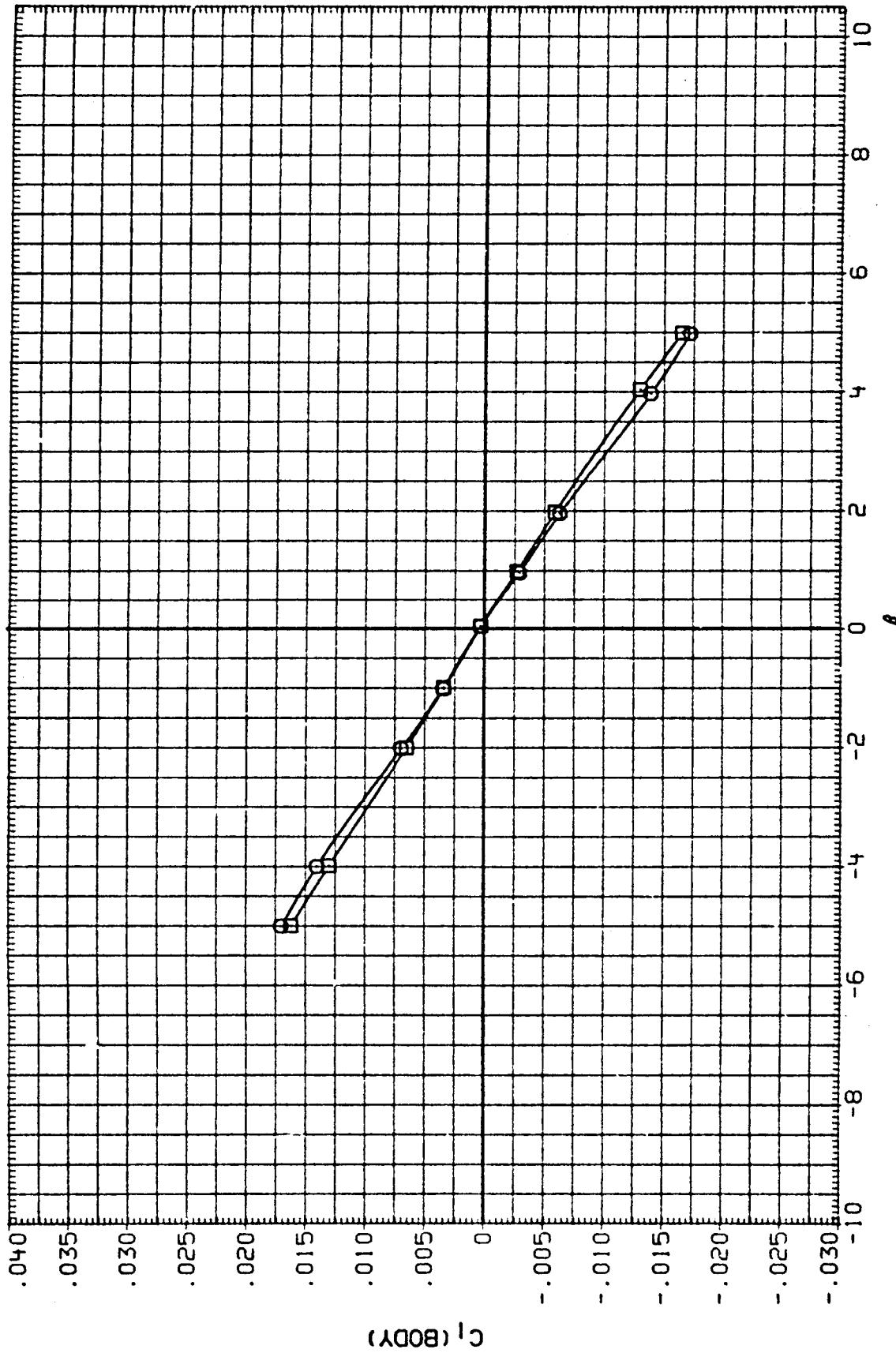
RF0016 O OA250 B2EC9615M16FBH116E43V8R5TC19

RF0020 O OA250 B2EC9615M16N24N28H116E43V8R5

ELEVON H/B AILRON ALPHA BDFLAP RUDDER SPOILER

.000 .100 .000 10.000 -12.000 .000 .000

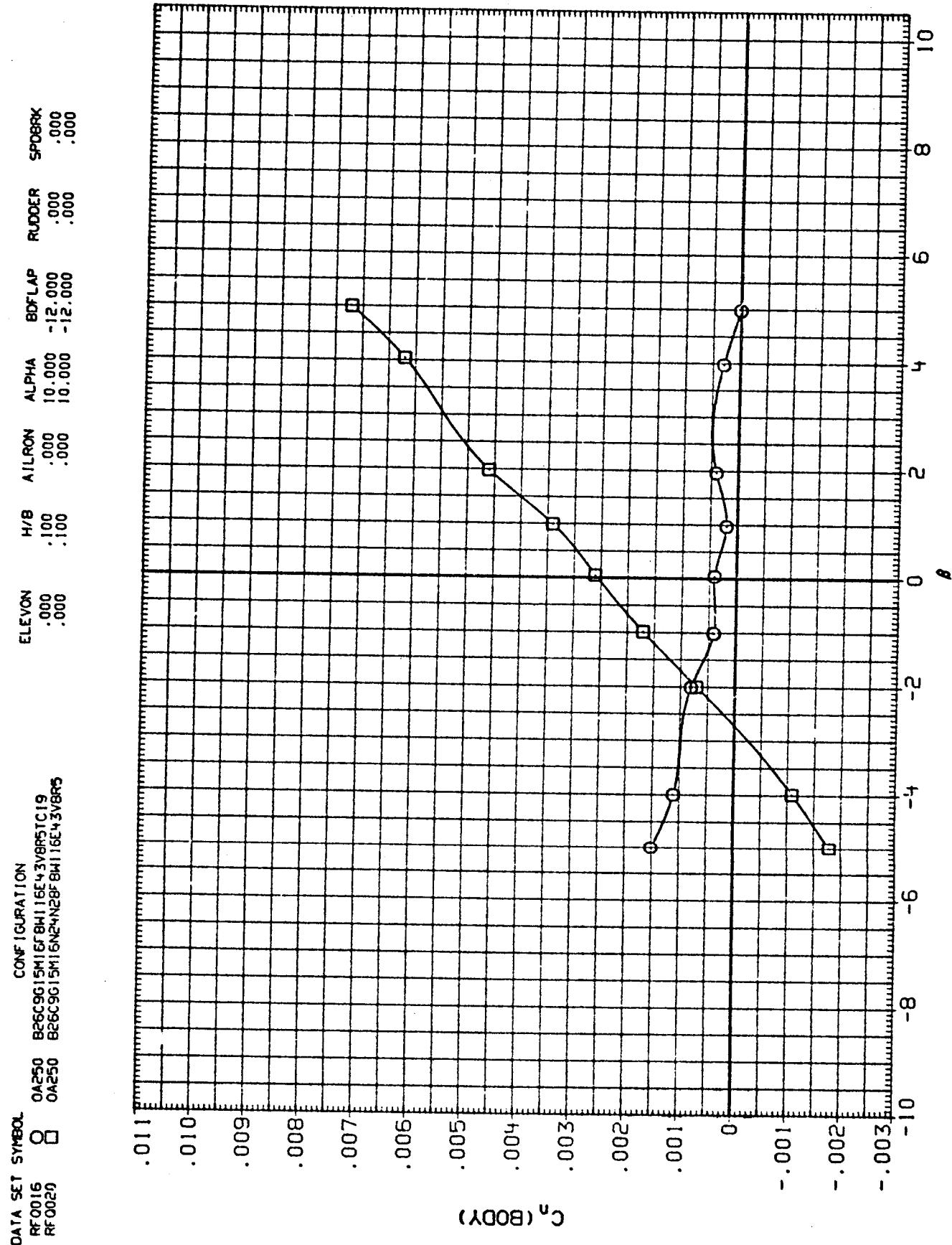
.000 .100 .000 10.000 -12.000 .000 .000



EFFECT OF TAILCONE, H/B = C.10

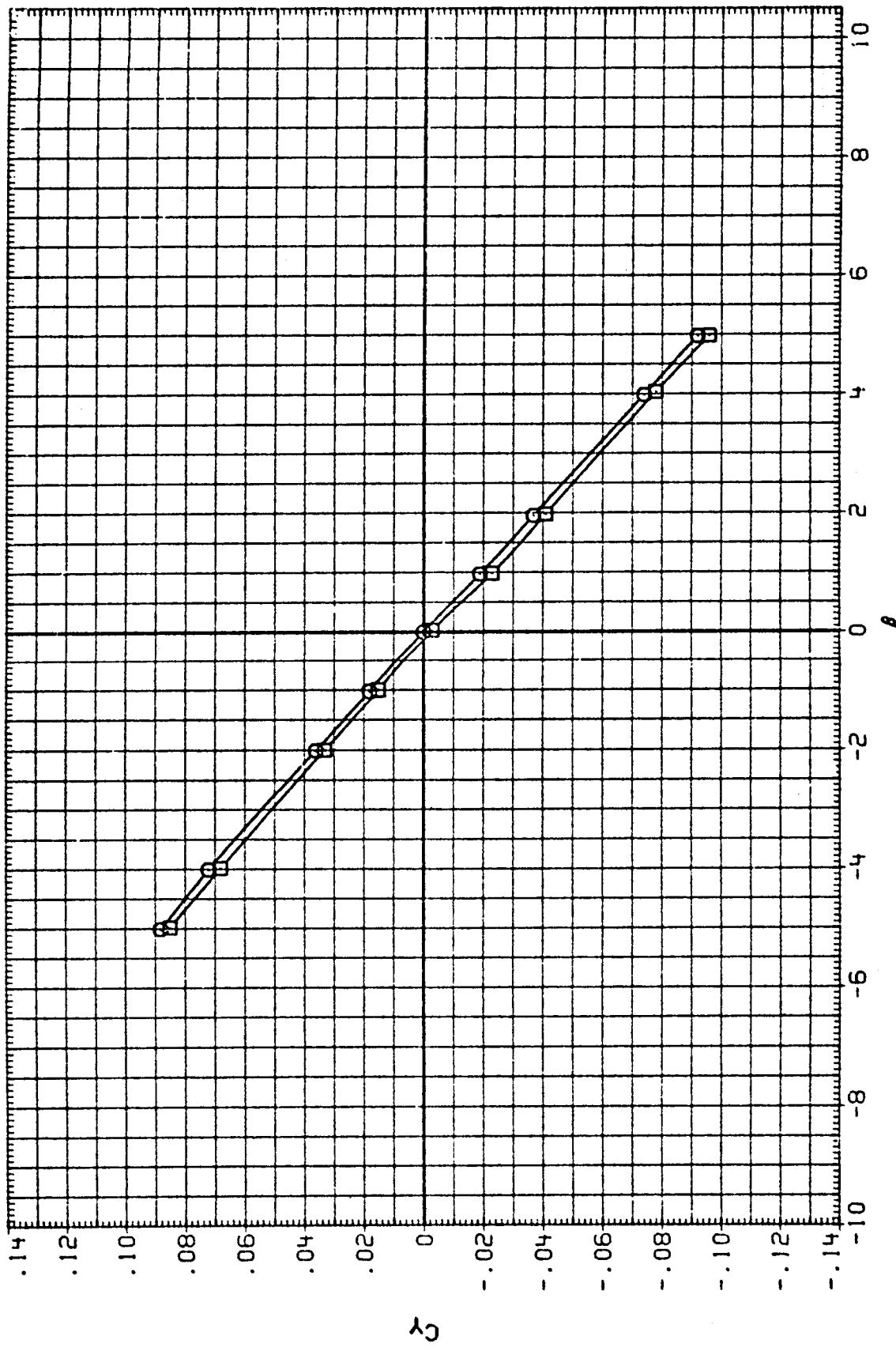
(A) MACH = .20

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DATA SET SYMBOL      CONFIGURATION  
 RF0007      O      B26C9015M16FBW116E43VBRSTC19  
 RF0015      O      B26C9015M16FBW116E43VBRSTC19

	ELEVON	H/B	AIRRON	ALPHA	BOFLAP	RUDDER	SPDBRK
RF0007	.000	.178	.000	10.000	-12.000	.000	.000
RF0015	.000	.178	.000	10.000	-12.000	.000	.000



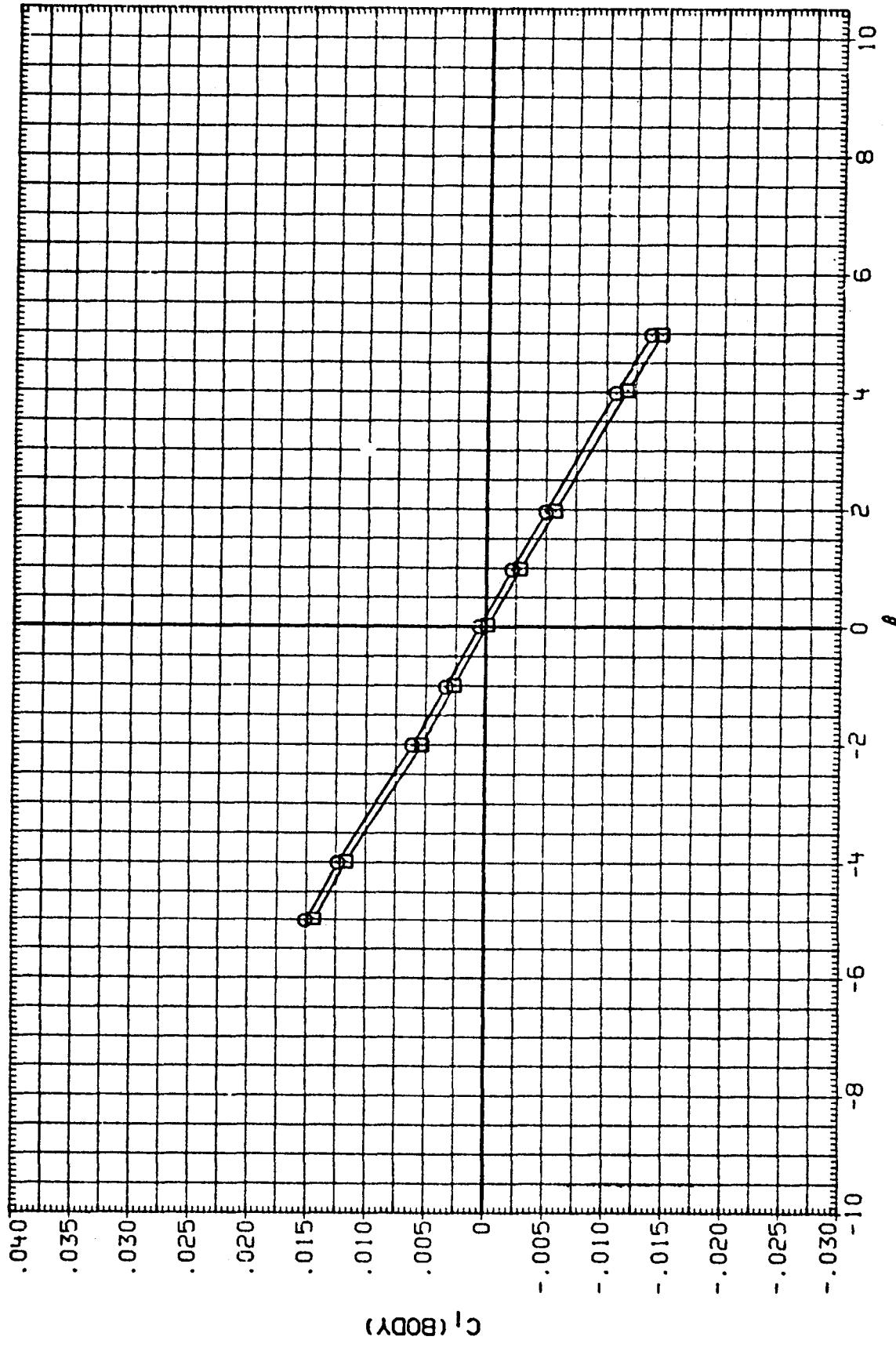
DATA REPEATABILITY, H/B = 0.178

(A) MACH = .20

PAGE 26

DATA SET SYMBOL      CONFIGURATION  
 RF0007      O      826C9G15M16F8W116E43V8R5TC19  
 RF0015      □      826C9G15M16F8W116E43V8R5TC19

	ELEVON	H/B	AILRON	ALPHA	BOFLAP	RUDDER	SPDBRK
RF0007	.000	.178	.000	10.000	-12.000	.000	.000
RF0015	.000	.178	.000	10.000	-12.000	.000	.000



DATA REPEATABILITY, H/B = 0.178

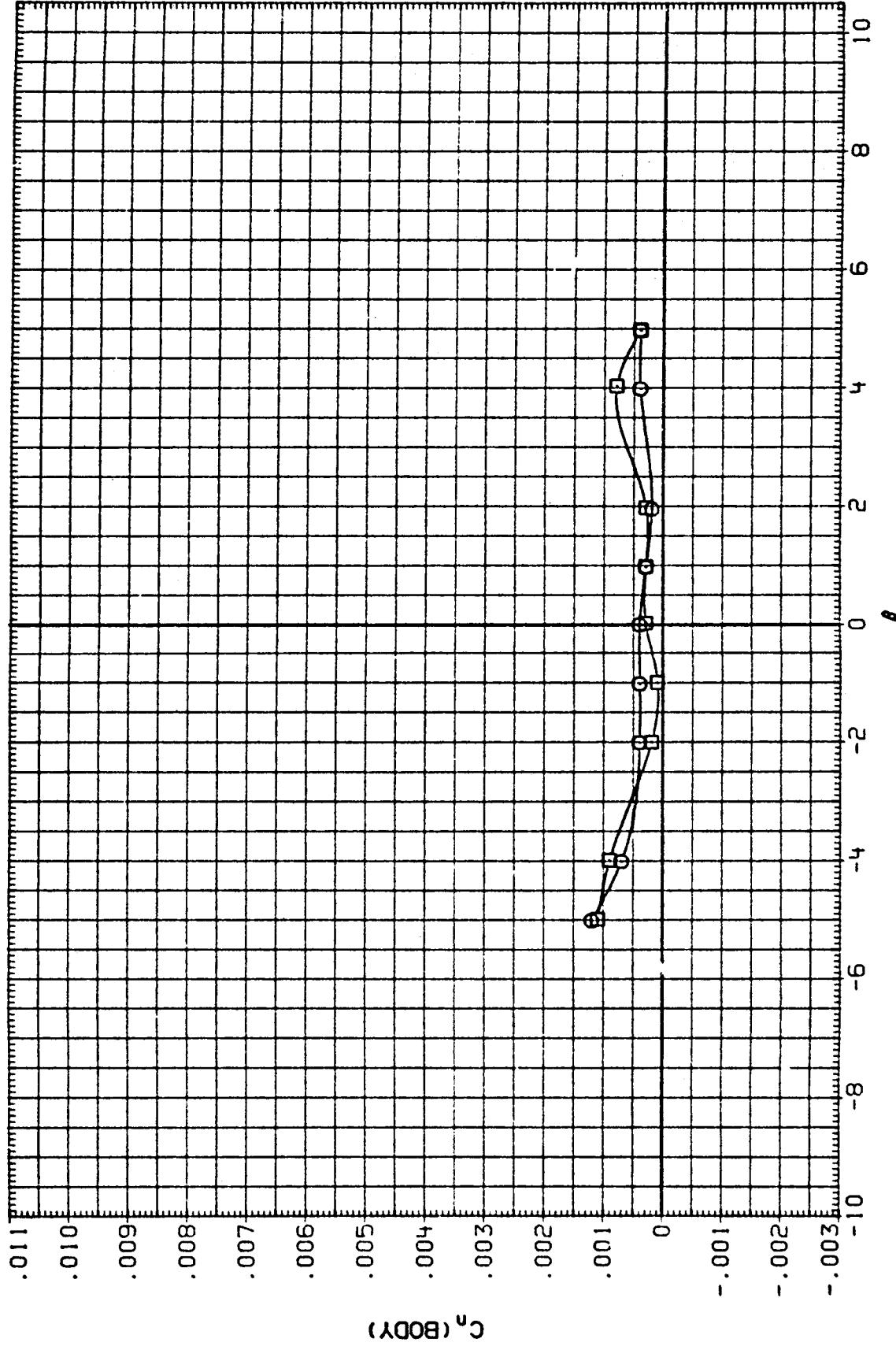
(A) MACH = .200

DATA SET SYMBOL      CONFIGURATION

RF0007	$\square$	0A250	B26C9615M16FBW116E43V8R5TC19
RF0015	$\square$	0A250	B26C9G15M16FBW116E43V8R5TC19

ELEVON      H/B      AILRON      ALPHA      BOFLAP      RUDDER      SPDBRK

.000	.178	.000	10.000	-12.000	.000	.000
.000	.178	.000	10.000	-12.000	.000	.000

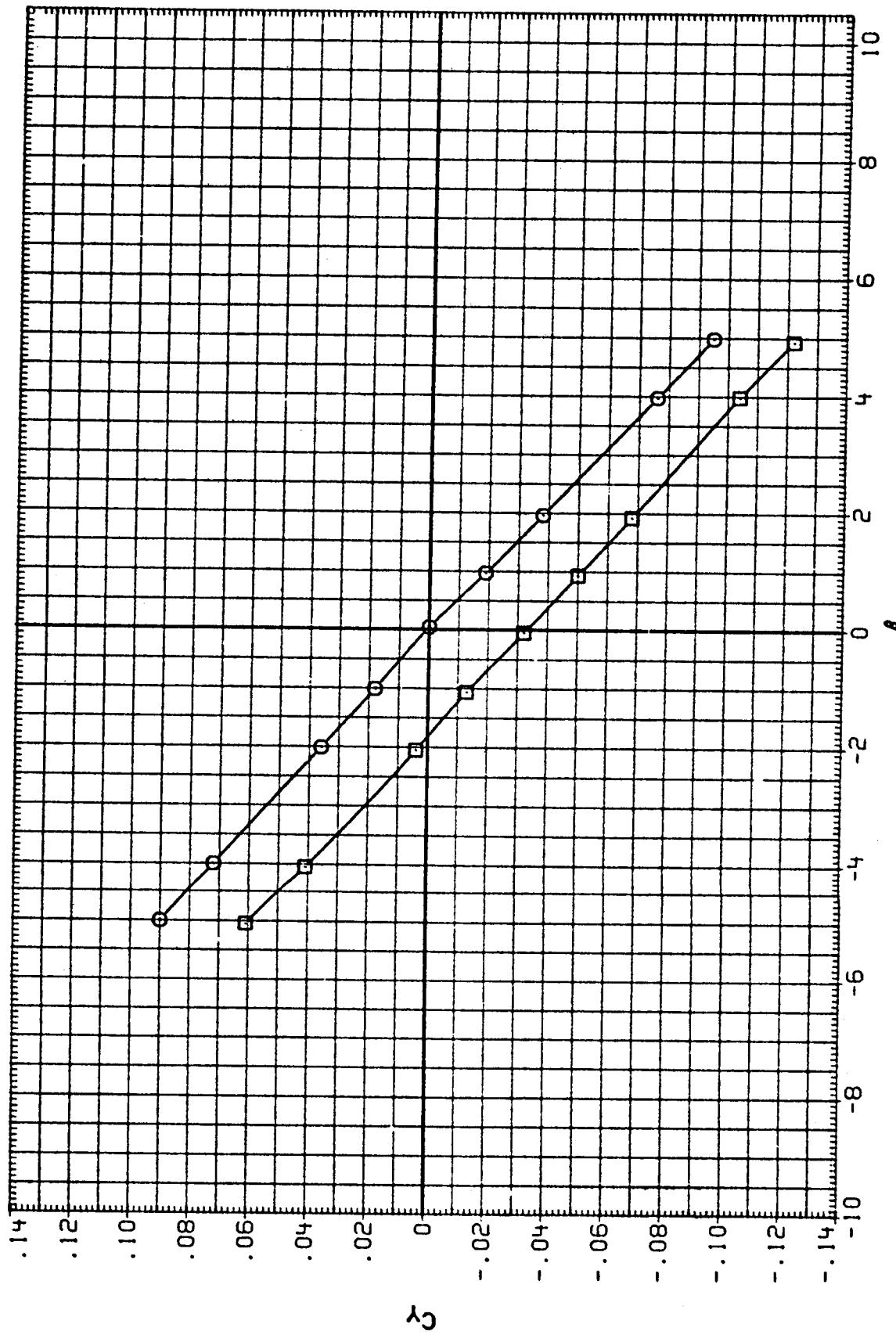


DATA REPEATABILITY, H/B = 0.178

(A) MACH = .20

DATA SET SYMBOL CONFIGURATION  
 RF0016 O B26C9G15M16F8W116E43V8R5TC19  
 RF0019 □ B26C9G15M16F8W116E43V8R5TC19

ELEVON	H/B	AILERON	ALPHA	BDFLAP	RUDDER	SPDBRK
.000	.100	.000	10.000	-12.000	.000	.000
.000	.100	.000	10.000	-12.000	.000	.000



EFFECT OF AILERON DEFLECTION, H/B = 0.10

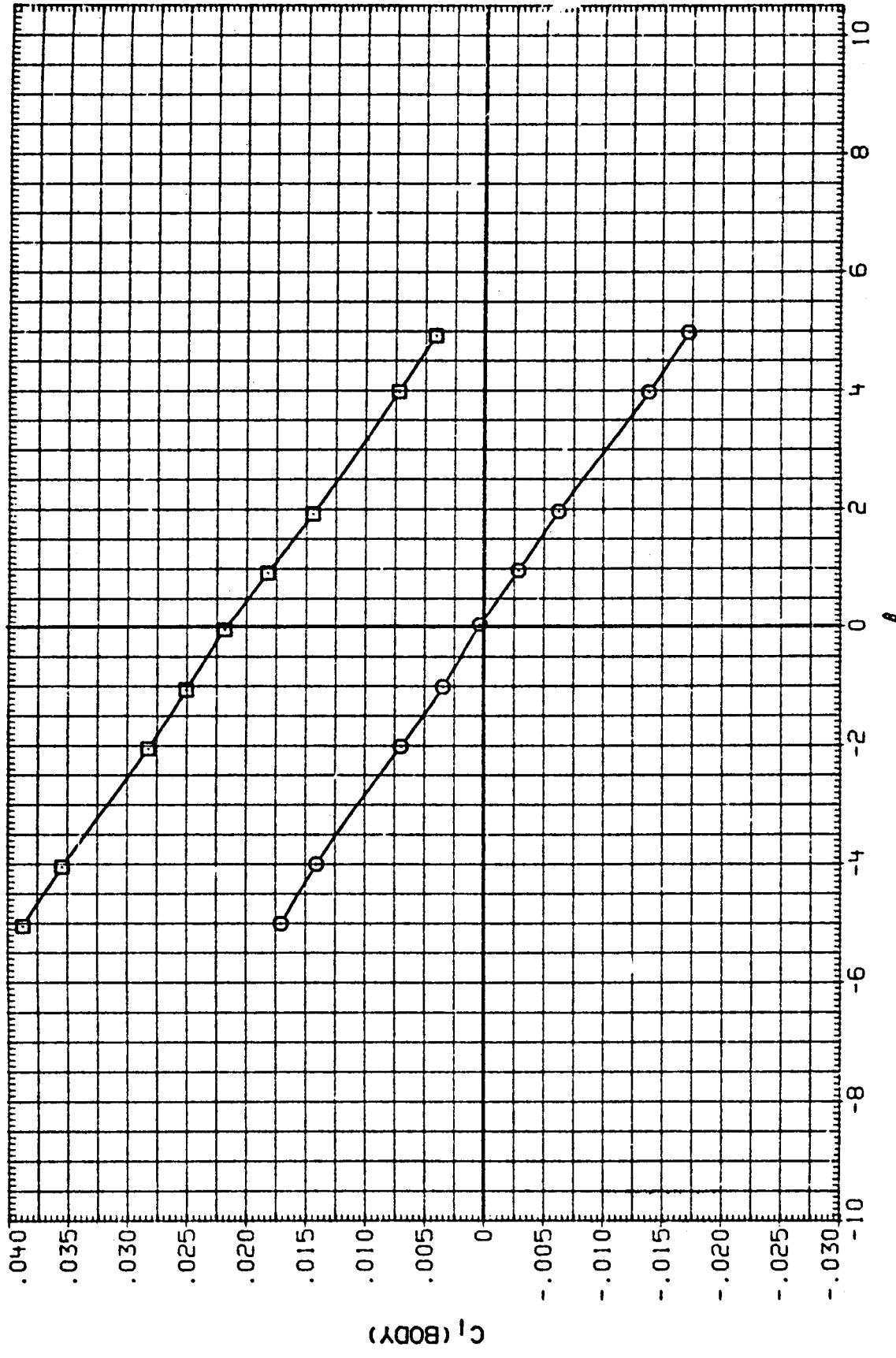
(A) MACH = .20

DATA SET SYMBOL CONFIGURATION

RF0016 O 0A250 B26C9G15M16F8W116E43V8R5TC19  
RF0019 O 0A250 B26C9G15M16F8W116E43V8R5TC19

ELEVON H/B AILERON ALPHALAP RUDER SPDBRK

.000 .100 .000 10.000 -12.000 .000  
.000 .100 .000 10.000 -12.000 .000

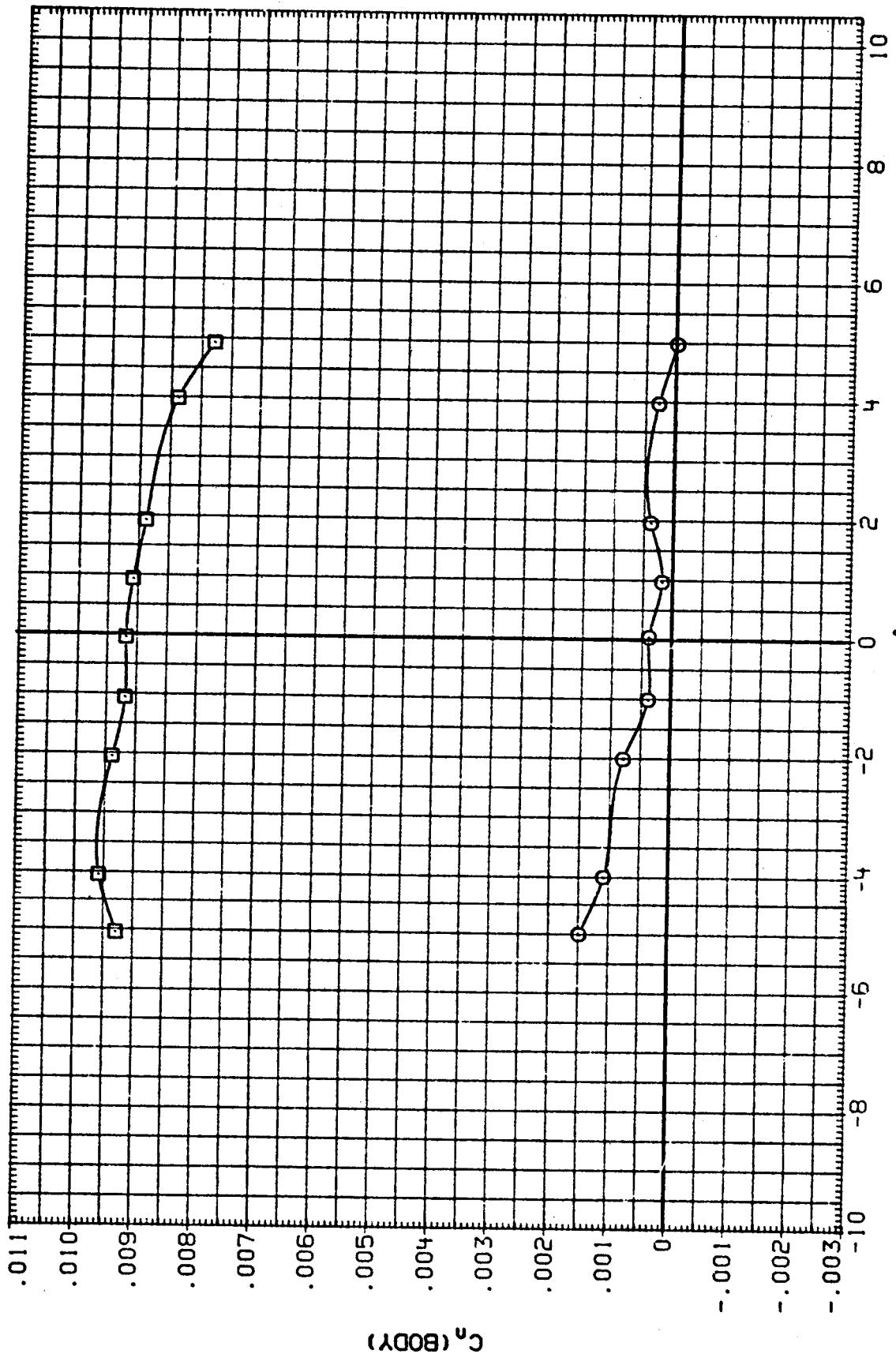


EFFECT OF AILERON DEFLECTION, H/B = 0.10

(A) MACH = .20

DATA SET SYMBOL      CONFIGURATION  
 RF0016      0      B26C9C15M16F8W116E43V8R5TC19  
 RF0019      0A250      B26C9G15M16F8H116E43V8R5TC19

ELEVON	H/B	AILRON	ALPHA	BLFLAP	RUDDER	SPOBRK
.000	.100	.000	10.000	-12.000	.000	.000
.000	.100	.500	10.000	-12.000	.000	.000



EFFECT OF AILERON DEFLECTION,  $H/B = 0.10$

(A) MACH = .20

**APPENDIX**

**Tabulated Source Data**

**Tabulations of plotted data are available on request from Data Management Services.**

DATE 18 NOV 77

OA250, NAAL TEST 775, TABULATED SOURCE DATA

PAGE 1

OA250 B26C9615M16F8W116E43V8R5TC19

(RFQ001) ( 18 NOV 77 )

## REFERENCE DATA

	SREF	LREF	BREF	SCALE	SQ.F.T.	XMRP	YMRP	ZMRP
	2690.0000	474.6000	936.6600	.0299	INCHES	1076.6800	INCHES	INCHES
						.0000	.0000	.0000
						375.0000	INCHES	

RUN NO.	1 / 0	RNL =	1.43	GRADIENT INTERVAL =	-6.00 / 6.00			
MACH	BETA	CL	CLM	CN	XCP/L	CA	CY	CYN
.201	-4.990	.56790	.11410	.04610	.57920	.67900	.01226	.16100
.201	-3.980	.57890	.10880	.04850	.58900	.68000	.00502	.13000
.201	-2.000	.59710	.10390	.05550	.60610	.68300	.00304	.06400
.201	-.990	.59970	.10320	.05340	.60850	.68300	.00416	.02900
.201	.050	.60080	.10290	.05570	.60950	.68300	.00462	.00200
.201	.990	.60130	.10260	.05640	.61000	.68400	.00501	.01800
.201	1.980	.60290	.10220	.05690	.61150	.68400	.00574	.03700
.201	4.040	.59390	.09820	.05210	.60190	.68100	.00802	.07500
.201	5.000	.59280	.09730	.05210	.60070	.68100	.00874	.09400
	GRADIENT	.00215	-.00142	-.00052	.00186	.00118	-.00178	-.02549
								.00205

OA250 B26C9 M16FBW116E43V8R5TC19

(RFQ002) ( 18 NOV 77 )

## REFERENCE DATA

	SREF	LREF	BREF	SCALE	SQ.F.T.	XMRP	YMRP	ZMRP
	2690.0000	474.6000	936.6600	.0299	INCHES	1076.6800	INCHES	INCHES
						.0000	.0000	.0000
						375.0000	INCHES	

RUN NO.	2 / 0	RNL =	1.43	GRADIENT INTERVAL =	-6.00 / 6.00			
MACH	BETA	CL	CLI'	CN	XCP/L	CA	CY	CYN
.201	-4.990	.58150	.07690	-.04610	.58600	.67800	-.02680	.07900
.201	-3.950	.58350	.07760	-.04680	.58810	.67900	-.02645	.06500
.201	-1.990	.58820	.07920	-.04980	.59300	.68000	-.02574	.03200
.201	-1.000	.58950	.07950	-.05020	.59430	.68100	-.02566	.01600
.201	.060	.58950	.07980	-.05020	.59430	.68100	-.02543	.00000
.201	1.000	.59100	.07960	-.05060	.59580	.68100	-.02589	.00010
.201	1.990	.58890	.07910	-.05070	.59370	.68100	-.02599	.03300
.201	4.050	.58390	.07780	-.04700	.58840	.67900	-.02637	.06500
.201	5.000	.58280	.07720	-.04630	.58730	.67900	-.02673	.08100
	GRADIENT	.00012	-.00002	-.00004	.00011	.00008	-.00000	-.01612
								.00023

	ALPHA	AIRRON	BOFLAP	H/B	ELEVON	RUDDER	SPDBRK	CBL
					10.000	.000	-.12.000	.000
							.200	

(RFQ003) ( 18 NOV 77 )

DATE 18 NOV 77

OA250, NAAL TEST 775, TABULATED SOURCE DATA

PAGE 2

OA250 B26C9615M16F8W116E43V8R5TC19

(RFQ003) ( 18 NOV 77 )

## REFERENCE DATA

SREF = 2690.0000 SQ.FT. XMRP = 1076.6800 INCHES  
 LREF = 474.8000 INCHES YMRP = .0000 INCHES  
 BREF = 936.6600 INCHES ZMRP = 375.0000 INCHES  
 SCALE = .0299 SCALE

	RUN NO.	3 / 0	RNL =	1.43	GRADIENT	INTERVAL =	-6.00/	6.00		
MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
.201	-.4 .960	.59070	.09640	-.05220	.59840	.68200	-.00937	.09100	.00000	.01620
.201	-3 .980	.59010	.09690	-.05130	.59790	.68100	-.00882	.07300	-.00010	.01330
.201	-.1 .980	.59750	.10050	-.05550	.60580	.68300	-.00647	.03800	-.00050	.00650
.201	-.070	.59860	.10150	-.05550	.60710	.68300	-.00566	.01900	-.00040	.00330
.201	1 .010	.59870	.10180	-.05550	.60730	.68300	-.00542	.00000	-.00020	.0030
.201	2 .000	.59920	.10140	-.05610	.60770	.68300	-.00587	.01900	.00000	-.00260
.201	4 .000	.59830	.10060	-.05610	.60670	.68100	-.00652	.03700	.00000	-.00560
.201	5 .040	.59320	.09720	-.05210	.60100	.68100	-.00890	.07400	.00030	-.01230
.201	GRADIENT	.59030	.09620	-.05200	.59800	.68200	-.00942	.09200	.00040	-.01560
		.00004	.00000	-.00004	.00014	.00002	-.00001	.01841	.00005	-.00317

OA250 B26C9615M16F8W116E43V8R5TC19

## REFERENCE DATA

SREF = 2690.0000 SQ.FT. XMRP = 1076.6800 INCHES  
 LREF = 474.8000 INCHES YMRP = .0000 INCHES  
 BREF = 936.6600 INCHES ZMRP = 375.0000 INCHES  
 SCALE = .0299 SCALE

	RUN NO.	4 / 0	RNL =	1.43	GRADIENT	INTERVAL =	-6.00/	6.00		
MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
.201	-.5 .030	.48600	.08100	.00000	.49270	.64900	-.00573	.08700	.00110	.01410
.201	-.4 .030	.48540	.08150	-.00140	.49220	.64860	-.00514	.06900	.00080	.01160
.201	-2 .030	.49100	.08510	-.00190	.49840	.65100	-.00259	.3400	.00050	.00560
.201	-1 .030	.49060	.08580	-.00190	.49800	.65100	-.00182	.01600	.00040	.00280
.201	.010	.49150	.08620	-.00140	.49900	.65100	-.00161	.00200	.00030	.00000
.201	.950	.49130	.08570	-.00190	.49870	.65100	-.00208	.02000	.00030	.00260
.201	1 .940	.49200	.08540	-.00210	.49940	.65100	-.00248	.03900	.00030	.00520
.201	4 .010	.48670	.08180	-.00080	.49350	.64900	-.00511	.07500	.00060	.01090
.201	4 .970	.48790	.08140	-.00043	.49460	.65000	-.00569	.09400	.00050	.01380
	GRADIENT	.00019	.00004	-.00004	.00019	.00010	-.00000	.01806	-.00005	-.00279

(RFQ004) ( 18 NOV 77 )

## PARAMETRIC DATA

	ALPHA	AIRRON	ELEVON	RUDDER	SPDBRK
	=	=	=	=	=
	.10 .000	.00 .000	.10 .000	.00 .000	.00 .000
	BDFLAP	H/B			
	=	=			
	-12 .000	.200			

(RFQ003) ( 18 NOV 77 )

DATE 18 NOV 77

0A250, NAAL TEST 775, TABULATED SOURCE DATA  
0A150 B26C9G15M16F8W116E43V8R5TC19

## REFERENCE DATA

SREF	=	2690.0000	SQ.F.T.	XMRP	=	1076.6800	INCHES
LREF	=	474.8000	INCHES	YMRP	=	.0000	INCHES
BREF	=	936.6600	INCHES	ZMRP	=	375.0000	INCHES
SCALE	=	.0299	SCALE				

RUN NO.	/	0	RNL =	1.43	GRADIENT	INTERVAL =	-6.00/	6.00
MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CYN
.201	-5.000	.37210	.06940	.05660	.37850	.59400	.00296	.01190
.201	-4.000	.37100	.07010	.05890	.37760	.59200	.00385	.00150
.201	-2.000	.37370	.07330	.05640	.38070	.59500	.00650	.00150
.201	- .990	.37400	.07400	.05650	.38120	.59500	.00718	.00150
.201	.040	.37390	.07420	.05680	.38110	.59500	.00739	.00100
.201	.980	.37360	.07380	.05630	.38070	.59500	.00702	.00100
.201	1.970	.37410	.07350	.05660	.38120	.59500	.00669	.00240
.201	3.980	.37120	.07010	.05910	.37770	.59200	.00386	.00490
.201	4.990	.37230	.06950	.05680	.37680	.59400	.00303	.01030
	GRADIENT	.00002	.00001	.00002	.00003	.00000	.00001	.01290
								.00261

0A250 B26C9G15M16F8W116E43V8R5TC19

## REFERENCE DATA

SREF	=	2690.0000	SQ.F.T.	XMRP	=	1076.6800	INCHES
LREF	=	474.8000	INCHES	YMRP	=	.0000	INCHES
BREF	=	936.6600	INCHES	ZMRP	=	375.0000	INCHES
SCALE	=	.0299	SCALE				

RUN NO.	/	0	RNL =	1.43	GRADIENT	INTERVAL =	-6.00/	6.00
MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CYN
.201	-4.990	.48650	.08120	-.00050	.49320	.65000	-.00559	.00090
.201	-3.990	.48600	.08180	-.00070	.49280	.64900	-.00498	.00060
.201	-1.990	.49240	.08530	-.00260	.49970	.65100	-.00260	.01170
.201	-1.000	.49170	.08600	-.00260	.49920	.65100	-.00178	.00570
.201	-.050	.49330	.08660	-.00230	.50080	.65100	-.00155	.00290
.201	.990	.49300	.08600	-.00320	.50050	.65200	-.00208	.00100
.201	1.980	.49240	.08580	-.00320	.50160	.65200	-.00248	.00020
.201	4.040	.48830	.08210	-.00000	.49510	.64900	-.00505	.00530
.201	5.020	.48790	.08150	-.00100	.49460	.65000	-.00561	.01110
	GRADIENT	.00023	.00004	-.00008	.00023	.00003	-.00001	.01400
								.00283

ALPHA	=	10.000	ELEVON	=	-5.000
AIRRON	=	.0000	RUDDER	=	.000
BDFLAP	=	-12.000	SPDBRK	=	.000
H/B	=	.200			

(RFQ003) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	.01310
AIRRON	=	.0000	RUDDER	=	.01070
BDFLAP	=	-12.000	SPDBRK	=	.00520
H/B	=	.200			.00260

ALPHA	=	10.000	ELEVON	=	.01310
AIRRON	=	.0000	RUDDER	=	.01070
BDFLAP	=	-12.000	SPDBRK	=	.00520
H/B	=	.200			.00260

(RFQ005) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	.01310
AIRRON	=	.0000	RUDDER	=	.01070
BDFLAP	=	-12.000	SPDBRK	=	.00520
H/B	=	.200			.00260

(RFQ006) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	.01310
AIRRON	=	.0000	RUDDER	=	.01070
BDFLAP	=	-12.000	SPDBRK	=	.00520
H/B	=	.200			.00260

ALPHA	=	10.000	ELEVON	=	.01310
AIRRON	=	.0000	RUDDER	=	.01070
BDFLAP	=	-12.000	SPDBRK	=	.00520
H/B	=	.200			.00260

(RFQ006) ( 18 NOV 77 )

DATE 18 NOV 77

## OA250. NAAL TEST 775, TABULATED SOURCE DATA

OA250 B26C9615M16F8H116E43V8R5TC19

## REFERENCE DATA

SREF	=	2690.0000	SQ.FT.	XMRP	=	1076.6800	INCHES
LREF	=	474.8000	INCHES	YMRP	=	.0000	INCHES
BREF	=	936.6600	INCHES	ZMRP	=	375.0000	INCHES
SCALE	=	.0299	SCALE				

MACH	BETA	RUN NO.	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
			.08100	.00010	.49210	.6900	-.00567	.08800	.00120	.01500
.201	-5.020		.48550	.00160	.49320	.64800	-.00508	.07200	.00070	.01230
.201	-4.010		.48640	.08170	.49870	.65100	-.00277	.03600	.00040	.00610
.201	-2.010		.49140	.08500	.49940	.65100	-.00201	.01800	.00040	.00330
.201	-1.020		.49190	.08590	.50090	.65100	-.00169	.00000	.00040	.00050
.201	.000		.49340	.08640	.50190	.65100	-.00204	.01900	.00030	-.00220
.201	.970		.49200	.08580	.49940	.65100	-.00259	.03700	.00020	-.00500
.201	1.950		.49420	.08570	.50150	.65100	-.00240	.00507	.00040	.01080
.201	3.980		.48940	.08230	.49630	.64900	-.00070	.07400	.00040	.01380
.201	4.970		.48900	.08150	.49570	.65000	-.00050	.09200	.00040	-.01380
	GRADIENT		.00038	.00007	.00039	.00010	-.00005	.01815	-.00005	-.00288

OA250 B26C9615M16F8H116E43V8R5TC19

## REFERENCE DATA

SREF	=	2690.0000	SQ.FT.	XMRP	=	1076.6800	INCHES
LREF	=	474.8000	INCHES	YMRP	=	.0000	INCHES
BREF	=	936.6600	INCHES	ZMRP	=	375.0000	INCHES
SCALE	=	.0299	SCALE				

MACH	BETA	RUN NO.	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
			.05110	-.04990	.59740	.68100	-.00964	.09100	.00020	.01670
.201	-5.020		.58970	.09590	.59660	.68000	-.00898	.07300	-.00010	.01370
.201	-3.990		.58880	.09640	.59530	.68200	-.00684	.03900	-.00060	.00660
.201	-2.020		.59670	.10000	.60500	.68200	-.00603	.02000	-.00050	.00330
.201	-1.020		.59880	.10100	.60410	.68200	-.00561	.00100	-.00030	.00040
.201	.030		.59990	.10160	.60410	.68300	-.00603	.01700	-.00020	.00260
.201	.960		.59910	.10120	.60460	.68300	-.00665	.03600	-.00010	.00570
.201	1.960		.59940	.10070	.60510	.68300	-.00895	.07200	-.00000	.01260
.201	4.020		.59320	.09730	.60140	.68100	-.00960	.09100	.00030	-.01570
.201	4.990		.59260	.09650	.60220	.60030	-.00015	.00042	-.01822	-.00323
	GRADIENT		.00042	.00009		.00007				

(RFQ007) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	.000
AILRDN	=	10.000	RUDDER	=	.000
BDFLAP	=	-12.000	SPDBRK	=	.000
H/B	=	.178			

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(RFQ008) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	5.000
AILRDN	=	10.000	RUDDER	=	.000
BDFLAP	=	-12.000	SPDBRK	=	.000
H/B	=	.178			

DATE 18 NOV 77

0A250. NAAL TEST 775. TABULATED SOURCE DATA

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0A250 B26C9615M16F8W116E43V8R5TC19

## REFERENCE DATA

SREF	2690.0000 SQ.FT.	XMRP	1076.6800 INCHES
LREF	474.8000 INCHES	YMRP	.0000 INCHES
BREF	936.6600 INCHES	ZMRP	375.0000 INCHES
SCALE	.0299 SCALE		

RUN NO. 9/ 0 RN/L = 1.43 GRADIENT INTERVAL = -6.00/ 6.00

MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
.201	-5.010	.36860	.06920	.05890	.37500	.59200	.00356	.08700	.00200	.01350
.201	-4.000	.36690	.06970	.06140	.37340	.58900	.00434	.07000	.00140	.01100
.201	-2.960	.36720	.07130	.06110	.37400	.58900	.00580	.05100	.00160	.00820
.201	-2.000	.37030	.07390	.05930	.37730	.59200	.00695	.03400	.00100	.00530
.201	-1.010	.37060	.07390	.05920	.37780	.59200	.00775	.01700	.00040	.00270
.201	.000	.37100	.07410	.05940	.37820	.59200	.00793	.00000	.0010	.00020
.201	.980	.37130	.07370	.05920	.37840	.59200	.00751	.01700	.00020	.00220
.201	1.970	.37200	.07350	.05880	.37910	.59200	.00715	.03500	.00050	.00480
.201	4.020	.37080	.07040	.05970	.37750	.59100	.00434	.07200	.0010	.01030
.201	4.980	.37120	.06980	.05740	.37770	.59400	.00363	.08900	.00030	.01310
.201	GRADIENT	.00041	.00008	-.00020	.00043	.00025	.00001	-.01760	-.00024	-.00265

0A250 B26C9615M16F8W116E43V8R5TC19

## REFERENCE DATA

SREF	2690.0000 SQ.FT.	XMRP	1076.6800 INCHES
LREF	474.8000 INCHES	YMRP	.0000 INCHES
BREF	936.6600 INCHES	ZMRP	375.0000 INCHES
SCALE	.0299 SCALE		

RUN NO. 10/ 0 RN/L = 1.43 GRADIENT INTERVAL = -6.00/ 6.00

MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
.201	-5.000	.39140	.07140	.06750	.39780	.58700	.00162	.09000	.00180	.01540
.201	-3.980	.38780	.07150	.06990	.39430	.58400	.00240	.07200	.00130	.01270
.201	-1.990	.39270	.07500	.06830	.39980	.58700	.00498	.03600	.00100	.00630
.201	-1.000	.39360	.07570	.06710	.40080	.58800	.00548	.01900	.00040	.00340
.201	.000	.39520	.07640	.06700	.40240	.58800	.00586	.00000	.00020	.00050
.201	.990	.39600	.07630	.06680	.40330	.58800	.00565	.01800	.00000	.00230
.201	1.980	.39730	.07580	.06640	.40440	.58900	.00488	.03600	.0010	.00490
.201	4.040	.39840	.07340	.06600	.40520	.58900	.00239	.07400	.00030	.01130
.201	4.990	.40220	.07350	.06270	.40880	.59300	.00188	.09200	-.00020	-.01440
.201	GRADIENT	.00117	.00022	-.00047	.00120	.00059	.00001	-.01821	-.00018	-.00297

(RFQ009) ( 18 NOV 77 )

PARAMETRIC DATA

ALPHA	10.000	ELEVON	-5.000
AIRLON	.000	RUDDER	.000
BDFLAP	-12.000	SPDBRK	.000
H/B	.125		

(RFQ10) ( 18 NOV 77 )

PARAMETRIC DATA

ALPHA	10.000	ELEVON	-5.000
AIRLON	.000	RUDDER	.000
BDFLAP	-12.000	SPDBRK	.000
H/B	.125		

DATE 18 NOV 77

## OA250. NAAL TEST 775. TABULATED SOURCE DATA

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OA250 B26C9G15M16F8W116E43V8R5TC19

## REFERENCE DATA

SREF	2690.0000	SQ.FT.	XMRP	=	1076.6800	INCHES						
LREF	474.8000	INCHES	YMRP	=	.0000	INCHES						
BREF	936.6600	INCHES	ZMRP	=	375.0000	INCHES						
SCALE	.0299	SCALE										
MACH	BETA		RUN NO.	11 / 0	R/N/L =	1.43	GRADIENT INTERVAL =	-6.00/	6.00			
.201	-5.050	CL	CD	.09740	CLM	CN	XCP/L	CA	CY	CYN	CBL	
.201	-4.050	.61620	.09740	-.04660	.62370	.67700	-.01285	.09100	.00120	.01930		
.201	-2.050	.61420	.09740	-.04540	.62180	.67600	-.01252	.07300	.00070	.01570		
.201	-1.050	.62280	.10040	-.05030	.63070	.67900	-.01112	.03900	.00010	.00760		
.201	.000	.62550	.10150	-.05130	.63360	.67900	-.01052	.02100	-.00020	.00400		
.201	.930	.62600	.10250	-.05160	.63430	.67900	-.00960	.00200	-.00020	.00040		
.201	1.910	.62730	.10310	-.05200	.63570	.68000	-.00930	.01700	.00000	-.00270		
.201	3.980	.62680	.10350	-.05240	.63520	.68000	-.00878	.03500	.00000	-.00620		
.201	4.960	.62390	.10160	-.04980	.63210	.67900	-.01014	.07200	-.00330	-.01420		
.201	GRADIENT	.62530	.10110	-.05130	.63330	.67900	-.01082	.09100	-.00440	-.01760		
		.00102	.00047	-.00050	.00109	.00027	.00028	.01820	-.00013	-.00368		

OA250 B26C9G15M16F8W116E43V8R5TC19

## REFERENCE DATA

SREF	2690.0000	SQ.FT.	XMRP	=	1076.6800	INCHES						
LREF	474.8000	INCHES	YMRP	=	.0000	INCHES						
BREF	936.6600	INCHES	ZMRP	=	375.0000	INCHES						
SCALE	.0299	SCALE										
MACH	BETA		RUN NO.	12 / 0	R/N/L =	1.43	GRADIENT INTERVAL =	-6.00/	6.00			
.201	-5.010	CL	CD	.09710	CLM	CN	XCP/L	CA	CY	CYN	CBL	
.201	-4.000	.61760	.09710	-.04830	.62510	.67800	-.01344	.09200	.00090	.01930		
.201	-2.010	.61520	.09690	-.04680	.62270	.67700	-.01314	.07400	.00030	.01560		
.201	-1.020	.62420	.09960	-.05150	.63200	.67900	-.01210	.03900	-.00010	.00740		
.201	-.010	.62610	.10060	-.05200	.63400	.68000	-.01153	.02100	-.00020	.00380		
.201	.970	.62570	.10110	-.05200	.63370	.68000	-.01095	.00200	-.00010	.00030		
.201	1.960	.62780	.10110	-.05280	.63580	.68000	-.01134	.01700	-.00000	-.00310		
.201	3.980	.62700	.10000	-.05300	.63490	.68000	-.01224	.03500	.00010	-.00660		
.201	4.980	.62440	.09850	-.05100	.63200	.67900	-.01327	.07200	.00000	-.01450		
		.00015	.09830	-.05230	.63300	.68000	-.01368	.09100	-.00020	-.01800		
		.00091	.00015	-.00044	.00092	.00022	-.00002	.01836	-.00007	-.00372		

(RFQ011) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	5.000
AIRRON	=	.000	RUDER	=	.000
BDFLAP	=	-12.000	SPDBRK	=	.000
H/B	=	.125			
CBL					

(RFQ012) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	5.000
AIRRON	=	.000	RUDER	=	.000
BDFLAP	=	-12.000	SPDBRK	=	.000
H/B	=	.125			
CBL					

DATE 18 NOV 77

## OA250. NAAL TEST 775. TABULATED SOURCE DATA

OA250 B26C9615M16F8W116E43V8R5TC19

## REFERENCE DATA

SREF	2690.0000	SQ.FT.	XMRP	=	1076.6800	INCHES
LREF	474.8000	INCHES	YMRP	=	.0000	INCHES
BREF	936.6600	INCHES	ZMRP	=	375.0000	INCHES
SCALE	.0299	SCALE				

RUN NO.	13/ 0	RNL =	1.43	GRADIENT INTERVAL =	-6.00/ 6.00
MACH	BETA	CL	CD	CN	XCP/L
.201	-4.980	.49390	.08640	.50140	.64300
.201	-3.990	.49340	.08670	.00910	.64300
.201	-1.990	.49780	.08980	.00740	.50590
.201	-1.010	.49910	.09040	.00630	.50720
.201	.000	.49890	.09080	.00730	.50710
.201	.990	.50000	.09090	.00680	.50820
.201	1.980	.50080	.09020	.00680	.50890
.201	4.040	.50140	.08780	.00740	.50910
.201	4.990	.50480	.08780	.00490	.51240
	GRADIENT	.00102	.00014	-.00026	.00103
				.00023	
				-.00004	
				-.01853	

OA250 B26C9615M16F8W116E43V8R5TC19

## REFERENCE DATA

SREF	2690.0000	SQ.FT.	XMRP	=	1076.6800	INCHES
LREF	474.8000	INCHES	YMRP	=	.0000	INCHES
BREF	936.6600	INCHES	ZMRP	=	375.0000	INCHES
SCALE	.0299	SCALE				

RUN NO.	14/ 0	RNL =	1.43	GRADIENT INTERVAL =	-6.00/ 6.00
MACH	BETA	CL	CD	CN	XCP/L
.201	-4.990	.50900	.08270	.00660	.51560
.201	-3.990	.50800	.08310	.00830	.51470
.201	-2.000	.51450	.08620	.00530	.52160
.201	-1.010	.51650	.08710	.00430	.52380
.201	.040	.51730	.08760	.00400	.52460
.201	.980	.51750	.08750	.00400	.52480
.201	1.970	.51840	.08690	.00390	.52560
.201	3.980	.51800	.08470	.00510	.52490
.201	4.990	.51780	.08440	.00340	.52460
	GRADIENT	.00101	.00018	-.00035	.00103
				.00027	
				-.00000	
				-.01853	

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(RFQ013) ( 18 NOV 77 )

PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	.000
AIRRON	=	.000	RUDDER	=	.000
BDFLAP	=	-12.000	SPDBRK	=	.000
H/B	=	.125			

(RFQ014) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	.000
AIRRON	=	.000	RUDDER	=	.000
BDFLAP	=	-12.000	SPDBRK	=	.000
H/B	=	.125			

(RFQ014) ( 18 NOV 77 )

0A250 B26C9615M16FBW116E43V8R5TC19

(RFQ015) ( 18 NOV 77 )

## REFERENCE DATA

SREF	=	2690.0000	SQ.FT.	XMRP	=	1076.6800	INCHES
LREF	=	474.8000	INCHES	YMRP	=	.0000	INCHES
BREF	=	936.6600	INCHES	ZMRP	=	.375.0000	INCHES
SCALE	=	.0299	SCALE				

MACH	BETA	RUN NO.	15/ 0	RNL = 1.43	GRADIENT INTERVAL = -6.00/ 6.00
.201	-4.990	CL	.08250	.00410	.50140 .64600 -.00579 .08500 .00110 .01430
.201	-3.990	CD	.08330	.00510	.50150 .64600 -.00498 .06800 .00090 .01160
.201	-2.000	CL	.08630	.00170	.50480 .64800 -.00249 .03300 .00020 .00530
.201	-1.000	CD	.08740	.00150	.50510 .64800 -.00148 .01500 .00010 .00260
.201	.030	CL	.08770	.00180	.50430 .64800 -.00102 .00300 .00030 .00290
.201	.980	CD	.08760	.00140	.50530 .64800 -.00132 .02300 .00030 .00290
.201	1.970	CL	.08660	.00150	.50310 .64800 -.00196 .04100 .00030 .00580
.201	4.030	CD	.08400	.00350	.49780 .64700 -.00363 .07800 .00080 .01180
.201	4.980	CL	.08360	.00200	.49750 .64800 -.00393 .09600 .00040 .01470
.201	GRADIENT		.00044	-.00019	-.00041 .00017 -.01824 -.00004 -.00290

0A250 B26C9615M16FBW116E43V8R5TC19

## REFERENCE DATA

SREF	=	2690.0000	SQ.FT.	XMRP	=	1076.6800	INCHES
LREF	=	474.8000	INCHES	YMRP	=	.0000	INCHES
BREF	=	936.6600	INCHES	ZMRP	=	.375.0000	INCHES
SCALE	=	.0299	SCALE				

MACH	BETA	RUN NO.	16/ 0	RNL = 1.43	GRADIENT INTERVAL = -6.00/ 6.00
.201	-5.010	CL	.08220	.00730	.53030 .64400 -.01132 .09000 .00150 .01710
.201	-4.000	CD	.08240	.00900	.52980 .64300 -.01100 .07200 .00110 .01410
.201	-2.010	CL	.08280	.08490	.00660 .53190 .64500 -.00938 .03600 .00700
.201	-1.010	CD	.08290	.08560	.00650 .53610 .64500 -.00906 .01800 .00350
.201	-0.040	CL	.53040	.08590	.00610 .53730 .64500 -.00887 .00000 .00040
.201	.970	CD	.53250	.08580	.00560 .53930 .64600 -.00929 .01900 .00020
.201	1.960	CL	.53180	.08500	.00570 .53840 .64600 -.00999 .03800 .00040
.201	3.970	CD	.53100	.08370	.00730 .53750 .64600 -.01113 .07600 .00620
.201	4.980	CL	.53230	.08370	.00520 .53980 .64600 -.01138 .09500 .00350
.201	GRADIENT		.00089	.00014	-.00022 .00090 .00019 -.01855 -.00002 -.00013

(RFQ016) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	.000
AIRRON	=	.000	RUDDER	=	.000
BDFLAP	=	-12.000	SPDBRK	=	.000
H/B	=	.100			

(RFQ016) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	.000
AIRRON	=	.000	RUDDER	=	.000
BDFLAP	=	-12.000	SPDBRK	=	.000
H/B	=	.100			

(RFQ016) ( 18 NOV 77 )

DATE 18 NOV 77

OA250, NAAL TEST 775, TABULATED SOURCE DATA

(RFC017) ( 18 NOV 77 )

## REFERENCE DATA

SREF	=	2690.0000	SQ.FT.	XMRP	=	1076.6800	INCHES
LREF	=	474.8000	INCHES	YMRP	=	.0000	INCHES
BREF	=	936.6600	INCHES	ZMRP	=	375.0000	INCHES
SCALE	=	.0299	SCALE				

RUN NO.	17/ 0	RNL =	1.43	GRADIENT INTERVAL =	-6.00/ 6.00				
MACH	BETA	CL	CD	CN	XCP/L	CA	CY	CYN	CBL
.201	-5.010	.62850	.09610	.63560	.67600	-.01632	.06900	.00170	.01870
.201	-3.990	.62740	.09620	.63450	.67500	-.01611	.07100	.00130	.01530
.201	-2.000	.63570	.09850	.64820	.64310	-.01526	.03700	.00050	.00740
.201	-1.000	.63650	.09890	.64870	.64401	-.01517	.01900	.00000	.00340
.201	.010	.63900	.09980	.64910	.64661	-.01469	.00200	-.00030	-.00020
.201	.980	.63790	.09910	.64900	.64540	-.01505	.01500	-.00080	-.00410
.201	1.970	.63740	.09850	.64840	.64480	-.01558	.03400	-.00080	-.00820
.201	4.030	.63520	.09770	.64680	.64250	-.01597	.07100	.00120	.01660
.201	4.980	.63670	.09770	.64780	.64400	-.01623	.09100	-.00130	-.02020
	GRADIENT	.00084	.00016	.00023	.00086	.00010	.00001	-.01788	-.00391

OA250 B26C9G15M16F8W116E43V8R5TC19

## REFERENCE DATA

RUN NO.	18/ 0	RNL =	1.43	GRADIENT INTERVAL =	-6.00/ 6.00				
MACH	BETA	CL	CD	CN	XCP/L	CA	CY	CYN	CBL
.201	-5.010	.40020	.07120	.07030	.40650	.58600	-.00015	.09000	.00160
.201	-4.010	.39660	.07120	.07320	.40300	.58300	.00048	.07200	.00100
.201	-2.010	.39980	.07400	.07250	.40560	.58400	.00266	.03500	.00660
.201	-1.020	.40180	.07460	.07210	.40860	.58500	.00288	.01800	.00050
.201	-.010	.40350	.07520	.07160	.41040	.58500	.00325	-.00100	.00350
.201	.970	.40640	.07520	.07030	.41330	.58700	.00266	-.01900	.00060
.201	1.960	.40810	.07470	.06970	.41490	.58800	.00194	-.03900	-.00040
.201	3.970	.40860	.07330	.06900	.41110	.58800	.00043	-.07700	-.00060
.201	4.990	.41390	.07390	.06610	.42040	.59200	.00005	-.09600	-.00000
	GRADIENT	.00150	.00026	-.00049	.00152	.00065	-.00001	-.01863	-.00012

(RFC018) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON	=	-5.000
AIRLON	=	.0000	RUDDER	=	.000
EDFLAP	=	-.12.000	SPDBRK	=	.000
H/B	=	.100			

(RFC018) ( 18 NOV 77 )

DATE 18 NOV 77

0A250, NAAL TEST 775.

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TABULATED SOURCE DATA

(RFQ019) ( 18 NOV 77 )

## REFERENCE DATA

SREF =	2690.0000	SQ.FT.	XMRP =	1076.6800	INCHES
LREF =	474.8000	INCHES	YMRP =	.0000	INCHES
BREF =	936.6600	INCHES	ZMRP =	375.0000	INCHES
SCALE =	.0299	SCALE			

RUN NO. 19/ 0 RN/L = 1.43 GRADIENT INTERVAL = -6.00/ 6.00

MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
.201	-5.050	.51390	.08450	.01370	.52080	.64000	-.00731	.06100	.00930	.03880
.201	-4.050	.51230	.08470	.01500	.51920	.63900	-.00683	.04100	.00960	.03550
.201	-2.050	.51420	.08650	.01440	.52140	.63900	-.00534	.00400	.00940	.02830
.201	-1.060	.51590	.08730	.01370	.52320	.64000	-.00486	.01300	.00920	.02510
.201	- .050	.51760	.08780	.01250	.52490	.64000	-.00468	.03200	.00920	.02190
.201	.930	.51870	.08730	.01130	.52590	.64200	-.00538	.05000	.00910	.01830
.201	1.920	.51890	.08660	.01180	.52610	.64000	-.00613	.06800	.00890	.01450
.201	3.980	.51640	.08530	.01290	.52340	.64000	-.00696	.10400	.00840	.00730
.201	4.930	.52150	.08580	.00990	.52840	.64300	-.00742	.12200	.00780	.00420
	GRADIENT	.00072	.00010	-.00038	.00073	.00027	-.00003	-.01823	-.00015	-.00348

0A250 B26C9G15M16N24N28F8W116E43V8R5

## REFERENCE DATA

SREF =	2690.0000	SQ.FT.	XMRP =	1076.6800	INCHES
LREF =	474.8000	INCHES	YMRP =	.0000	INCHES
BREF =	936.6600	INCHES	ZMRP =	375.0000	INCHES
SCALE =	.0299	SCALE			

RUN NO. 20/ 0 RN/L = 1.43 GRADIENT INTERVAL = -6.00/ 6.00

MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
.201	-5.000	.47470	.11330	.04050	.48730	.61900	.02808	.09400	-.00180	.01630
.201	-3.990	.47330	.11210	.04210	.48560	.61800	.02716	.07300	-.00110	.01310
.201	-2.000	.47460	.11380	.04330	.48720	.61700	.02854	.03300	-.00070	.00660
.201	-.990	.47530	.11430	.04340	.48800	.61700	.02895	.01300	.00170	.00350
.201	.050	.47650	.11530	.04330	.48940	.61700	.02977	-.00700	.00260	.00040
.201	.990	.47760	.11480	.04300	.49040	.61700	.02908	-.02800	-.00340	-.00260
.201	1.980	.47940	.11440	.04190	.49200	.61800	.02834	.04900	.00460	.00580
.201	4.040	.48040	.11380	.04120	.49300	.61700	.02759	.08800	.00620	.01290
.201	4.930	.48310	.11550	.04010	.49590	.62000	.02872	-.11100	.00720	-.01640
	GRADIENT	.00089	.00021	-.00010	.00012	.00092	.00005	-.02037	.00091	-.00324

(RFQ020) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA =	10.000	ELEVON =	.000
AIRRON =	.000	RUDDER =	.000
BOFLAP =	-.12.000	SPDBRK =	.000
H/B =	.100		

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0A250, NAAL TEST 775, TABULATED SOURCE DATA  
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(RFQ021) ( 18 NOV 77 )

## REFERENCE DATA

SREF	2690 .0000	SQ.FT.	XMRP	=	1076 .6800	INCHES
LREF	474 .8000	INCHES	YMRP	=	.0000	INCHES
BREF	936 .6600	INCHES	ZMRP	=	375 .0000	INCHES
SCALE	.0299	SCALE				

RUN NO. 21 / 0 RN/L = 1.43 GRADIENT INTERVAL = -6.00 / 6.00

MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
.201	-5.010	.53090	.08130	.00830	.53700	.64400	-.01347	.09100	.00210	.01930
.201	-3.990	.53020	.08110	.00970	.52620	.64300	-.01359	.07200	.00160	.01580
.201	-2.000	.53720	.08290	.00630	.54340	.64500	-.01306	.03500	.00120	.00810
.201	-1.010	.53720	.08300	.00570	.54340	.64600	-.01298	.01600	.00090	.00420
.201	.000	.53890	.08410	.00590	.54530	.64500	-.01218	-.00100	.00080	.00070
.201	.980	.53970	.08350	.00520	.54600	.64600	-.01281	-.01900	.00040	.00290
.201	1.970	.53910	.08270	.00530	.54510	.64600	-.01353	-.03800	.00050	.00670
.201	4.030	.53570	.08160	.00760	.54170	.64400	-.01410	-.07000	.00000	.01500
.201	4.990	.53790	.08220	.00610	.54400	.64500	-.01383	-.09600	-.00050	-.01880
	GRADIENT	.00068	.00007	-.0002+	.00068	.00012	-.00005	-.01861	-.00023	-.00381

0A250 826C9G15M16F8H116E43V8R5TC19

## REFERENCE DATA

SREF	2690 .0000	SQ.FT.	XMRP	=	1076 .6800	INCHES
LREF	474 .8000	INCHES	YMRP	=	.0000	INCHES
BREF	936 .6600	INCHES	ZMRP	=	375 .0000	INCHES
SCALE	.0299	SCALE				

RUN NO. 22 / 0 RN/L = 1.43 GRADIENT INTERVAL = -6.00 / 6.00

MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
.201	-4.990	.63350	.09480	-.04470	.64220	.67500	-.01895	.08900	.00260	.02060
.201	-3.990	.63160	.09470	-.04380	.64140	.67500	-.01891	.07000	.00210	.01670
.201	-2.010	.64170	.09630	-.04750	.64860	.67600	-.01860	.03400	.00150	.00840
.201	-.990	.64330	.09660	-.04820	.65030	.67700	-.01854	.01600	.00090	.00410
.201	.050	.64330	.09750	-.04830	.65040	.67700	-.01765	.00000	.00020	.00000
.201	.980	.64430	.09710	-.04830	.65140	.67700	-.01825	-.01600	-.00060	-.00400
.201	1.980	.64150	.09580	-.04770	.64830	.67700	-.01902	-.03400	-.00090	-.00830
.201	4.040	.63970	.09510	-.04640	.64640	.67600	-.01939	-.07300	-.00160	-.01750
.201	4.990	.64040	.09570	-.04700	.64720	.67600	-.01898	-.09200	-.00180	-.02160
	GRADIENT	.00049	.00006	-.00024	.00049	.00012	-.00003	-.01789	-.00047	-.00423

(RFQ022) ( 18 NOV 77 )

## PARAMETRIC DATA

ALPHA	=	10.000	ELEVON
AIRLON	=	.000	RUDDER
BDFLAP	=	-12.000	SPDBRK
H/B	=	.075	

(RFQ022) ( 18 NOV 77 )

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OA250, NAAL TEST 775, TABULATED SOURCE DATA

OA250 B2659G15M16FBW116E43V8R5TC19

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## REFERENCE DATA

SREF	=	2680.0000	SQ.FT.	XMRP	=	1076.6800	INCHES
LREF	=	.474.8000	INCHES	YMRP	=	.0000	INCHES
BREF	=	.936.6600	INCHES	ZMRP	=	.375.0000	INCHES
SCALE	=	.0299	SCALE				

RUN NO. 23/ 0 RN/L = 1.43

MACH	BETA	CL	CD	CLM	CN	XCP/L	CA	CY	CYN	CBL
.201	-5.000	.41200	.07150	.07050	.41820	.58700	-.00191	.09200	.00210	.01690
.201	-3.990	.40860	.07090	.07260	.41470	.58500	-.00193	.07300	.00150	.01370
.201	-2.000	.41430	.07310	.07010	.42070	.58800	-.00076	.03500	.00120	.00680
.201	-1.000	.41560	.07330	.06920	.42200	.58900	-.00076	.01700	.00080	.00340
.201	.000	.41600	.07390	.06970	.42250	.58900	-.00027	.00100	.00070	.00020
.201	.990	.41830	.07370	.06830	.42480	.59000	-.00091	.01900	.00040	.00310
.201	1.980	.41880	.07310	.06840	.42510	.59000	-.00158	.03800	.00060	.00650
.201	4.040	.41670	.07180	.06980	.42280	.58900	-.00249	.07700	.00050	.01370
.201	4.990	.42110	.07280	.06730	.42730	.59100	-.00225	.09600	.00000	.01720
	GRADIENT	.00097	.00011	-.00034	.00097	.00044	-.00006	-.01871	-.00017	-.00340

## PARAMETRIC DATA

(RFQ023)	( 18 NOV 77 )	
ALPHA	= 10.000	ELEVON = -5.000
AIRON	= .000	RUDDER = .000
BDFLAP	= -12.000	SPDBRK = .000
H/B	= .075	